

TEMPORARY REVISION
TR-MÄM-40-227
1200 kg Maximum Take-Off Mass

This Temporary Revision TR-MÄM-40-227 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-227 and is valid in conjunction with the latest revision of the DA 40 Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

The technical information contained in this document has been verified for EASA by the Austrian Civil Aviation Authority Austro Control (ACG) as Primary Certification Authority (PCA) in accordance with the valid Certification Procedures and approved by EASA with approval no. EASA. A.C. 02792.

Doc. No. 6.01.01-E	TR-MÄM- 40-227	20-Feb-2007	COVER PAGE 1
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DA 40 AFM
DA 40 D AFM



Temporary Revision
Landing Mass 1150 kg

TEMPORARY REVISION

TR-MÄM-40-123/d

supersedes TR-MÄM-40-123/c

Maximum Landing Mass 1150 kg

This Temporary Revision TR-MÄM-40-123/d is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-123/d and is valid in conjunction with the latest revision of the DA 40 or DA 40 D Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

The technical information contained in this document has been approved under the authority of DOA no. EASA.21J.052.

Affected Pages:

Doc. No.	Section	Affected Pages
6.01.01-E	-	Cover Page
6.01.05-E	1	1-2c
	2	2-10a
	4A	4A-29a
	4B	4B-9a

Instruction:

-) Print this document on yellow paper (single-sided).
-) Insert this cover page as the first page of the AFM.
-) Insert the other pages of this Temporary Revision in front of the corresponding AFM pages.

Doc. No. 6.01.01-E	TR-MÄM-	14-Sep-2005	Cover Page
Doc. No. 6.01.05-E	40-123/d		

TEMPORARY REVISION
TR-MÄM-40-175
Procedures for the Use of the
Electrical Fuel Pump

This Temporary Revision TR-MÄM-40-175 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-175 and is valid in conjunction with the latest revision of the DA 40 Airplane Flight Manual until this temporary revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

The technical information contained in this document has been approved under the authority of DOA no. EASA.21J.052

Affected Pages:

Doc. No.	Chapter	Pages
6.01.01-E	3	3-14a
	4A	4A-24a, 4A-26a, 4A-28a

Instruction:

- Print this document on yellow paper (single-sided).
- Insert this cover page as the first page of the AFM.
- Insert the other pages of this Temporary Revision in front of the corresponding AFM pages.

Doc. No. 6.01.01-E	TR-MÄM- 40-175	28-May-2005	Cover Page
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TEMPORARY REVISION
TR-MÄM-40-069
Corrected Take-Off Chart

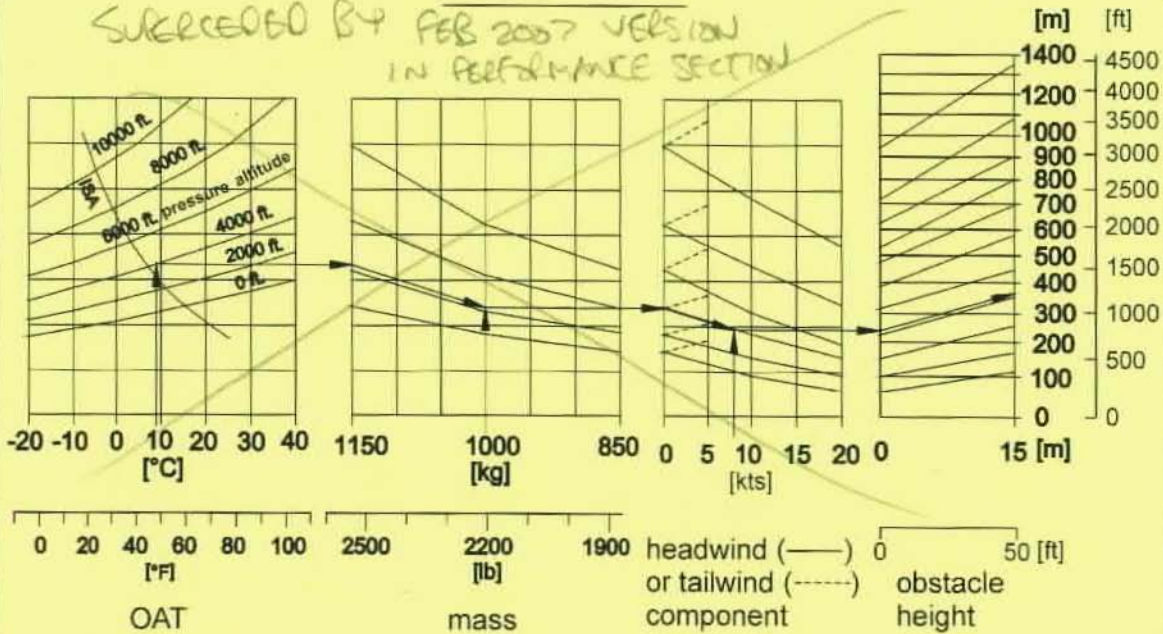
This Temporary Revision TR-MÄM-40-069 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-069 and is valid in conjunction with the Airplane Flight Manual Revision 5.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

Affected Chapters:

5.3.6 TAKE-OFF DISTANCE

In the following graph, the take-off distances from metres to feet have been corrected due to a conversion error in publishing. Moreover, tailwind information has been added.

**Example:**

Pressure altitude : 4000 ft
 OAT : 8 °C (46 °F)
 Mass : 1000 kg (2205 lb)
 Headwind comp. : 8 kts

Result:

T/O roll : approx. 240 m (787 ft)
 T/O distance over 50 ft obstacle : approx. 370 m (1214 ft)

TEMPORARY REVISION
TR-OÄM-40-078
Operation of the DA 40 with Winter Kit

This Temporary Revision TR-OÄM-40-078 is approved in conjunction with the Optional Design Change Advisory OÄM 40-078 and is valid in conjunction with the Airplane Flight Manual Revision 5.

Affected Chapters:

9.2 LIST OF SUPPLEMENTS

The following is added to the List of Supplements:

Airplane S/N: 40.646 Registration: N566DS Date: FEB 23, 2006					
Sup. No.	Title	Rev. No.	Date	applicable	
				YES	NO
O2	Operation of the DA 40 with Winter Kit	0	26-Nov-2001	✓	

Doc. No. 6.01.01-E	TR-OÄM-40-078	26 Nov 2001	Page 1 of 1
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TEMPORARY REVISION**TR-OÄM-40-071/b****Long Range Tank**

This Temporary Revision TR-OÄM-40-071/b is approved in conjunction with the Optional Design Change Advisory OÄM 40-071/b and is valid in conjunction with the Airplane Flight Manual Revision 5.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

AFFECTED CHAPTERS:**1. GENERAL****1.1 INTRODUCTION**

The following Entry must be added to the Modification List:

Modification	Source	Installed	
		yes	no
Long Range Tank	OAM 40-071	✓	

2. OPERATING LIMITATIONS**2.5 ENGINE INSTRUMENT MARKINGS**

Indi- cation	Red arc/bar = lower prohibited range	Yellow arc/bar = caution range	Green arc/bar = normal operating range	Yellow arc/bar = caution range	Red arc/bar = upper prohibited range
Fuel quantity	0 US gal	--	0 - 16 US gal	--	--

2.8 CENTER OF GRAVITY

Most Rearward CG:

2.55 m (100.4 in) aft of DP

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2.14 FUEL

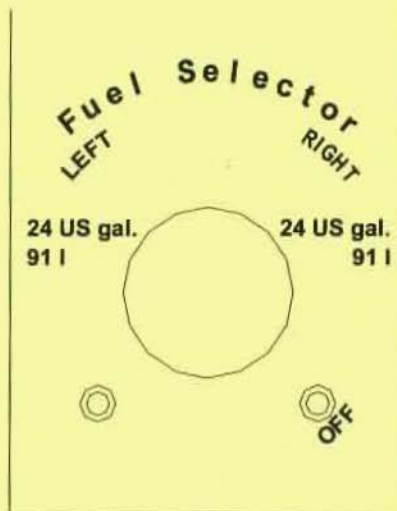
Fuel Quantity: Total fuel quantity	:	2 x 24.5 US gal (approx. 185 liters)
Unusable fuel	:	2 x 0.5 US gal (approx. 3.8 liters)
Max. indicated fuel quantity	:	16 US gal per tank
Indicated auxiliary fuel quantity	:	3 to 9 US gal per tank
Max. permissible difference between right and left tank	:	8 US gal (approx. 30.3 liters)

CAUTION

If a fuel indicator shows 16 US gal and the aux. fuel indicator 0 US gal for the same fuel tank, then 19 US gal must be assumed for the calculation of the difference between right and left tank.

2.15 LIMITATION PLACARDS

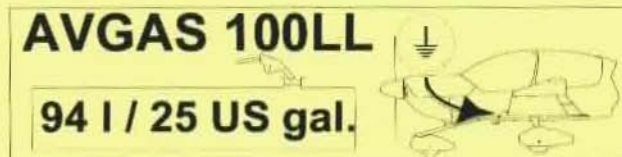
On the fuel tank selector:



Next to the fuel quantity indication:

Fuel qty. indication: 16 + 9 US gal
max. difference LH/RH tank: 8 US gal
AUX FUEL QTY switch for LH/RH auxiliary fuel quantity
NOTE: See AFM for more information on AUX FUEL

Next to each of the two fuel filler necks:



3. EMERGENCY PROCEDURES

No change.

4A. NORMAL OPERATING PROCEDURES**4A.3 CHECKLISTS FOR NORMAL OPERATING PROCEDURES****4.A3.1 PRE-FLIGHT INSPECTION****I. Cabin check**

m) Fuel quantity check

NOTE

When the fuel quantity indicator reads 16 US gal, read the auxiliary fuel quantity by setting the AUX FUEL QTY switch to the appropriate position (LH or RH), and add the auxiliary fuel quantity to the 16 US gal.

NOTE

An auxiliary fuel quantity below 3 US gal cannot be indicated by the system. For auxiliary fuel quantities below 3 US gal, the correct fuel quantity must be determined with the fuel quantity measuring device (see Section 7.10 FUEL SYSTEM).

CAUTION

After selecting a fuel tank the fuel quantity indication will be incorrect for 2 minutes.

4B. ABNORMAL OPERATING PROCEDURES

No change.

5. PERFORMANCE**5.3.2 TABLE FOR SETTING ENGINE PERFORMANCE****NOTE**

An auxiliary fuel below 3 US gal cannot be indicated by the system. If a fuel indicator shows 16 US gal and the auxiliary fuel indicator 0 US gal for the same fuel tank, for in-flight fuel consumption / flight planning a fuel quantity available of 16 US gal must be assumed.

6. MASS AND BALANCE**6.4 FLIGHT MASS AND CENTER OF GRAVITY**

The first paragraph of step 2 of the procedure is amended as follows:

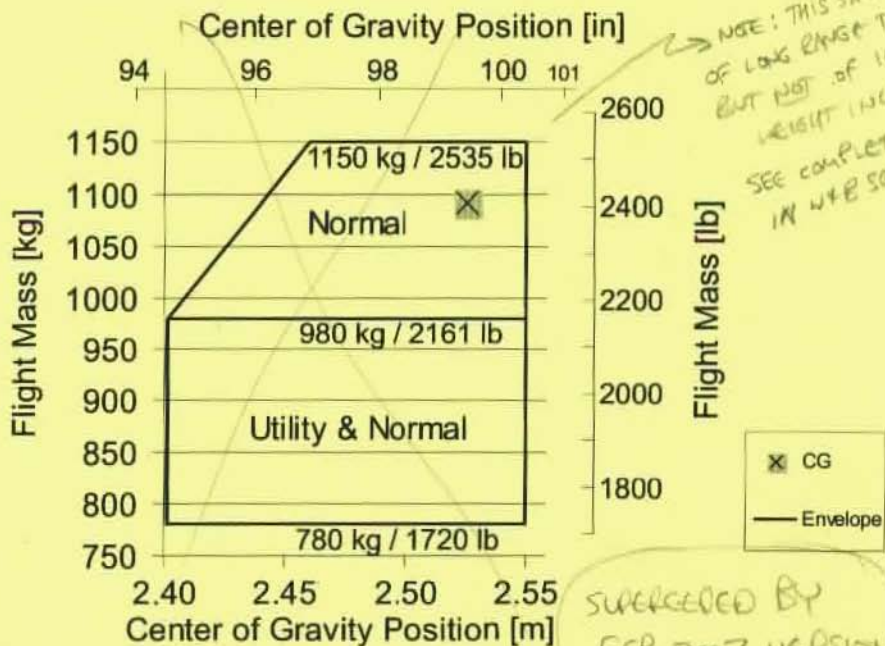
2. Read the fuel quantity indicators to determine the fuel quantity.

NOTE

When the fuel quantity indicator reads 16 US gal, read the auxiliary fuel quantity by setting the AUX FUEL QTY switch to the appropriate position (LH or RH), and add the auxiliary fuel quantity to the 16 US gal.

NOTE

An auxiliary fuel quantity below 3 US gal cannot be indicated by the system. For auxiliary fuel quantities below 3 US gal, the correct fuel quantity must be determined with the fuel quantity measuring device (see Section 7.10 FUEL SYSTEM).

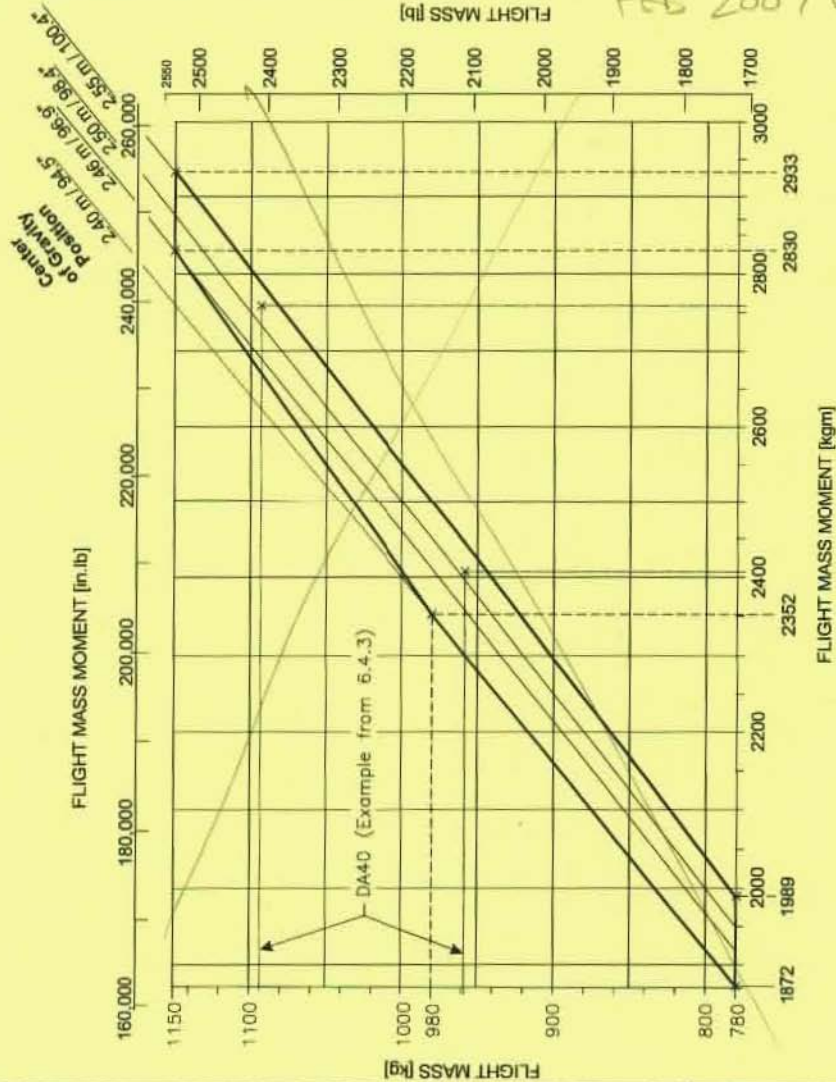
6.4.4 PERMISSIBLE CENTER OF GRAVITY RANGE


Most rearward flight CG:

2.55 m (100.4 in) aft of Datum Plane

*SUPERSEDED BY
FEB 2007 VERSION*

6.4.5 PERMISSIBLE MOMENT RANGE



6.5 EQUIPMENT LIST AND EQUIPMENT INVENTORY

The following items must be added to the Equipment List:

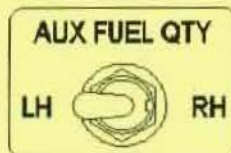
Airplane Serial No.:		Registration:		Date:	
Description	Type	Part No.	Manufacturer	S/N	in- stalled
FUEL					
Fuel qty sensor LH (auxiliary fuel)	VM 1000	3010050	Vision Microsystems		
Fuel qty sensor RH (auxiliary fuel)	VM 1000	3010050	Vision Microsystems		
ENGINE INDICATING					
Data processing unit ¹	DPU	4010067	Vision Microsystem		
Data processing unit ¹	DPU	4010081	Vision Microsystem		
Data processing unit ¹	DPU	4015340	Vision Microsystem		

- One of the following data processing units may be installed:
 DPU with P/N 4010067 or DPU with P/N 4010081 or DPU with P/N 4015340 (MÄM 40-039) with EPROM P/N 3034219 and software revision V1.002. With installed long range tank (OÄM 40-071), the DPU must have EPROM P/N 4015340 and software revision V1.002 containing fuel calibration DA40ER.

7. DESCRIPTION OF THE AIRPLANE AND ITS SYSTEMS

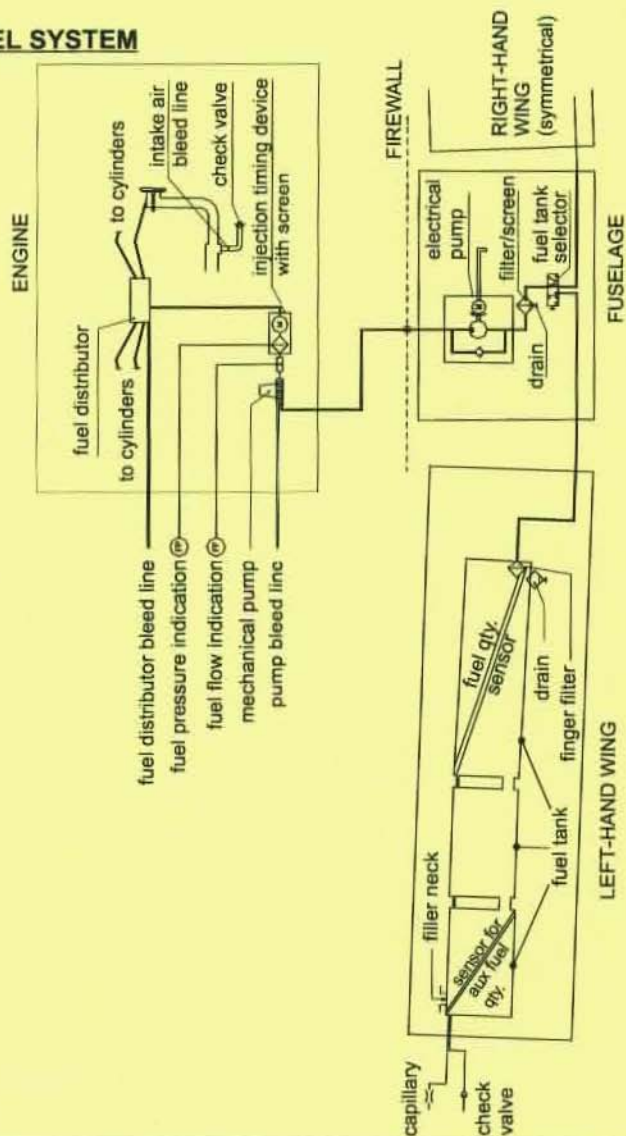
7.4 INSTRUMENT PANEL

The following switch is installed adjacent to the fuel quantity indicator:

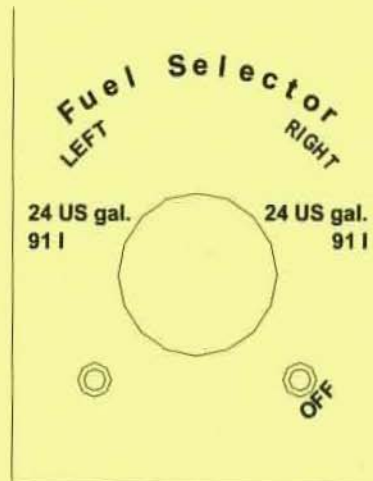


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7.10 FUEL SYSTEM



Fuel Tank Selector



Fuel Tanks

Each of the two wing tanks consists of *three* aluminum chambers which are joined by a piece of flexible hose and two independent vent hoses. There are two separate vents per tank. The hose terminations are situated on the underside of the wing, approx. 2 meters (7 ft) from the wing tip. One vent acts as a capillary, both to equalize the air pressure, and to provide a safety factor in the event of a failure of the other vent. The second vent is a check valve, to allow air to enter the tank, but prevent flow to the outside.

A coarse filter (finger filter) is fitted before the outlet. To allow draining of the tank, there is an outlet valve at its lowest point. A gascolator sits at the lowest point in the fuel system. A drain valve is fitted to the gascolator, which can be used to remove water and sediment which has collected in the fuel system. This valve is fitted centrally on the underside of the fuselage, approximately 30 cm (1 ft) forward of the wing leading edge.

Fuel Quantity Indication

To determine the fuel quantity in the larger tanks an additional capacity probe is installed in each wing tank (LH / RH). When the fuel quantity indicator reads zero, only the unusable fuel remains in the tank. The usable capacity of each tank is 24 US gal (approximately 91 liters).

Fuel quantities up to 16 US gal (approximately 61 liters) in each wing tank (LH / RH) are measured by the standard sensors and indicated on the fuel quantity indicator either on the left or right side in 1 US gal (approximately 4 liters) increments.

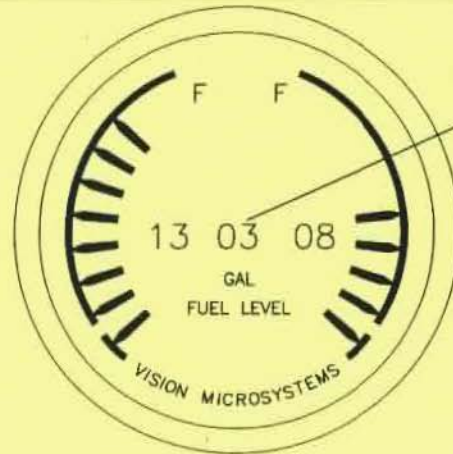
Fuel quantities between 16 US gal (approximately 61 liters) and 24 US gal (approximately 91 liters) are measured by additional sensors and indicated in the center of the fuel quantity indicator. The indication increments are 3 US gal (approximately 11 liters) between 0 and 3 US gal (approximately 11 liters) and 1 US gal (approximately 4 liters) between 3 US gal (approximately 11 liters) and 9 US gal (approximately 34 liters). Which side (LH / RH) is being indicated depends on the position of the AUX FUEL QTY. switch (refer to Section 7.4 - INSTRUMENT PANEL). The indication on the left and right side of the instrument (from 0 US gal to 16 US gal (approximately 61 liters)) will not be affected.

The actual total fuel quantity in each wing tank is the sum of the individual indications.

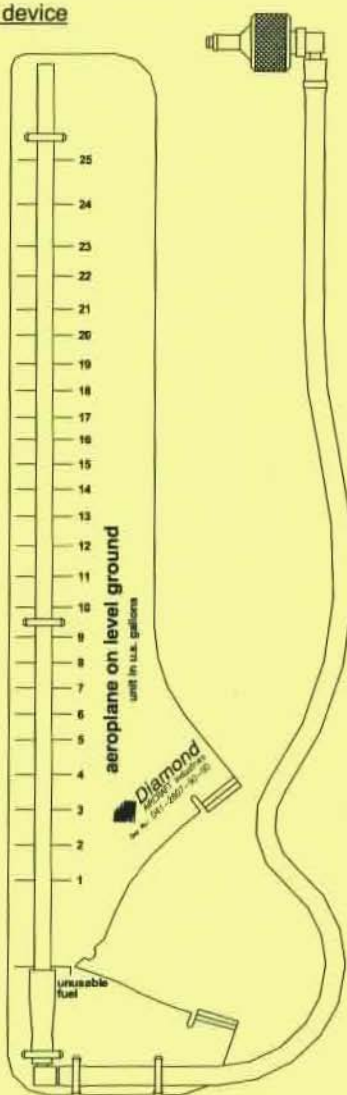
CAUTION

After selecting a fuel tank with the AUX FUEL QTY. switch, the fuel quantity indication will be incorrect for 2 minutes.

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Additional fuel quantity indication between 16 US gal (approximately 61 liters) and 24 US gal (approximately 91 liters) of the wing tank selected with the AUX FUEL QTY. switch..

Fuel quantity measuring device

8. AIRPLANE HANDLING, CARE AND MAINTENANCE

No change.

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TEMPORARY REVISION
TR-MÄM-40-078
Low Temperature Limit

This Temporary Revision TR-MÄM-40-078 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-078. For the operation of the aircraft with the new low temperature limit this TR must be included in the AFM or that AFM revision must be used in which the TR is incorporated.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

Affected Chapters:

2.16 OTHER LIMITATIONS

2.16.1 TEMPERATURE

The airplane may only be operated when its temperature is not less than -40°C (-40°F).

CAUTION

For cold weather starting of the engine refer to the latest instructions given by the engine manufacturer.

TEMPORARY REVISION

TR-MÄM-40-099

P/N Change Autopilot Computer KC 140

This Temporary Revision TR-MÄM-40-099 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-099 and is valid in conjunction with the latest revision of the DA 40 or DA 40 D Airplane Flight Manual until this temporary revision has been incorporated into the Airplane Flight Manuals.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

Affected Chapters:

6.5 EQUIPMENT LIST AND EQUIPMENT INVENTORY

The following items must be added to the Equipment List:

Airplane Serial No.: 40646		Registration: N566DS		Date: FEB 23, 2003	
Description	Type	Part No.	Manufacturer	S/N	In- stalled
AUTOPILOT SYSTEM					
Flight computer ¹	KC 140	065-00176-5403 (post MÄM 40-099 or MSB 40-018)	Bendix/King		
Flight computer ¹	KC 140	065-00176-7703 (post MÄM 40-099 or MSB 40-018)	Bendix/King	9791	X
Flight computer ¹	KC 140	065-00176-5402 (pre MÄM 40-099 or MSB 40-018)	Bendix/King		
Flight computer ¹	KC 140	065-00176-7702 (pre MÄM 40-099 or MSB 40-18)	Bendix/King		

1. One of the following Flight Computers may be installed:
P/N 065-00176-5402, or P/N 065-00176-5403, or P/N 065-00176-7702, or P/N
065-00176-7703.

9.2 LIST OF SUPPLEMENTS

The following supplement must be added to the List of Supplements:

- Supplement A13, Autopilot System, KAP 140, Revision 1 or later

TEMPORARY REVISION TR-MÄM-40-133 Canopy Operating Procedure

This Temporary Revision TR-MÄM-40-133 is approved in conjunction with the Mandatory Design Change Advisory MÄM 40-133 and is valid in conjunction with the latest revision of the DA 40 or DA 40 D Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

Affected Chapters:

4A.3.2 BEFORE STARTING ENGINE

After Item 7.

4A.3.6 BEFORE TAKE-OFF

After Item 5.

CAUTION

When operating the canopy, pilots / operators are to ensure that there are no obstructions between the canopy and the mating frame, for example seat belts, clothing, etc. When operating the locking handle do NOT apply undue force.

A slight downward pressure on the canopy may be required to ease the handle operation.

TEMPORARY REVISION
TR-OÄM-40-124
Main Landing Gear Tire 15x6.0-6

This Temporary Revision TR-OÄM-40-124 is approved in conjunction with the Optional Design Change Advisory OÄM 40-124 and is valid in conjunction with the latest revision of the DA 40 or DA 40 D Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

Affected Chapters:

1.4 DIMENSIONS

Landing gear

- Main wheel :
- (a) 6.00-6; 6 PR, 120 mph or
 - (b) 15x6.0-6, 6 PR, 120 mph (OÄM 40-124; approved only in combination with MÄM 40-123, main landing gear strut with 18 mm / 0.71" thickness)

TEMPORARY REVISION
TR-OÄM-40-190
Turn Coordinator 1394T100-(12RA)

This Temporary Revision TR-OÄM-40-190 is approved in conjunction with the Optional Design Change Advisory OÄM 40-190 and is valid in conjunction with the latest revision of the DA 40 Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

The technical information contained in this document has been approved under the authority of DOA No. EASA.21J.052.

Affected Chapters:

6.5 EQUIPMENT LIST AND EQUIPMENT INVENTORY

The following items must be added to the Equipment List:

Airplane Serial No.: 40.646		Registration: N566DS		Date: FEB 23, 2006	
Description	Type	Part No.:	Manufacturer	S/N	in-stalled
NAVIGATION Turn coordinator	1394T100-(12RA)		Mid Continent Instr.	305-11265	X



AIRPLANE FLIGHT MANUAL

DA 40

Airworthiness Category : Normal, Utility
Requirement : JAR-23
Serial Number : 40.646
Registration : N566DS
Doc. No. : 6.01.01-E
Date of Issue : 26 June 2000

Signature :

Authority :

Stamp :

Date of approval :



AUSTRO CONTROL GmbH
Abteilung Flugtechnik
Wien
A-1030 Wien, Schairchgasse 11
09. DEZ. 2004

- | This Flight Manual has been verified for EASA by the Austrian Civil Aviation Authority Austro Control (ACG) as Primary Certification Authority (PCA) in accordance with the valid Certification Procedures and approved by EASA with approval no. 2004-12326
- | This Flight Manual has been approved by EASA on behalf of CAAC-AAD.

DIAMOND AIRCRAFT INDUSTRIES GMBH
N.A. OTTO-STR. 5
A-2700 WIENER NEUSTADT
AUSTRIA

FOREWORD

We congratulate you on the acquisition of your new DIAMOND STAR.

Skillful operation of an airplane increases both safety and the enjoyment of flying. Please take the time therefore, to familiarize yourself with your new DIAMOND STAR.

This airplane may only be operated in accordance with the procedures and operating limitations of this Airplane Flight Manual.

Before this airplane is operated for the first time, the pilot must familiarize himself with the complete contents of this Airplane Flight Manual.

In the event that you have obtained your DIAMOND STAR second-hand, please let us know your address, so that we can supply you with the publications necessary for the safe operation of your airplane.

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**COVER PAGE TO
TEMPORARY REVISION
TR-OÄM-40-183
Winter Baffle Fresh Air Inlet**

This Temporary Revision TR-OÄM-40-183 is approved in conjunction with the Optional Design Change Advisory OÄM 40-183 and is valid in conjunction with the latest revision of the DA 40 or DA 40 D Airplane Flight Manual until this Temporary Revision has been incorporated into the Airplane Flight Manual.

The limitations and information contained herein either supplement or, in the case of conflict, override those in the Airplane Flight Manual.

The technical information contained in this document has been approved under the authority of DOA no. EASA.21J.052

Affected Pages:

Doc. No.	Rev.	Page
6.01.01-E	5	0-1a, 1-2a, 4A-5a, 6-12a, 9-3a
6.01.05-E	3	0-1a, 1-2a, 4A-5a, 6-15a, 9-3a

Instruction:

Insert pages of Temporary Revision in front of original manual page.

Insert this cover page as first page of the manual.

Doc. No. 6.01.01-E	TR-OÄM-40-183	22-Nov-2004	Page 0-1a
Doc. No. 6.01.05-E			

9.2 LIST OF SUPPLEMENTS

The Supplement E7 must be added to the List of Supplements.

Airplane S/N: 40.646		Registration: N56605		Date: FEB 23, 2006	
Sup. No.	Title	Rev. No.	Date	applicable	
				YES	NO
E7	Winter Baffle Fresh Air Inlet		04-Nov-2004	✓	

0.1 RECORD OF REVISIONS

All revisions of this manual, with the exception of -

- Temporary Revisions,
- updates of the modification level (Section 1.1),
- updated mass and balance information (Section 6.3),
- updates of the Equipment Inventory (Section 6.5), and
- updates of the List of Supplements (Section 9.2)

must be recorded in the following table. Revisions of approved Chapters require the countersignature of Austro Control GmbH.

The new or amended text is indicated by a vertical black line at the left hand side of the revised page, with the revision number and date appearing at the bottom of the page.


If pages are revised which contain information valid for your particular serial number (modification level of the airplane, weighing data, Equipment Inventory, List of Supplements), then this information must be transferred to the new pages in hand-writing.

Temporary Revisions, if applicable, are inserted behind the cover page of this manual. Temporary Revisions are used to provide information on systems or equipment until the next 'permanent' Revision of the Airplane Flight Manual. When a 'permanent' Revision covers a Mandatory or Optional Design Change Advisory (MÄM or OÄM), then the corresponding Temporary Revision is superseded. For example: Revision 5 covers OÄM 40-039, therefore the Temporary Revision TR-OÄM-40-039 is superseded by the 'permanent' Revision 5.

Rev. No.	Reason	Chapter	Page(s)	Date of Revision	Approval	Date of Approval	Date Inserted	Signature
1	corrections	all	all	26-Sep-2000	[approved by Ing. Andreas Winkler for ACG]	09-Oct-2000		

Rev. No.	Reason	Chapter	Page(s)	Date of Revision	Approval	Date of Approval	Date Inserted	Signature
2	OÄM 40-060 (White Wire optional)	0	0-2, 0-4 thru 0-7	19-Dec-2000	[approved by Ing. Andreas Winkler for ACG]	25-Jan-2001		
		1	1-16					
	2	2-1, 2-7 thru 2-9, 2-13 thru 2-19						
	3	3-7, 3-8, 3-19, 3-20, 3-25, 3-26						
	4A	4A-3 thru 4A-8, 4A-14, 4A-15						
	4B	4B-4 thru 4B-6						
	6	6-1, 6-2, 6-12 thru 6-14						
	OÄM 40-073 (LASAR optional) corrections	7	7-1, 7-8, 7-14, 7-28 thru 7-38					
3	OÄM 40-064 (Night VFR)	0	0-2 thru 0-7	05-Feb-2001	[approved by Ing. Andreas Winkler for ACG]	02 Jul 2001		
		1	1-2					
	2	2-1, 2-8, 2-9, 2-12, 2-15 thru 2-20						
	3	3-1, 3-25 thru 3-27						
	4A	4A-1, 4A-8 thru 4A-31						
	5	5-7, 5-14, 5-16						
	6	6-7, 6-9, 6-12 thru 6-14						
	7	7-32, 7-35, 7-36						
	OÄM 40-070 (tow bar) corrections	8	8-1 thru 8-9					
4	OÄM 40-057 (IFR) corrections	all	all	09 Apr 2001	[approved by Ing. Andreas Winkler for ACG]	02 Jul 2001		

Rev. No.	Reason	Chapter	Page(s)	Date of Revision	Approval	Date of Approval	Date Inserted	Signature
5	OAM 40-061 (KAP 140 autopilot)	0	0-1 thru 0-8	09 Sep 2001				
	OAM 40-073 (Slick-START)	1	1-2, 1-5, 1-14					
	OAM 40-081 (door lock)	2	2-1, 2-16, 2-22, 2-23, 2-24					
	OAM 40-085 (OX 155A as COM 1)	3	3-13, 3-18, 3-22, 3-23, 3-24, 3-31, 3-36					
	OAM's 40-092 thru 40-094 (Mikrotechna ASI, altimeter, VSI)	4A	4A-8, 4A-10, 4A-22, 4A-23, 4A-26					
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CHAPTER 1

GENERAL

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1.1 INTRODUCTION

This Airplane Flight Manual has been prepared in order to provide pilots and instructors with all the information required for the safe and efficient operation of the airplane.

The Airplane Flight Manual includes all the data which must be made available to the pilot according to the JAR-23 requirement. Beyond this, it contains further data and operating instructions which, in the manufacturer's opinion, could be of value to the pilot.

This Airplane Flight Manual is valid for all serial numbers. Equipment and modification level (design details) of the airplane may vary from serial number to serial number. Therefore, some of the information contained in this manual is applicable depending on the respective equipment and modification level. The exact equipment of your serial number is recorded in the Equipment Inventory in Section 6.5. The modification level is recorded in the following table (as far as necessary for this manual).

Modification	Source	Installed	
Autopilot	OÄM 40-061	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
Emergency switch	OÄM 40-067	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
Essential Bus	OÄM 40-068	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
Alternate Static Valve	OÄM 40-072	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
SlickSTART Ignition System	OÄM 40-073	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
Door Locking System	OÄM 40-081	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
RH Emergency Window	MÄM 40-048	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no

1. GENERAL

INTRODUCTION

The following entry must be added to the Modification List:

Modification	Source	Installed	
		yes	no
Winter Baffle Fresh Air Inlet	OÄM 40-183	yes	no

Affected Chapters:

1.1 INTRODUCTION

[...]

Modification	Source	Installed	
[...]			
Modified MLG-Strut	MÂM 40-123	<input checked="" type="checkbox"/> yes	<input type="checkbox"/> no
[...]			

[...]

AFFECTED CHAPTERS:**1.1 INTRODUCTION**

The following line is inserted in the existing table:

Modification	Source	Installed
1200 kg Maximum Take-Off Mass	MÄM 40-227	<input checked="" type="checkbox"/> yes <input type="checkbox"/> no

This Airplane Flight Manual must be kept on board the airplane at all times. Its designated place is the side bag of the forward left seat.

This Airplane Flight Manual constitutes an FAA Approved Airplane Flight Manual for US registered airplanes in accordance with FAA regulation 14 CFR, Part 21.29.

CAUTION

The DA 40 is a single engine airplane. When the operating limitations and maintenance requirements are complied with, it has the high degree of reliability which is required by the certification basis. Nevertheless, an engine failure is not completely impossible. For this reason, flights during the night, on top, under instrument meteorological conditions (IMC), or above terrain which is unsuitable for a landing, constitute a risk. It is therefore highly recommended to select flight times and flight routes such that this risk is minimized.

1.2 CERTIFICATION BASIS

This airplane has been type certified in accordance with the JAA JC/VP procedure. The certification basis is JAR-23, published on 11-Mar-1994.

1.3 WARNINGS, CAUTIONS AND NOTES

Special statements in the Airplane Flight Manual concerning the safety or operation of the airplane are highlighted by being prefixed by one of the following terms:

WARNING

means that the non-observation of the corresponding procedure leads to an immediate or important degradation in flight safety.

CAUTION

means that the non-observation of the corresponding procedure leads to a minor or to a more or less long term degradation in flight safety.

NOTE

draws the attention to any special item not directly related to safety but which is important or unusual.

1.4 DIMENSIONS

Overall dimensions

Span	:	appr. 11.94 m	appr. 39 ft 2 in
Length	:	appr. 8.01 m	appr. 26 ft 3 in
Height	:	appr. 1.97 m	appr. 6 ft 6 in

Wing

Airfoil	:	Wortmann FX 63-137/20 - W4	
Wing Area	:	appr. 13.54 m ²	appr. 145.7 sq.ft.
Mean aerodynamic chord (MAC)	:	appr. 1.121 m	appr. 3 ft 8.1 in
Aspect ratio	:	appr. 10.53	
Dihedral	:	appr. 5°	
Leading edge sweep	:	appr. 1°	

Aileron

Area (total, left + right)	:	appr. 0.654 m ²	appr. 7.0 sq.ft.
----------------------------	---	----------------------------	------------------

Wing flaps

Area (total, left + right)	:	appr. 1.56 m ²	appr. 16.8 sq.ft.
----------------------------	---	---------------------------	-------------------

Horizontal tail

Area	:	appr. 2.34 m ²	appr. 25.2 sq.ft.
Elevator area	:	appr. 0.665 m ²	appr. 7.2 sq.ft.
Angle of incidence	:	appr. -3.0° relative to longitudinal axis of airplane	

Vertical tail

Area	:	appr. 1.60 m ²	appr. 17.2 sq.ft.
Rudder area	:	appr. 0.47 m ²	appr. 5.1 sq.ft.

Landing gear

Track	:	appr. 2.97 m	appr. 9 ft 9 in
Wheelbase	:	appr. 1.68 m	appr. 5 ft 6 in
Nose wheel	:	5.00-5; 6 PR, 120 mph	
Main wheel	:	6.00-6; 6 PR, 120 mph	

1.5 DEFINITIONS AND ABBREVIATIONS

(a) Airspeeds

CAS: Calibrated Airspeed. Indicated airspeed, corrected for installation and instrument errors. CAS equals TAS at standard atmospheric conditions at MSL.

KCAS: CAS in knots.

IAS: Indicated Airspeed as shown on an airspeed indicator.

KIAS: IAS in knots.

TAS: True Airspeed. The speed of the airplane relative to the air. TAS is CAS corrected for errors due to altitude and temperature.

V_A : Maneuvering Speed. Full or abrupt control surface movement is not permissible above this speed.

V_{FE} : Max. Flaps Extended Speed. This speed must not be exceeded with the given flap setting.

V_{NE} : Never Exceed Speed in smooth air. This speed must not be exceeded in any operation.

- V_C : Maximum Structural Cruising Speed. This speed may be exceeded only in smooth air, and then only with caution.
- V_S : Stalling Speed, or the minimum continuous speed at which the airplane is still controllable in the given configuration.
- V_{S0} : Stalling Speed, or the minimum continuous speed at which the airplane is still controllable in the landing configuration.
- V_x : Best Angle-of-Climb Speed.
- V_y : Best Rate-of-Climb Speed.

(b) Meteorological terms

ISA: International Standard Atmosphere. Conditions at which air is identified as an ideal dry gas. The temperature at mean sea level is 15 °C (59 °F), air pressure at MSL is 1013.25 hPa (29.92 inHg); the temperature gradient up to the altitude at which the temperature reaches -56.5 °C (-69.7 °F) is -0.0065 °C/m (-0.00357 °F/ft), and above this 0 °C/m (0 °F/ft).

MSL: Mean Sea Level.

OAT: Outside Air Temperature.

QNH: Theoretical atmospheric pressure at MSL, calculated from the elevation of the measuring point above MSL and the actual atmospheric pressure at the measuring point.

Indicated Pressure Altitude:

Altitude reading with altimeter set to 1013.25 hPa (29.92 inHg).

Pressure Altitude:

Altitude above MSL, indicated by a barometric altimeter which is set to 1013.25 hPa (29.92 inHg). The Pressure Altitude is the Indicated Pressure Altitude corrected for installation and instrument errors.

In this Airplane Flight Manual altimeter instrument errors are regarded as zero.

Density Altitude:

Altitude in ISA conditions at which the air density is equal to the current air density.

Wind: The wind speeds which are shown as variables in the diagrams in this manual should be regarded as headwind or downwind components of the measured wind.

(c) Flight performance and flight planning**Demonstrated Crosswind Component:**

The speed of the crosswind component at which adequate maneuverability for take-off and landing has been demonstrated during type certification.

MET: Weather, weather advice.

NAV: Navigation, route planning.

(d) Mass and balance

DP: Datum Plane; an imaginary vertical plane from which all horizontal distances for center of gravity calculations are measured.

Moment Arm:

The horizontal distance from the Datum Plane to the Center of Gravity of a component.

Moment: The mass of a component multiplied by its moment arm.

CG: Center of Gravity, also called 'center of mass'. Imaginary point in which the airplane mass is assumed to be concentrated for mass and balance calculations. Its distance from the Datum Plane is equal to the Center of Gravity Moment Arm.

Center of Gravity Moment Arm:

The Moment Arm which is obtained if one divides the sum of the individual moments of the airplane by its total mass.

Center of Gravity Limits:

The Center of Gravity range within which the airplane, at a given mass, must be operated.

Usable Fuel:

The quantity of fuel available for flight planning.

Unusable Fuel:

The quantity of fuel remaining in the tank which cannot be used for flight.

Empty Mass:

The mass of the airplane including unusable fuel, all operating consumables and the maximum quantity of oil.

Useful Load:

The difference between take-off mass and empty mass.

Maximum Take-off Mass:

The maximum permissible mass for take-off.

Maximum Landing Mass:

The highest mass for landing conditions at the maximum descent velocity. This velocity was used in the strength calculations to determine the landing gear loads during a particularly hard landing.

(e) Engine**Take-off Power:**

Maximum permissible engine output power for take-off.

Maximum Continuous Power:

Maximum permissible engine output power used continuously during flight.

CHT: Cylinder Head Temperature.

EGT: Exhaust Gas Temperature.

(f) Designation of the circuit breakers on the instrument panel**AVIONICS:**

ADF	Automatic Direction Finder
AUDIO	Audio Panel / Intercom
AUTOPILOT	Autopilot
AVIONIC BUS	Avionic Bus
DME	Distance Measuring Equipment
ESSENTIAL AVIONIC	Essential Avionic Bus
GPS	Global Positioning System
NAV/COM	Navigation/Communication Equipment
XPDR	Transponder

ENGINE:

IGNITION	Ignition
INST. 1	Engine Instrument VM 1000
START	Starter

LIGHTING:

FLOOD	Flood Light
INST.	Instrument Lights
LANDING	Landing Light
POSITION	Position Lights
STROBE	Strobe Light (= Anti Collision Light = ACL)
TAXI/MAP	Taxi Light/Map Light

SYSTEMS:

ANNUN.	Annunciator Panel
DG	Directional Gyro
FAN/OAT	Fan/Outside Air Temperature Indicator
FLAPS	Flaps
FUEL PUMP	Fuel pump
HORIZON	artificial horizon (attitude gyro)
PITOT HEAT	Pitot Heating System
T&B	Turn & Bank Indicator

ELECTRICAL:

ALT.	Alternator
ALT. PROT.	Alternator Protection
ALT. CONT.	Alternator Control
BATT.	Battery
ESSENTIAL TIE	Bus Interconnection
MAIN TIE	Bus Interconnection
MASTER CONTROL	Master Control (avionic master switch, essential bus switch, essential avionics relay, bus interconnection relay, avionics master relay).

(g) Equipment

ELT: Emergency Locator Transmitter.

(h) Design Change Advisories

MÄM: Mandatory Design Change Advisory.

OÄM: Optional Design Change Advisory.

(i) Miscellaneous

ACG: Austro Control GmbH (formerly BAZ, Federal Office of Civil Aviation).

ATC: Air Traffic Control.

CFRP: Carbon Fiber Reinforced Plastic.

GFRP: Glass Fiber Reinforced Plastic.

JAR: Joint Aviation Requirements.

JC/VP: Joint Certification/Validation Procedure.

PCA: Primary Certification Authority.

1.6 UNITS OF MEASUREMENT

1.6.1 CONVERSION FACTORS

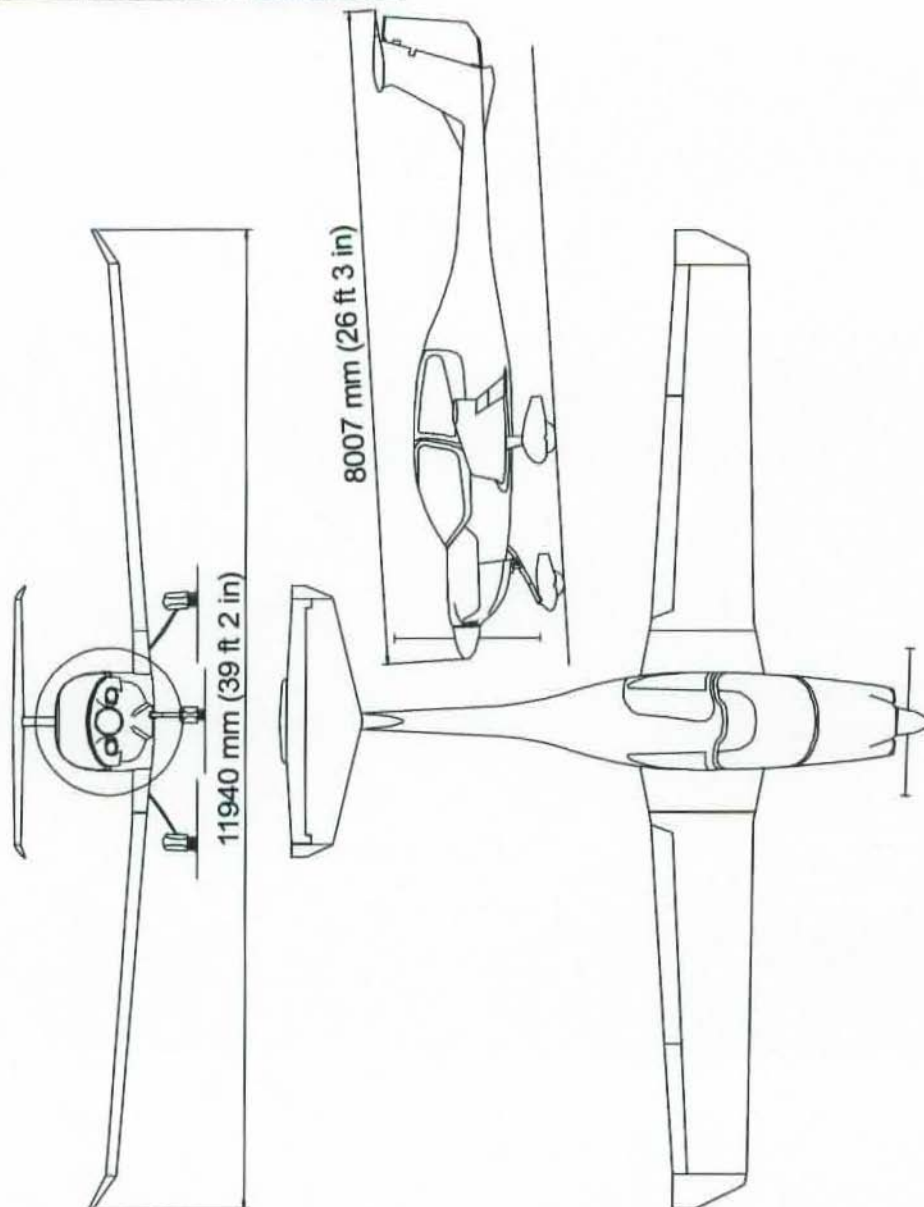
Dimension	SI-Units	US Units	Conversion
Length	[mm] millimeters	[in] inches	$[mm] / 25.4 = [in]$
	[m] meters	[ft] feet	$[m] / 0.3048 = [ft]$
	[km] kilometers	[NM] nautical miles	$[km] / 1.852 = [NM]$
Volume	[l] liters	[US gal] US gallons	$[l] / 3.7854 = [US\ gal]$
		[qts] US quarts	$[l] / 0.9464 = [qts]$
Speed	[km/h] kilometers per hour	[kts] knots	$[km/h] / 1.852 = [kts]$
	[m/s] meters per second	[mph] miles per hour	$[km/h] / 1.609 = [mph]$
		[fpm] feet per minute	$[m/s] \times 196.85 = [fpm]$
Speed of rotation	[RPM] revolutions per minute		--
Mass	[kg] kilograms	[lb] pounds	$[kg] \times 2.2046 = [lb]$
Force, weight	[N] newtons	[lbf] pounds force	$[N] \times 0.2248 = [lbf]$
Pressure	[hPa] hecto-pascals	[inHg] inches of mercury	$[hPa] = [mbar]$
	[mbar] millibars	[psi] pounds per square inch	$[hPa] / 33.86 = [inHg]$
	[bar] bars		$[bar] \times 14.504 = [psi]$
Temperature	[°C] degrees Celsius	[°F] degrees Fahrenheit	$[°C] \times 1.8 + 32 = [°F]$ $([°F] - 32) / 1.8 = [°C]$

Dimension	SI-Units	US Units	Conversion
Intensity of electric current	[A] ampères		--
Electric charge (battery capacity)	[Ah] ampère-hours		--
Electric potential	[V] volts		--
Time	[sec] seconds		--

1.6.2 CONVERSION CHART LITERS / US GALLONS

Liters	US Gallons
5	1.3
10	2.6
15	4.0
20	5.3
25	6.6
30	7.9
35	9.2
40	10.6
45	11.9
50	13.2
60	15.9
70	18.5
80	21.1
90	23.8
100	26.4
110	29.1
120	31.7
130	34.3
140	37.0
150	39.6
160	42.3
170	44.9
180	47.6

US Gallons	Liters
1	3.8
2	7.6
4	15.1
6	22.7
8	30.3
10	37.9
12	45.4
14	53.0
16	60.6
18	68.1
20	75.7
22	83.3
24	90.9
26	98.4
28	106.0
30	113.6
32	121.1
34	128.7
36	136.3
38	143.8
40	151.4
45	170.3
50	189.3

1.7 THREE-VIEW DRAWING

1.8 SOURCE DOCUMENTATION

This section lists documents, manuals and other literature that were used as sources for the Airplane Flight Manual, and indicates the respective publisher. However, only the information given in the Airplane Flight Manual is valid.

1.8.1 ENGINE

Address: Textron Lycoming
652 Oliver Street
WILLIAMSPORT, PA 17701
USA

Phone: +1-570-323-6181

Documents: a) Textron Lycoming Operator's Manual, Aircraft Engines
60297-12 (Part No.)
b) Service Bulletins (SB)
Service Instructions (SI); (e.g. SI 1014, SI 1070)
Service Letters (SL); (e.g. SL114 (subscriptions))

1.8.2 PROPELLER

Address: mt-propeller
Airport Straubing Wallmühle
D-94348 ATTING
GERMANY

Phone: +49-9429-9409-0

E-mail: sales@mt-propeller.com

Internet: www.mt-propeller.de

Documents: E-124, Operation and Installation Manual
Hydraulically controlled variable pitch propeller
MTV -5, -6, -9, -11, -12, -14, -15, -16, -21, -22, -25

1.8.3 ENGINE INSTRUMENTS

Address: VISION MICROSYSTEMS, INC.
ADVANCED ELECTRONIC INSTRUMENTATION
4071 Hannegan Road, Suite T
BELLINGHAM, WA 98226
USA

Phone: +1-360-714-8203

Documents: 5010002 REV F, VM 1000 Owner's Manual

1.8.4 IGNITION CONTROL UNIT

The electronic ignition control unit LASAR is optional equipment.

Address: UNISON Industries
7575 Baymeadows Way
JACKSONVILLE, FL 32256
USA

Phone: +1-904-739-4066

Internet: www.unisonindustries.com

Documents: L-1502
LASAR Installation, Operation, and Troubleshooting Manual

CHAPTER 2

OPERATING LIMITATIONS

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2.1 INTRODUCTION

Chapter 2 of this Airplane Flight Manual includes operating limitations, instrument markings, and placards necessary for safe operation of the airplane, its power-plant, standard systems and standard equipment.

The limitations included in this Chapter are approved.

WARNING

Operation of the airplane outside of the approved operating limitations is not permissible.

2.2 AIRSPEED

The first line of the table is amended to read:

	Airspeed	IAS	Remarks
v _A	Maneuvering speed	108 KIAS (above 980 kg / 2161 lb up to 1150 kg / 2535 lb) 94 KIAS (780 kg / 1720 lb up to 980 kg / 2161 lb) If MÄM 40-227 is carried out: 111 KIAS (above 1036 kg / 2284 lb up to 1200 kg / 2646 lb) 94 KIAS (780 kg / 1720 lb up to 1036 kg / 2284 lb)	Do not make full or abrupt control surface movement above this speed. → w/ 2 PEOPLE 200LB EACH + 15 GAL FUEL

2.2 AIRSPEED

	Airspeed	IAS	Remarks
V_A	Maneuvering speed	108 KIAS (above 980 kg / 2161 lb up to 1150 kg / 2535 lb) 94 KIAS (780 kg / 1720 lb up to 980 kg / 2161 lb)	Do not make full or abrupt control surface movement above this speed.
V_{FE}	Max. flaps extended speed	LDG: 91 KIAS T/O: 108 KIAS	Do not exceed these speeds with the given flap setting.
V_C	Max. structural cruising speed	<u>129 KIAS</u>	Do not exceed this speed except in smooth air, and then only with caution.
V_{NE}	Never exceed speed in smooth air	<u>178 KIAS</u>	Do not exceed this speed in any operation.

2.3 AIRSPEED INDICATOR MARKINGS

Marking	IAS	Significance
White arc	49 KIAS - 91 KIAS	Operating range with flaps fully extended
Green arc	52 KIAS - 129 KIAS	Normal operating range.
Yellow arc	129 KIAS - 178 KIAS	'Caution' range - "Only in smooth air".
Red line	178 KIAS	Maximum speed for all operations - V_{NE} .

2.4 POWER-PLANT LIMITATIONS

- a) Engine manufacturer : Textron Lycoming
- b) Engine designation : IO-360 M1-A
- c) RPM limitations
- Max. take-off RPM : 2700 RPM
 - Max. continuous RPM : 2400 RPM
- d) Manifold pressure limitations
- Maximum : FULL throttle
- e) Oil pressure
- Minimum (IDLE) : 25 psi / 1.72 bar
 - Maximum : 98 psi / 6.76 bar
 - Normal operating range : 55 to 95 psi / 3.8 to 6.55 bar
- f) Oil quantity
- Minimum : 4 qts
 - Maximum : 8 qts
- g) Oil temperature
- Maximum : 245 °F (118 °C)
- h) Fuel pressure
- Minimum : 14 psi / 0.97 bar
 - Maximum : 35 psi / 2.4 bar
- i) Cylinder head temperature
- Maximum : 500 °F (260 °C)

- j) Propeller manufacturer : mt-Propeller
- k) Propeller designation : MTV-12-B/180-17
- l) Propeller diameter : 1.80 m (+ 0 mm, - 50 mm)
5 ft 10.9 in (+ 0.0 in, - 2.0 in)
- m) Propeller pitch angle (0.75 R) : 10.5° to 30°
- n) Oil specification:

Airplane engine oil should be used which meets SAEJ1899 (MIL-L-22851) Standard (ashless dispersant type). During the first 50 hours of operation of a new or newly overhauled engine, or after replacement of a cylinder, airplane engine oil should be used which meets SAEJ1966 (MIL-L-6082) Standard (straight mineral type). The viscosity should be selected according to the recommendation given in the following table:

OAT at ground level	<i>During the first 50 hours:</i>	<i>After 50 hours:</i>
	SAEJ1966 / MIL-L-6082 Mineral Oil	SAEJ1899 / MIL-L-22851 Ashless Dispersant Oil
All temperatures	---	SAE 15-W50, SAE 20-W50
above 80 °F (above 27 °C)	SAE 60	SAE 60
above 60 °F (above 16 °C)	SAE 50	SAE 40 or SAE 50
30 °F to 90 °F (-1 °C to 32 °C)	SAE 40	SAE 40
0 °F to 90 °F (-18 °C to 32 °C)	SAE 20-W50	SAE 20-W50 or SAE 15-W50
0 °F to 70 °F (-18 °C to 21 °C)	SAE 30	SAE 30, SAE 40, or SAE 20-W40
below 10 °F (below -12 °C)	SAE 20	SAE 30 or SAE 20-W30

2.5 ENGINE INSTRUMENT MARKINGS

Engine instrument markings and their color code significance are shown in the table below:

NOTE

When an indication lies in the upper or lower prohibited range, the numerical indication will begin flashing as well.

Indi- cation	Red arc/bar = lower prohibited range	Yellow arc/bar = caution range	Green arc/bar = normal operating range	Yellow arc/bar = caution range	Red arc/bar = upper prohibited range
Manifold Pressure	--	--	13 - 30 inHg	--	--
RPM	--	--	500 - 2400 RPM	2400 - 2700 RPM	> 2700 RPM
Oil temp.	--	--	149 - 230 °F	231 - 245 °F	> 245 °F
Cylinder head temp.	--	--	150 - 475 °F	476 - 500 °F	> 500 °F
Oil pressure	< 25 psi	25 - 55 psi	56 - 95 psi	96 - 97 psi	> 97 psi
Fuel Pressure	< 14 psi	--	14 - 35 psi	--	> 35 psi
Fuel Flow	--	--	1 - 20 US gal/hr	--	> 20 US gal/hr
Voltage	< 24.1 V	24.1 - 25 V	25.1 - 30 V	30.1 - 32 V	> 32 V
Ammeter	--	--	2 - 75 A	--	--
Fuel quantity	0 US gal	--	0 - 15 US gal	--	--

2.6 WARNING, CAUTION AND STATUS LIGHTS

The following tables show the color and significance of the warning, caution and status lights on the annunciator panel. There are two variants of the annunciator panel ('DAI' and 'White Wire', see Section 7.11).

NOTE

Section 7.11 includes a detailed description of the lights on the annunciator panel.

Color and significance of the warning lights (red)

Warning lights (red)			Cause
Variant 'DAI'	Variant 'White Wire'	Meaning	
OIL PR	OIL PRESS	oil pressure	Oil pressure below 25 psi
FUEL PR	FUEL PRESS	fuel pressure	Fuel pressure below 14 psi
ALT	ALTERNATOR	alternator (generator)	Alternator failure
START	START	starter	Operation of starter, or failure of the starter motor to disengage from the engine after starting
DOOR	DOORS	doors	Front canopy and/or rear door not completely closed and locked
	TRIM FAIL	trim failure	Failure in the automatic trim system of the autopilot (if installed)

Color and significance of the caution lights (amber)

Caution lights (amber)			Cause
Variant 'DAI'	Variant 'White Wire'	Meaning	
L FUEL		fuel quantity left tank	Fuel quantity in the left tank less than 3 US gal (± 1 US gal)
R FUEL		fuel quantity right tank	Fuel quantity in the right tank less than 3 US gal (± 1 US gal)
	LOW FUEL	fuel quantity	1 st caution: fuel quantity in one tank less than 3 US gal (± 1 US gal) 2 nd caution: fuel quantity in second tank less than 3 US gal (± 1 US gal)
VOLT	LOW VOLTS	voltage	On-board voltage below 24 V
PITOT	PITOT	Pitot heating	Pitot heating not switched ON, or fault in the Pitot heating system

Color and significance of the status light (white)

Status light (white)			Cause
Variant 'DAI'	Variant 'White Wire'	Meaning	
IGN	IGNITION	ignition	Electronic ignition control unit (if installed) not in operation

2.7 MASS (WEIGHT)*Subceeded*

Maximum take-off mass (Normal Category)	: 1150 kg	2535 lb
Maximum take-off mass (Utility Category)	: 980 kg	2161 lb
Maximum landing mass	: 1092 kg	2407 lb
Max. load in baggage compartment	: 30 kg	66 lb

WARNING

Exceeding the mass limits will lead to an overstressing of the airplane as well as to a degradation of flight characteristics and flight performance.

NOTE

The maximum landing mass is the highest mass for landing conditions at the maximum descent velocity. This velocity was used in the strength calculations to determine the landing gear loads during a particularly hard landing.

DA 40 AFM
DA 40 D AFM



Temporary Revision
Landing Mass 1150 kg

2.7 MASS (WEIGHT)

NOTE THIS IS NOW LESS THAN THE
MAX TAKEOFF WT OF 2646!

Maximum landing mass : 1150 kg (2535 lb) if landing gear struts with 18 mm
(0.71 in) thickness are installed (MÄM
40-123)

1092 kg 2407 lb otherwise

2.7 MASS (WEIGHT)

The first line is amended to read:

Maximum take-off mass (Normal Category) : ~~1150~~ kg ~~2535~~ lb

! if MAM 40-227 is carried out \Rightarrow 1200 kg 2646 lb

2.8 CENTER OF GRAVITY

Center of gravity limitations

The paragraph is amended to read:

The center of gravity (CG) for flight conditions must lie between the following limits:

Most forward CG:

~~2.40 m (94.5 in) aft of DP from 780 kg to 980 kg (1720 lb to 2161 lb)~~

~~2.46 m (96.9 in) aft of DP at 1150 kg (2535 lb)~~

linear variation between these values

if MAM 40-227 is carried out: *IT IS : THIS IS 100 LB WT INCREASE*

2.40 m (94.5 in) aft of DP from 780 kg to 980 kg (1720 lb to 2161 lb)

2.48 m (97.6 in) aft of DP at 1200 kg (2646 lb)

linear variation between these values

Most rearward CG:

a) Standard Tank : 2.59 m (102.0 in) aft of DP

~~★~~ b) Long Range Tank : 2.55 m (100.4 in) aft of DP

WARNING

Exceeding the center of gravity limitations reduces the controllability and stability of the airplane.

<p>1. Name of the project</p> <p>2. Date of the project</p>	<p>3. Name of the organization</p>	<p>4. Name of the person</p>
---	------------------------------------	------------------------------

5. Name of the project

6. Name of the organization

7. Name of the person

8. Name of the organization

9. Name of the person

10. Name of the organization

11. Name of the person

12. Name of the organization

13. Name of the person

14. Name of the organization

15. Name of the person

16. Name of the organization

17. Name of the person

18. Name of the organization

19. Name of the person

20. Name of the project

21. Name of the organization

22. Name of the person

<p>23. Name of the project</p>	<p>24. Name of the organization</p>	<p>25. Name of the person</p>
--------------------------------	-------------------------------------	-------------------------------

2.8 CENTER OF GRAVITY

Datum Plane

The Datum Plane (DP) is a plane which is normal to the airplane's longitudinal axis and in front of the airplane as seen from the direction of flight. The airplane's longitudinal axis is parallel with the upper surface of a 600:31 wedge which is placed on top of the rear fuselage in front of the vertical stabilizer. When the upper surface of the wedge is aligned horizontally, the Datum Plane is vertical. The Datum Plane is located 2.194 meters (86.38 in) forward of the most forward point of the root rib on the stub wing.

Center of gravity limitations

The center of gravity (CG) for flight conditions must lie between the following limits:

Most forward CG:

Suppressed (Previous Page)

2.40 m (94.5 in) aft of DP from 780 kg to 980 kg (1720 lb to 2161 lb)

2.46 m (96.9 in) aft of DP at 1150 kg (2535 lb)

linear variation between these values

Most rearward CG:

2.59 m (102.0 in) aft of DP

WARNING

Exceeding the center of gravity limitations reduces the controllability and stability of the airplane.

The NOTE is amended to read:

NOTE

In some countries the beginning of a flight is defined by starting the engine. In those countries a maximum ramp mass 4 kg (9 lb) above the maximum take-off mass is approved. At the time of lift-off the maximum permitted take-off mass must not be exceeded.

2.9 APPROVED MANEUVERS

The airplane is certified in the Normal Category and in the Utility Category in accordance with JAR-23.

Approved maneuvers

a) Normal Category:

- 1) all normal flight maneuvers;
- 2) stalling (with the exception of dynamic stalling); and
- 3) Lazy Eights, Chandelles, as well as steep turns and similar maneuvers, in which an angle of bank of not more than 60° is attained.

CAUTION

Aerobatics, spinning, and flight maneuvers with more than 60° of bank are not permitted in the Normal Category.

b) *Utility Category:*

- 1) all normal flight maneuvers;
- 2) stalling (with the exception of dynamic stalling); and
- 3) Lazy Eights, Chandelles, as well as steep turns and similar maneuvers, in which an angle of bank of not more than 90° is attained.

CAUTION

Aerobatics, spinning, and flight maneuvers with more than 90° of bank are not permitted in the Utility Category.

CAUTION

The accuracy of the attitude gyro (artificial horizon) and the directional gyro is affected by the maneuvers approved under item 3 if the bank angle exceeds 60°. Such maneuvers may therefore only be flown when the above mentioned instruments are not required for the present kind of operation.

2.10 MANEUVERING LOAD FACTORS

Table of maximum structural load factors:

Normal Category

	at v_A	at v_{NE}	with flaps in T/O or LDG position
Positive	3.8	3.8	2.0
Negative	-1.52	0	

Utility Category

	at v_A	at v_{NE}	with flaps in T/O or LDG position
Positive	4.4	4.4	2.0
Negative	-1.76	-1.0	

WARNING

Exceeding the maximum load factors will lead to an overstressing of the airplane.

2.11 OPERATING ALTITUDE

The maximum demonstrated operating altitude is 16,400 ft (5,000 meters).

The maximum approved operating altitude for US registered airplanes is 14,000 ft MSL unless an approved supplemental oxygen system is installed.

2.12 FLIGHT CREW

Minimum crew number : 1 (one person)

Maximum number of occupants:

Normal Category : 4 (four persons)

Utility Category : 2 (two persons), both of whom must sit in front

2.13 KINDS OF OPERATION

Provided that national operational requirements are met, the following kinds of operation are approved:

- * daytime flights according to Visual Flight Rules (VFR)
- * with the appropriate equipment: night flights according to Visual Flight Rules (VFR)
- * with the appropriate equipment: flights according to Instrument Flight Rules (IFR)

Flights into known or forecast icing conditions are prohibited.

Flights into known thunderstorms are prohibited.

Minimum operational equipment (serviceable) MEL (?)

The following table lists the minimum serviceable equipment required by JAR-23. Additional minimum equipment for the intended operation may be required by national operating rules and also depends on the route to be flown.

	for daytime VFR flights	in addition for night VFR flights	in addition for IFR flights
Flight and navigation instruments	<ul style="list-style-type: none"> * airspeed indicator * altimeter * magnetic compass 	<ul style="list-style-type: none"> * vertical speed indicator (VSI) * attitude gyro (artificial horizon) * turn & bank indicator * directional gyro * OAT indicator * chronometer with indication of hours, minutes, and seconds * VHF radio (COM) with speaker and microphone * VOR receiver * transponder (XPDR), mode A and mode C * 1 headset 	<ul style="list-style-type: none"> * second VHF radio (COM) * VOR-LOC-GP receiver * marker beacon receiver

	for daytime VFR flights	in addition for night VFR flights	in addition for IFR flights
engine instruments	<ul style="list-style-type: none"> * fuel indicators * integrated engine instrument * annunciator panel (all lights, see 2.6) 	<ul style="list-style-type: none"> * ammeter (included in VM 1000) * voltmeter (included in VM 1000) 	
lighting		<ul style="list-style-type: none"> * position lights * strobe lights (anti collision lights) * landing light * instrument lighting * flood light * flashlight 	
other operational minimum equipment	<ul style="list-style-type: none"> * stall warning system * fuel quantity measuring device (see 7.10) * safety belts for each occupied seat * airplane flight manual 	<ul style="list-style-type: none"> * Pitot heating system * alternate static valve * essential bus 	* emergency battery

NOTE

A list of approved equipment can be found in Chapter 6.

NOTE

For the upgrade of an airplane for Night VFR or IFR operation it is not sufficient to install the required equipment. The retrofit must be carried out in accordance with the requirements of the manufacturer (see Service Bulletin No. OSB 40-001) and the national airworthiness authority. Any additional equipment (equipment which is not listed in the Equipment List in Section 6.5) must also be approved for the intended kind of operation by the national airworthiness authority.

2.15 LIMITATION PLACARDS

On the instrument panel:

The following is added:

| If MÄM 40-227 is carried out, the following placard is installed:
|

Maneuvering Speed:

| $v_A = 111$ KIAS (above 1036 up to 1200 kg, above 2284 up to 2646 lb)

| $v_A = 94$ KIAS (780 to 1036 kg, 1720 to 2284 lb)

The airplane may only be operated in accordance with the Airplane Flight Manual. It can be operated in the „Normal“ and the „Utility“ categories in non-icing conditions. Provided that national operational requirements are met and the appropriate equipment is installed, this airplane is approved for the following kinds of operation: day VFR, night VFR and IFR. All aerobatic maneuvers including spinning are prohibited.

For further operational limitations refer to the Airplane Flight Manual.

No smoking.

2.14 FUEL

SUPERCEDED

Fuel grade	: AVGAS 100LL	
Fuel quantity	: Total fuel quantity	: 2 x 20,6 US gal (approx. 156 liters)
	Unusable fuel	: 2 x 0,5 US gal (approx. 3,8 liters)
	Max. indicated fuel quantity	: 15 US gal per tank
	Max. permissible difference between right and left tank	: 10 US gal (approx. 38 liters)

2.15 LIMITATION PLACARDS

All *limitation* placards are shown below. A list of *all* placards is included in the Airplane Maintenance Manual (Doc. No. 6.02.01), Chapter 11.

On the instrument panel:

SUPERCEDED

Maneuvering speed:

$v_A = 108$ KIAS (above 980 up to 1150 kg / above 2161 up to 2535 lb)

$v_A = 94$ KIAS (780 to 980 kg / 1720 to 2161 lb)

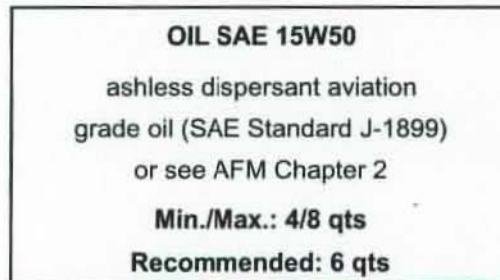
This airplane may only be operated in accordance with the Airplane Flight Manual. It can be operated in the "Normal" and "Utility" categories in non-icing conditions. Provided that national operational requirements are met and the appropriate equipment is installed, this airplane is approved for the following kinds of operation: day VFR, night VFR and IFR. All aerobatic maneuvers including spinning are prohibited. For further operational limitations refer to the Airplane Flight Manual.

No smoking.

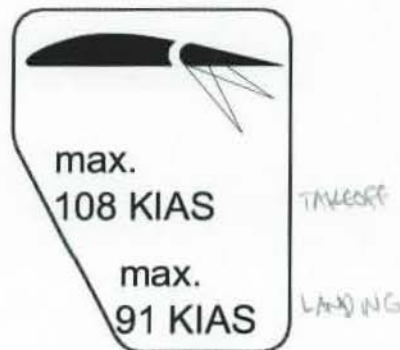
Next to each of the two fuel filler necks:



In the cowling, on the door for the oil filler neck:



Next to the flap selector switch:



Next to the essential bus switch (if installed)

Ess. Bus NOT for normal operation. See AFM.

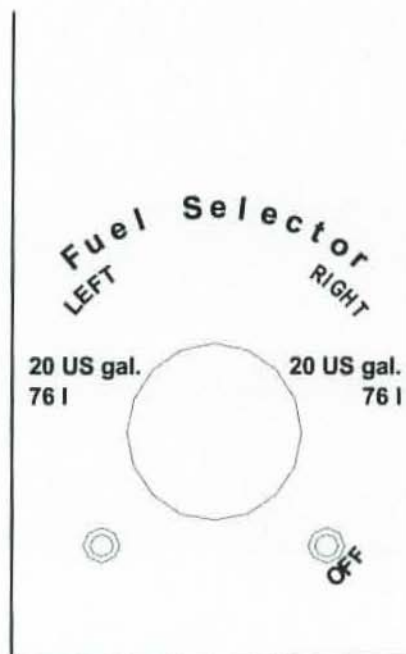
Next to the fuel quantity indication:

max. indicated fuel quantity: 15 US gal

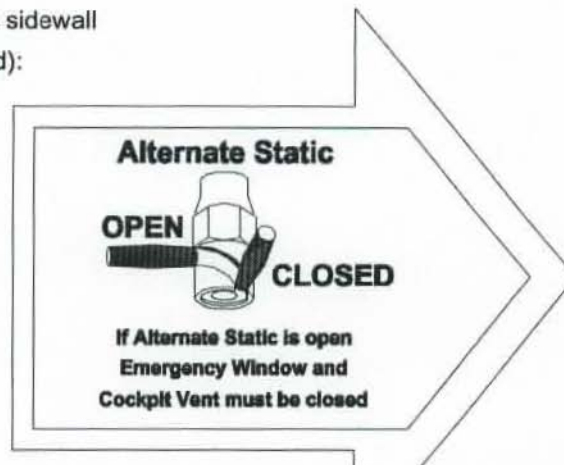
left and right tank max. 10 US gal difference

For use of max. tank capacity see AFM

On the fuel tank selector:



In the cockpit, on the left fuselage sidewall
 (if alternate static valve is installed):



Next to the baggage compartment:



▮ Beside the door locking device
 ▮ (if installed):
 ▮

EMERGENCY EXIT:

The keylock must be
 unlocked during flight

2.16 OTHER LIMITATIONS

2.16.1 TEMPERATURE

The airplane may only be operated when its temperature is not less than -20 °C (-4 °F).

2.16.2 BATTERY CHARGE

Taking off for a Night VFR or IFR flight with an empty battery is not permitted.

The use of an external power supply for engine starting with an empty airplane battery is not permitted if the subsequent flight is intended to be an IFR flight. In this case the airplane battery must first be charged.

2.16.3 EMERGENCY SWITCH

IFR flights are not permitted when the seal on the emergency switch is broken.

2.16.4 OPERATION TIME OF ELECTRICAL EQUIPMENT

Following an alternator failure and with the Essential Bus (if installed) switched ON, it can be expected that the systems listed under 3.7.2 FAILURES IN THE ELECTRICAL SYSTEM are supplied with power for half an hour. After this, electrical power is available for the attitude gyro (artificial horizon) and flood light for another 1.5 hours when the emergency power pack (if installed) is used.

2.16.5 DOOR LOCKING DEVICE

The canopy and the passenger door must not be blocked by the door locking device during operation of the airplane.

2.16.6 ELECTRONIC EQUIPMENT

The use and switching on of electronic equipment other than that which is part of the equipment of the airplane is not permitted, as it could lead to interference with the airplane's avionics.

Examples of undesirable items of equipment are:

- Mobile telephones
- Remote radio controls
- Video screens employing CRTs
- Minidisc recorders when in the record mode.

This list is not exhaustive.

The use of laptop computers, including those with CD-ROM drives, CD and minidisc players in the replay mode, cassette players and video cameras is permitted. All this equipment however should be switched off for take-off and landing.

CHAPTER 3

EMERGENCY PROCEDURES

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NOTE

Procedures for uncritical system faults are given in Chapter
4B ABNORMAL OPERATING PROCEDURES.

3.1 INTRODUCTION

3.1.1 GENERAL

This Chapter contains checklists as well as the description of recommended procedures to be followed in the event of an emergency. Engine failure or other airplane-related emergencies are most unlikely to occur if the prescribed procedures for pre-flight checks and airplane maintenance are followed.

If, nonetheless, an emergency does arise, the guidelines given here should be followed and applied in order to clear the problem.

As it is impossible to foresee all kinds of emergencies and cover them in this Airplane Flight Manual, a thorough understanding of the airplane by the pilot is, in addition to his knowledge and experience, an essential factor in the solution of any problems which may arise.

WARNING

In each emergency, control over the flight attitude and the preparation of a possible emergency landing have priority over attempts to solve the current problem ("first fly the aircraft"). Prior to the flight the pilot must consider the suitability of the terrain for an emergency landing for each phase of the flight. For a safe flight the pilot must constantly keep a safe minimum flight altitude. Solutions for various adverse scenarios should be thought over in advance. Thus it should be guaranteed that the pilot is at no time shocked by an engine failure and that he can act calmly and with determination.

3.1.2 CERTAIN AIRSPEEDS IN EMERGENCIES

The table is amended to read:

		Flight Mass	850 kg	1000 kg	1150 kg	1200 kg
		Event	1874 lb	2205 lb	2535 lb	2646 lb
Engine failure after take-off (Flaps T/O)			59 KIAS	66 KIAS	72 KIAS	74 KIAS
Airspeed for best glide angle (Flaps UP)			60 KIAS	68 KIAS	73 KIAS	76 KIAS
Emergency landing with engine off	Flaps UP		60 KIAS	68 KIAS	73 KIAS	76 KIAS
	Flaps T/O		59 KIAS	66 KIAS	72 KIAS	74 KIAS
	Flaps LDG		58 KIAS	63 KIAS	71 KIAS	73 KIAS

3.1.2 CERTAIN AIRSPEEDS IN EMERGENCIES

Event		850 kg 1874 lb	1000 kg 2205 lb	1150 kg 2535 lb
Engine failure after take-off (Flaps T/O)		59 KIAS	66 KIAS	72 KIAS
Airspeed for best glide angle (Flaps UP)		60 KIAS	68 KIAS	73 KIAS
Emergency landing with engine off	Flaps UP	60 KIAS	68 KIAS	73 KIAS
	Flaps T/O	59 KIAS	66 KIAS	72 KIAS
	Flaps LDG	58 KIAS	63 KIAS	71 KIAS

SURVIVED

3.2 ENGINE PROBLEMS

3.2.1 ENGINE PROBLEMS ON THE GROUND

1. Throttle IDLE
2. Brakes as required
3. Engine switch off, if considered necessary;
otherwise establish the cause of the
problem and re-establish engine
performance

CAUTION

If the oil pressure is below the green sector, the engine must be switched off immediately.

WARNING

If the problem cannot be cleared, the airplane must not be flown.

3.2.2 ENGINE PROBLEMS DURING TAKE-OFF

(a) Take-off can still be abandoned (sufficient runway length available)

land straight ahead:

1. Throttle IDLE

on the ground:

2. Brakes as required

CAUTION

If sufficient time is remaining, the risk of fire in the event of a collision can be reduced as follows:

- Fuel tank selector OFF
Mixture control lever LEAN-shut engine off
Ignition switch OFF
Master switch OFF

3.2.2 ENGINE PROBLEMS DURING TAKE-OFF

(b) Take-off can no longer be abandoned

Item 1 of the checklist is amended to read:

- | | |
|-------------------|----------------------------|
| 1. Airspeed | 74 KIAS (1200 kg, 2646 lb) |
| | 72 KIAS (1150 kg, 2535 lb) |
| | 66 KIAS (1000 kg, 2205 lb) |
| | 59 KIAS (850 kg, 1874 lb) |

(b) Take-off can no longer be abandoned

1. Airspeed ~~72 KIAS (1150 kg, 2535 lb)
66 KIAS (1000 kg, 2205 lb)
59 KIAS (850 kg, 1874 lb)~~
- SUPERSEDED*

WARNING

If, in the event of an engine problem occurring during take-off, the take-off can no longer be abandoned and a safe height has not been reached, then a straight-ahead emergency landing should be carried out. Turning back can be fatal.

if time allows:

2. Fuel tank selector check selected tank
3. Electrical fuel pump check ON
4. Ignition switch check BOTH
5. Throttle check MAX PWR
6. RPM lever check HIGH RPM
7. Mixture control lever check RICH (leaner above 5000 ft)
8. Alternate Air OPEN

WARNING

If the problem does not clear itself immediately, and the engine is no longer producing sufficient power, then an emergency landing must be carried out.

3.2.3 ENGINE PROBLEMS IN FLIGHT**(a) Engine running roughly*****Item 1 of the checklist is amended to read:***

- | | | |
|--|-------------------|----------------------------|
| | 1. Airspeed | 76 KIAS (1200 kg, 2646 lb) |
| | | 73 KIAS (1150 kg, 2535 lb) |
| | | 68 KIAS (1000 kg, 2205 lb) |
| | | 60 KIAS (850 kg, 1874 lb) |

3.2.3 ENGINE PROBLEMS IN FLIGHT**(a) Engine running roughly****WARNING**

An engine which is running very roughly can lead to the loss of the propeller. Only if there is no other alternative should an engine which is running roughly continue to be used.

1. Airspeed 73 KIAS (1150 kg, 2535 lb)
~~68 KIAS (1000 kg, 2205 lb)~~
~~60 KIAS (850 kg, 1874 lb)~~
2. Electrical fuel pump check ON
3. Fuel tank selector check selected tank
4. Engine instruments check
5. Throttle check
6. RPM lever check
7. Mixture control lever set for smooth running
8. Alternate Air OPEN
9. Ignition status light check (only if the electronic ignition control unit is installed)
10. Ignition switch check BOTH
11. Ignition circuit breaker (IGN) pull (only if the electronic ignition control unit is installed);
 if rough running is cleared by doing this, the circuit breaker should remain open
12. Throttle/RPM/Mixture try various settings

WARNING

If the problem does not clear itself immediately, and the engine is no longer producing sufficient power, then an emergency landing should be carried out.

(b) Loss of oil pressure

1. Check oil pressure warning light and oil pressure indicator.
2. Check oil temperature.
 - 2a. If the oil pressure indication drops below the green sector and the oil temperature is normal (oil pressure warning light does not illuminate or flash):
 - * Monitor the oil pressure warning light: it is probable that the oil pressure indication is defective.
 - * Monitor the oil and cylinder head temperatures.
 - 2b. If the oil pressure indication drops below the green sector while the oil or cylinder head temperature is rising, or

if the oil pressure warning light illuminates or flashes or

if both of these occur together:
 - * Reduce engine power to the minimum required.
 - * Land as soon as possible.
 - * Be prepared for engine failure and emergency landing.
 - 2c. Oil pressure tending to zero combined with:

Vibration, loss of oil, possibly unusual metallic noise and smoke:
 - * A mechanical failure in the engine is apparent.
 - * Shut off engine immediately and
 - * carry out emergency landing in accordance with 3.5.1 - EMERGENCY LANDING WITH ENGINE OFF.

(c) High oil pressure

Check oil temperature.

- * If the oil temperature is normal, it is probable that the fault lies in the oil pressure indication, which should thus be ignored (the airplane should be serviced).

(d) High oil temperature

Check cylinder head and exhaust gas temperature.

- * If neither of these is high, it is probable that the fault lies in the oil temperature indication. The airplane should be serviced. A stable oil temperature indication of 26 °F or 317 °F suggests a failure of the oil temperature sensor.
- * If the cylinder head temperature or exhaust gas temperature is also high:
 - Check oil pressure. If the oil pressure is low, proceed as in 3.2.3 (b) - Loss of oil pressure.
 - If the oil pressure is in the green sector:
 - Check mixture setting, enrich mixture if necessary.
 - Reduce power; if this produces no improvement, land at the nearest appropriate airfield.

(e) High cylinder head temperature

Cylinder head temperature in yellow sector or above:

1. Check mixture setting, enrich mixture if necessary.
2. Check oil temperature.
 - * If the oil temperature is also high:
 - Check oil pressure. If the oil pressure is low, proceed as in 3.2.3 (b) - Loss of oil pressure.
 - If the oil pressure is in the green sector:
 - Reduce power; if this produces no improvement, land at the nearest appropriate airfield.
 - Be prepared for possible emergency landing.

(f) High RPM

RPM moves on its own into the yellow sector, or is in the red sector:

1. Check friction adjuster for throttle quadrant.
2. Check oil pressure: Following a loss of oil or oil pressure, the propeller governor sets a high RPM. In this case the RPM should be regulated using the throttle. Proceed as in 3.2.3 (b) - Loss of oil pressure.

3. If oil pressure is normal:

- * Pull RPM lever back and listen for an associated drop in RPM:
 - If the indication does not change in spite of an audible drop in RPM, it is probable that the RPM indication is defective, which should thus be ignored (the airplane should be serviced).
 - If there is no audible drop in RPM, it is probable that the governor system is defective. In this case the RPM should be regulated using the throttle.

(g) Loss of RPM

1. Electrical fuel pump check ON
2. Fuel tank selector check
3. Friction adjuster for throttle quadrant check sufficiently tight
4. RPM lever HIGH RPM

- * Listen for rise in RPM.
 - If there is no audible rise in RPM, it is probable that the governor system is defective. In this case the RPM can be regulated within certain limits using the throttle.
 - Land at the nearest appropriate airfield.
 - Be prepared for possible emergency landing.
 - If the indication does not change in spite of an audible rise in RPM, it is probable that the RPM indication is defective, which should thus be ignored (the airplane should be serviced).

(h) High fuel flow

Fuel flow in the red sector:

1. Fuel pressure check:
 - * If the fuel pressure is low, there is possibly a leak (between the injection system and the injectors). Land at the nearest available airfield.
 - * If the fuel pressure is in the green sector there is no leak; the likely cause is a defective fuel flow indication, which should thus be ignored (the airplane should be serviced). Fuel flow data should be taken from the engine performance table in Chapter 5.
2. Check fuel quantity. A rapid reduction in fuel quantity confirms a high fuel flow.

(h) High fuel flow

Fuel flow in the red sector:

1. Fuel pump ON
2. Fuel pressure check after 10-15 sec:
 - * If the fuel pressure is low proceed with paragraph (i) 'Low fuel pressure with electrical pump ON.'
 - * If the fuel pressure is in the green sector or the fuel pressure warning light is not illuminated the likely cause is a defective fuel flow indication, which should thus be ignored (the airplane should be serviced). Fuel flow data should be taken from the engine performance table in Chapter 5.
3. Check fuel quantity. A rapid reduction in fuel quantity confirms a high fuel flow.

(i) Low fuel pressure with electrical pump ON

Fuel pressure warning light illuminates or fuel pressure indication below the green sector:

1. Fuel flow check:
 - * If the fuel flow is high, there is possibly a leak (between the injection system and the injectors). Land at the nearest available airfield.
 - * If the fuel flow is in the green sector and the engine is running smoothly, the likely cause is a defective fuel pressure indication, which should thus be ignored (the airplane should be serviced).

Monitor engine for power and rough operation that could indicate fuel starvation. If the engine is no longer producing sufficient power, then an emergency landing should be carried out.

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3.2.4 RESTARTING THE ENGINE WITH WINDMILLING PROPELLER

NOTE

Restarting the engine is possible at all airspeeds above 70 KIAS up to v_{NE} and up to the maximum demonstrated operating altitude.

NOTE

As long as an airspeed of at least 65 KIAS is maintained, and there is no major engine failure, the propeller will continue to windmill.

1. Airspeed 80 KIAS
2. Fuel tank selector fullest tank
3. Ignition switch check BOTH
4. Mixture control lever check appropriate position
5. Electrical fuel pump check ON
6. Alternate air OPEN

if engine does not start:

7. Mixture control lever LEAN
8. Mixture control lever push forward slowly
until engine starts

NOTE

If it is not possible to start the engine:

- adopt glide configuration as in 3.4 - GLIDING
- carry out emergency landing as in 3.5.1 - EMERGENCY
LANDING WITH ENGINE OFF

3.2.5 DEFECTIVE ENGINE CONTROLS

Defective Mixture Control Cable

(a) *Flight and Landing:*

1. Maintain altitude to the nearest airfield.
2. During descent, test the reaction of the engine to a higher power setting. A lean mixture can lead to engine roughness and a loss of power. The landing approach must be planned accordingly.

WARNING

Go-around may become impossible with the remaining power.

(b) *Engine shut-down:*

1. Parking brake set
2. Engine instruments check
3. Avionics master switch OFF
4. All electrical equipment OFF
5. Throttle IDLE
6. Ignition switch OFF
7. Master switch OFF

Defective Throttle Control Cable

(a) *Sufficient engine power available to continue flight:*

1. Approach nearest airfield, control engine power with RPM lever.
2. Perform landing with shut-down engine.

(b) *No sufficient engine power available to continue flight:*

1. Carry out emergency landing as in 3.5.1 - EMERGENCY LANDING WITH ENGINE OFF.

Defective RPM Lever Control Cable

(a) *Sufficient engine power available to continue flight:*

1. Approach nearest airfield, control engine power with throttle.
2. Perform normal landing

WARNING

Go-around may become impossible with the remaining power.

(b) *No sufficient engine power available to continue flight:*

1. Carry out emergency landing as in 3.5.1 - EMERGENCY LANDING WITH ENGINE OFF.

3.2.6 RESTARTING THE ENGINE WITH STATIONARY PROPELLER

NOTE

Restarting the engine is possible at all airspeeds above 80 KIAS up to v_{NE} and up to the maximum demonstrated operating altitude.

1. Airspeed 80 KIAS
2. Electrical equipment OFF
3. Avionics master switch OFF
4. Master switch (BAT) check ON
5. Mixture control lever check
6. Fuel tank selector check
7. Electrical fuel pump check ON
8. Alternate air OPEN
9. Ignition switch START

NOTE

By increasing the airspeed above approximately 130 KIAS, the propeller will begin to rotate and the engine can thus be started. For this, the ignition switch should be set at BOTH (see 3.2.4 RESTARTING THE ENGINE WITH WINDMILLING PROPELLER). An altitude loss of at least 1000 ft (300 meters) must be allowed for.

if it is not possible to start the engine:

- adopt glide configuration as in 3.4 - GLIDING
- carry out emergency landing as in 3.5.1 - EMERGENCY LANDING WITH ENGINE OFF

CAUTION

Engine restart following an engine fire should only be attempted if it is unlikely that a safe emergency landing can be made. It must be expected that engine restart is impossible after an engine fire.

3.3 SMOKE AND FIRE

3.3.1 SMOKE AND FIRE ON THE GROUND

(a) Engine fire when starting on the ground

1. Fuel tank selector OFF
2. Cabin heat OFF
3. Brakes apply

after standstill:

4. Throttle MAX PWR
5. Master switch (BAT) OFF

when the engine has stopped:

6. Ignition switch OFF
7. Canopy open
8. Airplane evacuate immediately

(b) Electrical fire with smoke on the ground

1. Master switch (BAT) OFF

if the engine is running:

2. Throttle IDLE
3. Mixture control lever LEAN - shut off engine

when the engine has stopped:

4. Ignition switch OFF
5. Canopy open
6. Airplane evacuate immediately

3.3.2 SMOKE AND FIRE DURING TAKE-OFF

(b) If take-off cannot be abandoned

Item 3 of the checklist is amended to read:

3. Airspeed 74 KIAS (1200 kg, 2646 lb)
72 KIAS (1150 kg, 2535 lb)
66 KIAS (1000 kg, 2205 lb)
59 KIAS (850 kg, 1874 lb)

3.3.2 SMOKE AND FIRE DURING TAKE-OFF

(a) If take-off can still be abandoned

1. Throttle IDLE
2. Cabin heat OFF
3. Brakes apply - bring the airplane to a stop
4. After stopping proceed as in 3.3.1 - SMOKE AND
FIRE ON THE GROUND

(b) If take-off cannot be abandoned

1. Cabin heat OFF
2. If possible, fly along a short-cut traffic circuit and land on the airfield.

WARNING

If, in the event of an engine problem occurring during take-off, the take-off can no longer be abandoned and a safe height has not been reached, then a straight-ahead emergency landing should be carried out. Turning back can be fatal.

3. Airspeed 73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

after climbing to a height from which the selected landing area can be reached safely:

4. Fuel tank selector OFF
5. Electrical fuel pump OFF
6. Cabin heat OFF
7. Master switch (BAT) OFF
8. Emergency window(s) open if required
9. Carry out emergency landing with engine off. Allow for increased landing distance due to the flap position.

CAUTION

In case of extreme smoke development, the front canopy may be unlatched during flight. This allows it to partially open, in order to improve ventilation. The canopy will remain open in this position. Flight characteristics will not be affected significantly.

3.3.3 SMOKE AND FIRE IN FLIGHT**(a) Engine fire in flight**

1. Cabin heat OFF
2. Select appropriate emergency landing field.

when it seems certain that the landing field will be reached:

3. Fuel tank selector OFF
4. Throttle MAX PWR
5. Electrical fuel pump OFF
6. Master switch (BAT) ON
7. Emergency window(s) open if required
8. Carry out emergency landing with engine off.

CAUTION

In case of extreme smoke development, the front canopy may be unlatched during flight. This allows it to partially open, in order to improve ventilation. The canopy will remain open in this position. Flight characteristics will not be affected significantly.

(b) Electrical fire with smoke in flight

1. Emergency switch ON if installed
2. Master switch (BAT) OFF
3. Cabin heat OFF
4. Emergency window(s) open if required
5. Land at an appropriate airfield as soon as possible

CAUTION

Switching OFF the master switch (BAT) will lead to total failure of all electronic and electric equipment. Also affected from this are - if installed - the attitude gyro (artificial horizon) and the directional gyro.

However, by switching the emergency switch ON (only installed in the IFR model), the emergency battery will supply power to the attitude gyro (artificial horizon) and the flood light.

In case of extreme smoke development, the front canopy may be unlatched during flight. This allows it to partially open, in order to improve ventilation. The canopy will remain open in this position. Flight characteristics will not be affected significantly.

3.4 GLIDING

The checklist is amended to read:

1. Flaps UP
2. Airspeed 76 KIAS (1200 kg, 2646 lb)
73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

3.4 GLIDING

1. Flaps UP
2. Airspeed 73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

NOTE

The glide ratio is 8.8; i.e., for every 1000 ft (305 meters) of altitude loss the maximum horizontal distance traveled in still air is 1.45 NM (2.68 km). During this the propeller will continue to windmill.

With a stationary propeller the glide ratio is 10.3; this corresponds to a maximum horizontal distance of 1.70 NM (3.14 km) for every 1000 ft altitude. In consideration of a safe airspeed however, this configuration may not be attainable.

3.5 EMERGENCY LANDING

3.5.1 EMERGENCY LANDING WITH ENGINE OFF

Item 4 of the checklist is amended to read:

4. Airspeed 76 KIAS (1200 kg, 2646 lb)
73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

3.5 EMERGENCY LANDINGS

3.5.1 EMERGENCY LANDING WITH ENGINE OFF

1. Select suitable landing area. If no level landing area is available, a landing on an upward slope should be sought.
2. Consider wind.
3. Approach: If possible, fly along a short-cut rectangular circuit. On the downwind leg of the circuit the landing area should be inspected for obstacles from a suitable height. The degree of offset at each part of the circuit will allow the wind speed and direction to be assessed.
4. Airspeed 73 KIAS (1150 kg, 2535 lb.)
68 KIAS (1000 kg, 2205 lb.)
60 KIAS (850 kg, 1874 lb.)
5. If time allows advise ATC
6. Fuel tank selector OFF

when it is certain that the landing field will be reached:

7. Flaps LDG
8. Safety harnesses tighten

CAUTION

If sufficient time is remaining, the risk of fire in the event of a collision with obstacles can be reduced as follows:

Ignition switch OFF

Master switch OFF

9. Touchdown with the lowest possible airspeed

3.5.2 LANDING WITH A DEFECTIVE TIRE ON THE MAIN LANDING GEAR**CAUTION**

A defective (e.g. burst) tire is not usually easy to detect. The damage normally occurs during take-off or landing, and is hardly noticeable during fast taxiing. It is only during the roll-out after landing or at lower taxiing speeds that a tendency to swerve occurs. Rapid and determined action is then required.

1. Advise ATC.
2. Land the airplane at the edge of the runway that is located on the side of the intact tire, so that changes in direction which must be expected during roll-out due to the braking action of the defective tire can be corrected on the runway.
3. Land with one wing low. The wing on the side of the intact tire should be held low.
4. Direction should be maintained using the rudder. This should be supported by use of the brake. It is possible that the brake must be applied strongly - if necessary to the point where the wheel locks. The wide track of the landing gear will prevent the airplane from tipping over a wide speed range. There is no pronounced tendency to tip even when skidding.

3.5.3 LANDING WITH DEFECTIVE BRAKES

In general, a landing on grass is recommended in order to reduce the landing run by virtue of the greater rolling resistance.

CAUTION

If sufficient time is remaining, the risk of fire in the event of a collision can be reduced as follows:

Fuel tank selector OFF
Mixture control lever LEAN - shut off engine
Ignition switch OFF
Master switch OFF

3.6 RECOVERY FROM AN UNINTENTIONAL SPIN

CAUTION

Steps 1 to 4 must be carried out **immediately** and **simultaneously**.

1. Throttle IDLE
2. Rudder full deflection against
direction of spin
3. Elevator (control stick) fully forward
4. Ailerons neutral
5. Flaps UP

when rotation has stopped:

6. Rudder neutral
7. Elevator (control stick) pull carefully
8. Return the airplane from a descending into a normal flight attitude. In so doing do not exceed the 'never exceed speed', V_{NE} .

3.7 OTHER EMERGENCIES

3.7.1 ICING

Unintentional flight into icing conditions

1. Leave the icing area (by changing altitude or turning back, in order to reach zones with a higher ambient temperature).
2. Pitot heating ON
3. Cabin heat ON
4. Air distributor lever ▲ (up)
5. RPM increase, in order to prevent ice
build-up on the propeller blades
6. Alternate Air OPEN
7. Emergency window(s) open if required

CAUTION

Ice build-up increases the stalling speed. If required for safety reasons, engine speeds up to 2700 RPM are admissible without time limit.

8. ATC advise if an emergency is expected

CAUTION

When the Pitot heating fails, and the alternate static valve is installed:

- Alternate static valve OPEN
 Emergency window(s) close

3.7.2 FAILURES IN THE ELECTRICAL SYSTEM

(a) Complete failure of the electrical system

Due to the strong mechanical design as well as due to the required check of the system during scheduled inspections, a total failure of the electrical system is extremely unlikely. If, nevertheless, a total failure should occur, all circuit breakers should be checked, pulled and re-set. If this does not help:

- Set emergency switch to ON (if installed).
- When necessary, use the flood light for lighting the instruments as well as levers and switches, etc.
- Set power based on lever positions and engine noise.
- Prepare landing with flaps in the given position.
- Land on the nearest appropriate airfield.

(b) Alternator failure

An alternator failure is indicated by an illuminated or flashing alternator warning light (ALT or ALTERNATOR) on the annunciator panel and a flashing ammeter on the Vision Microsystems VM 1000 engine instrument.

1. Circuit breakers check; if all are O.K., proceed
with step 2
2. Electrical equipment switch OFF all equipment which is
not needed
3. Voltmeter check regularly

CAUTION

Those items of equipment which are not needed for the safe operation and secure landing of the airplane can be switched off with the Essential Bus Switch (if installed). When the Essential Bus is switched ON, only the following items of equipment are supplied with power:

- NAV/COM 1
- transponder (XPDR)
- flood light
- attitude gyro (artificial horizon)
- VM 1000 engine instrument
- annunciator panel
- GPS (if installed)
- landing light
- Pitot heating system
- flaps

These items of equipment can be supplied with power by the battery for at least 30 minutes. Economical use, in particular of the Pitot heating, and switching off equipment that is not needed extends the time during which the other equipment remains available. During the 30 minutes period, the airplane must be landed at a suitable airfield.

For cases in which the battery capacity is not sufficient to reach a suitable airfield, an emergency battery is installed in the IFR model, serving as an additional back-up system for the attitude gyro (artificial horizon) and flood light. This battery is switched on with the Emergency Switch. It lasts for 1 hour and 30 minutes when the flood light is switched on,

(c) Starter malfunction

If the starter does not disengage from the engine after starting (starter warning light (START) on the annunciator panel remains illuminated or flashing after the engine has started):

1. Throttle IDLE
2. Mixture control lever LEAN - shut off engine
3. Ignition switch OFF
4. Master switch OFF

Terminate flight preparation!

(d) Overvoltage

If a voltage in the upper red sector (above 32 volts) is indicated:

1. Essential bus ON, if installed
2. Master switch (ALT) OFF

WARNING

Leave master switch (BAT) ON!

3. Equipment that is not needed,
in particular Pitot heat OFF
4. Land on the nearest appropriate airfield.

3.7.3 SUSPICION OF CARBON MONOXIDE CONTAMINATION IN THE CABIN

Carbon monoxide (CO) is a gas which is developed during the combustion process. It is poisonous and without smell. Since it occurs however usually together with flue gases, it can be detected. Increased concentration of carbon monoxide in closed spaces can be fatal. The occurrence of CO in the cabin is possible only due to a defect. If a smell similar to exhaust gases is noticed in the cabin, the following measures should be taken:

1. Cabin heat OFF
2. Ventilation open
3. Emergency window(s) open
4. Forward canopy open

CAUTION

In case of suspicion of carbon monoxide contamination in the cabin, the front canopy may be unlatched during flight. This allows it to partially open, in order to improve ventilation. The canopy will remain open in this position. Flight characteristics will not be affected significantly.

CHAPTER 4A

NORMAL OPERATING PROCEDURES

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4A.2 AIRSPEEDS FOR NORMAL OPERATING PROCEDURES

The table is amended to read:

Event \ Flight mass	850 kg 1874 lb	1000 kg 2205 lb	1150 kg 2535 lb	1200 kg 2646 lb
Airspeed for take-off climb (best rate-of-climb speed v_y) (Flaps T/O)	54 KIAS	60 KIAS	66 KIAS	67 KIAS
Airspeed for cruise climb (Flaps UP)	60 KIAS	68 KIAS	73 KIAS	76 KIAS
Approach speed for normal landing (Flaps LDG)	58 KIAS	63 KIAS	71 KIAS	73 KIAS
Minimum speed during touch & go (Flaps T/O)	54 KIAS	60 KIAS	66 KIAS	67 KIAS

4A.1 INTRODUCTION

Chapter 4A contains checklists and describes extended procedures for the normal operation of the airplane.

Superseded

4A.2 AIRSPEEDS FOR NORMAL OPERATING PROCEDURES

Flight mass	850 kg 1874 lb	1000 kg 2205 lb	1150 kg 2535 lb
Airspeed for take-off climb (best rate-of-climb speed v_y) (Flaps T/O)	54 KIAS	60 KIAS	66 KIAS
Airspeed for cruise climb (Flaps UP)	60 KIAS	68 KIAS	73 KIAS
Approach speed for normal landing (Flaps LDG)	58 KIAS	63 KIAS	71 KIAS
Minimum speed during touch & go (Flaps T/O)	54 KIAS	60 KIAS	66 KIAS

4A.3 CHECKLISTS FOR NORMAL OPERATING PROCEDURES

4.A3.1 PRE-FLIGHT INSPECTION

I. Cabin check

- a) MET, NAV, Mass & CG flight planning completed
- b) Airplane documents complete and up-to-date
- c) Ignition key pulled out
- d) Front canopy & rear door clean, undamaged,
check locking mechanism function
- e) All electrical equipment OFF
- f) Circuit breakers set in (if one has been pulled, check
reason)
- g) Engine control levers check condition, freedom of
movement and full travel of
throttle, RPM and mixture levers
- h) Throttle IDLE
- i) Mixture control lever LEAN
- j) RPM lever HIGH RPM
- k) Master switch (BAT) ON
- l) Annunciator panel check function (see 7.11)
- m) Fuel quantity check with fuel qty.
measuring device

NOTE

When the fuel quantity indicator reads 15 US gal, the correct fuel quantity must be determined with the fuel quantity measuring device. If this measurement is not carried out, the fuel quantity available for flight planning is 15 US gal.

- n) Position lights, strobe lights (ACL's) check
- o) Master switch (BAT) OFF
- p) Check for loose items complete
- q) Flight controls and trim free to move and correct
- r) Baggage stowed on seats and secure

II. Walk-around check, visual inspection

CAUTION

A visual inspection means: examination for damage, cracks, delamination, excessive play, load transmission, correct attachment and general condition. In addition control surfaces should be checked for freedom of movement.

CAUTION

In low ambient temperatures the airplane should be completely cleared of ice, snow and similar accumulations.

CAUTION

Prior to flight, remove such items as control surfaces gust lock, Pitot cover, tow bar, etc.

1. Left main landing gear:

- a) Landing gear strut visual inspection
- b) Wheel fairing visual inspection
- c) Tire inflation pressure (2.5 bar/36 psi) check
- d) Wear, tread depth of tire check
- e) Tire, wheel, brake visual inspection
- f) Brake line connection check for leaks
- g) Slip marks visual inspection
- h) Chocks remove

2. Left Wing:

- a) Entire wing surface visual inspection
- b) Step visual inspection
- c) Air intake on lower surface visual inspection
- d) Openings on lower surface check for traces of fuel
(if tank is full, fuel may spill over through the tank vent)
- e) Tank drain drain off a small quantity,
check for water and sediment
- f) Stall warning check (suck on opening)
- g) Tank filler visual inspection, fuel quantity
must agree with indicator
- h) 2 stall strips on wing visual inspection
- i) Pitot probe clean, orifices open
- j) Landing/taxi light visual inspection

4A.3 CHECKLISTS FOR NORMAL OPERATING PROCEDURES

4A.3.1 PRE-FLIGHT INSPECTION

II. Walk-around check, visual inspection

8. Front Fuselage:

h) Winter Baffle (if installed) visual inspection

- k) Wing tip visual inspection
- l) Position light, strobe light (ACL) visual inspection
- m) Mooring check, clear
- n) Aileron and linkage visual inspection
- o) Aileron hinges and safety pin visual inspection
- p) Foreign objects in aileron paddle visual inspection
- q) Flap and linkage visual inspection
- r) Flap hinges and safety pin visual inspection

3. Fuselage, left side:

- a) Canopy, left side visual inspection
- b) Rear cabin door & window visual inspection
- c) Fuselage skin visual inspection
- d) Antennas visual inspection

4. Empennage:

- a) Stabilizers and control surfaces visual inspection
- b) Hinges visual inspection
- c) Elevator trim tab visual inspection, check safetying
- d) Rudder trim tab visual inspection
- e) Mooring on fin check, clear
- f) Tail skid and lower fin visual inspection
- g) Towing assembly, if fitted visual inspection

5. *Fuselage, right side:*

- a) Fuselage skin visual inspection
- b) Window visual inspection
- c) Canopy, right side visual inspection

6. *Right wing:*

- a) Flap and linkage visual inspection
- b) Flap hinges and safety pin visual inspection
- c) Aileron and linkage visual inspection
- d) Aileron hinges and safety pin visual inspection
- e) Foreign objects in aileron paddle visual inspection
- f) Wing tip visual inspection
- g) Position light, strobe light (ACL) visual inspection
- h) Mooring check, clear
- i) Entire wing surface visual inspection
- j) 2 stall strips on wing visual inspection
- k) Tank filler visual check, fuel quantity
must agree with indicator
- l) Openings on lower surface check for traces of fuel
(if tank is full, fuel may spill over
through the tank vent)
- m) Tank drain drain off a small quantity,
check for water and sediment
- n) Step visual inspection

7. Right Main Landing Gear:

- a) Landing gear strut visual inspection
- b) Wheel fairing visual inspection
- c) Tire inflation pressure (2.5 bar/36 psi) check
- d) Wear, tread depth of tires check
- e) Tire, wheel, brake visual inspection
- f) Brake line connection check for leaks
- g) Slip marks visual inspection
- h) Chocks remove

8. Front fuselage:

- a) Oil level check dipstick,
min. 4 qts for VFR operation
min. 6 qts for IFR operation
- b) Cowling visual inspection
- c) 3 air intakes clear
- d) Propeller visual inspection;
bladeshake: max. 3 mm (1/8 in);
angular play of blade: max. 2°

WARNING

Never move the propeller by hand while the ignition is switched on, as it may result in serious personal injury.

- e) Spinner including attachment screws visual inspection

- f) Nose landing gear visual inspection
- g) Tire and wheel visual inspection,
check slip marks
- h) Wear, tread depth of tire check
- i) Wheel fairing visual inspection
- j) Tow bar removed
- k) Tire inflation pressure (2.0 bar/29 psi) check
- l) Chocks remove
- m) Exhaust visual inspection

WARNING

The exhaust can cause burns when it is hot.

Underside:

- n) Antennas (if fitted) visual inspection
- o) Gascolator drain off a small quantity of fuel,
check for water and sediment
- p) Venting pipes check for blockage
- q) Fuselage underside check for excessive contamination
particularly by oil, fuel, and other
fluids

4A.3.2 BEFORE STARTING ENGINE

1. Pre-flight inspection complete
2. Rudder pedals adjusted and locked
3. Passengers instructed
4. Safety harnesses all on and fastened
5. Rear door closed and locked
6. Door lock (if installed) unblocked, key removed
7. Front canopy Position 1 or 2 ("cooling gap")
8. Canopy lock (if installed) unblocked, key removed
9. Parking brake set
10. Flight controls free movement
11. Trim wheel T/O
12. Throttle IDLE
13. RPM lever HIGH RPM
14. Mixture control lever LEAN
15. Friction device, throttle quadrant adjusted
16. Alternate Air CLOSED
17. Alternate Static Valve CLOSED, if installed
18. Avionics master switch OFF
19. Essential Bus switch OFF, if installed

CAUTION

When the essential bus is switched ON, the battery will not be charged.

20. Master switch (BAT) ON
21. Annunciator panel test (see Section 7.11)
22. Fuel tank selector on full tank

WARNING

Never move the propeller by hand while the ignition is switched on, as it may result in serious personal injury.

Never try to start the engine by hand.

4A.3.3 STARTING ENGINE

(a) Cold engine

1. Strobe light (ACL) ON
2. Electrical fuel pump ON, note pump noise
(= functional check of pump)
3. Throttle 3 cm (1.2 in) forward from IDLE
(measured from rear of slot)
4. Mixture control lever RICH for 3 - 5 sec, then LEAN
5. Throttle 1 cm (0.4 in) forward from IDLE
(measured from rear of slot)

WARNING

Before starting the engine the pilot must ensure that the propeller area is free, and no persons can be endangered.

CAUTION

Do not overheat the starter motor. Do not operate the starter motor for more than 10 seconds. After operating the starter motor, let it cool off for 20 seconds. After 6 attempts to start the engine, let the starter cool off for half an hour.

CAUTION

The use of an external pre-heater and external power source is recommended whenever possible, in particular at ambient temperatures below 0 °C (32 °F), to reduce wear and abuse to the engine and electrical system. Pre-heat will thaw the oil trapped in the oil cooler, which can be congealed in extremely cold temperatures. After a warm-up period of approximately 2 to 5 Minutes (depending on the ambient temperature) at 1500 RPM, the engine is ready for take-off if it accelerates smoothly and the oil pressure is normal and steady.

6. Ignition switch START

when engine fires:

7. Mixture control lever rapidly move to RICH
 8. Oil pressure green sector within 15 sec
 9. Electrical fuel pump OFF

WARNING

If the oil pressure has not moved into the green sector within 15 seconds after starting, SWITCH OFF ENGINE and investigate problem.

10. Master switch (ALT) ON
 11. Ammeter check
 12. Fuel pressure check (14 psi to 35 psi)
 13. Annunciator panel check

(b) Warm engine

1. Strobe light (ACL) ON
2. Electrical fuel pump ON, note pump noise
(= functional check of pump)
3. Throttle 3 cm (1.2 in) forward from IDLE
(measured from rear of slot)
4. Mixture control lever RICH for 1 - 3 sec, then LEAN

WARNING

Before starting the engine the pilot must ensure that the propeller area is free and no persons can be endangered.

CAUTION

Do not overheat the starter motor. Do not operate the starter motor for more than 10 seconds. After operating the starter motor, let it cool off for 20 seconds. After 6 attempts to start the engine, let the starter cool off for half an hour.

5. Ignition switch START

when engine fires:

6. Mixture control lever rapidly move to RICH
7. Oil pressure green sector within 15 sec

WARNING

If the oil pressure has not moved into the green sector within 15 seconds after starting, SWITCH OFF ENGINE and investigate problem.

8. Electrical fuel pump OFF
9. Master switch (ALT) ON
10. Ammeter check
11. Fuel pressure check (14 psi to 35 psi)
12. Annunciator panel check

(c) Engine will not start after injection ("flooded engine")

1. Strobe light (ACL) ON
2. Electrical fuel pump OFF
3. Mixture control lever LEAN, fully aft
4. Throttle at mid position

WARNING

Before starting the engine the pilot must ensure that the propeller area is free and no persons can be endangered.

CAUTION

Do not overheat the starter motor. Do not operate the starter motor for more than 10 seconds. After operating the starter motor, let it cool off for 20 seconds. After 6 attempts to start the engine, let the starter cool off for half an hour.

5. Ignition switch START
6. Throttle pull back towards IDLE when engine fires

when engine fires:

7. Mixture control lever rapidly move to RICH
8. Oil pressure green sector within 15 sec

WARNING

If the oil pressure has not moved into the green sector within 15 seconds after starting, SWITCH OFF ENGINE and investigate problem.

9. Master switch (ALT) ON
10. Ammeter check
11. Fuel pressure check (14 psi to 35 psi)
12. Annunciator panel check

4A.3.4 BEFORE TAXIING

1. Avionics master switch ON
2. Electrical equipment ON as required
3. Flaps UP - T/O - LDG - T/O
(indicator and visual check)
4. Flight instruments and avionics set, test function, as required
5. Flood light ON, test function, as required
6. Ammeter check, if required increase RPM
7. Fuel tank selector change tanks, confirm that engine
also runs on other tank (at least 1
minute at 1500 RPM)
8. Pitot heating ON, test function;
ammeter must show rise
9. Pitot heating OFF if not required
10. Strobe lights (ACL's) check ON, test function,
as required
11. Position lights, landing and taxi lights ON, test function, as required

CAUTION

When taxiing at close range to other aircraft, or during night flight in clouds, fog or haze, the strobe lights should be switched OFF. The position lights must always be switched ON during night flight.

12. Idle RPM check, 600 to 800 RPM

4A.3.5 TAXIING

1. Parking brake release
2. Brakes test on moving off
3. Flight instrumentation and avionics
(particularly directional gyro and
turn and bank indicator) check for correct indications

CAUTION

When taxiing on a poor surface select the lowest possible RPM to avoid damage to the propeller from stones or similar items.

CAUTION

Following extended operation on the ground, or at high ambient temperatures, the following indications of fuel vapor lock may appear:

- arbitrary changes in idle RPM and fuel flow
- slow reaction of the engine to operation of throttle
- engine will not run with throttle in IDLE position

Remedy:

1. For about 1 to 2 minutes, or until the engine settles, run at a speed of 1800 to 2000 RPM. Oil and cylinder head temperatures must stay within limits.
2. Pull throttle back to IDLE to confirm smooth running.
3. Set throttle to 1200 RPM and mixture for taxiing, i.e., use mixture control lever to set the maximum RPM attainable.
4. Immediately before the take-off run set the mixture for take-off, apply full throttle and hold this position for 10 seconds.

NOTE

Vapor lock can be avoided if the engine is run at speeds of 1800 RPM or more. This results in lower fuel temperatures.

4A.3.6 BEFORE TAKE-OFF**CAUTION**

Before take-off, the engine must run on each tank for at least 1 minute at 1500 RPM. This can be done during taxiing and during the start check.

1. Position airplane into wind if possible
2. Parking brake set
3. Safety harnesses on and fastened
4. Rear door check closed and locked
5. Front canopy closed and locked
6. Door warning light (DOOR or DOORS) check
7. Fuel tank selector fullest tank
8. Engine instruments in green sector
9. Circuit breakers pressed in
10. Fuel pressure indicator check (approx. 15 - 25 psi)
11. Electrical fuel pump ON
12. Mixture control lever RICH (below 5000 ft)

NOTE

At a density altitude of 5000 ft or above or at high ambient temperatures a fully rich mixture can cause rough running of the engine or a loss of performance. The mixture should be set for smooth running of the engine.

13. Flaps check T/O
14. Trim check T/O
15. Flight controls free movement, correct sense
16. Throttle 2000 RPM
17. RPM lever pull back until a drop of max.
500 RPM is reached - HIGH RPM;
cycle 3 times
18. Magneto check L - BOTH - R - BOTH
Max. RPM drop 175 RPM
Max. difference 50 RPM
If the electronic ignition control unit
is installed, the ignition status light
must illuminate and extinguish after
approximately 20 to 30 sec

CAUTION

The lack of an RPM drop suggests a faulty grounding or incorrect ignition timing. In case of doubt the magneto check can be repeated with a leaner mixture, in order to confirm a problem. Even when running on only one magneto the engine should not run unduly roughly.

19. Throttle IDLE
20. Parking brake release
21. Alternate Air check CLOSED
22. Landing light ON as required

4A.3.7 TAKE-OFFNormal take-off procedure

1. Transponder ON/ALT
2. RPM lever check HIGH RPM
3. Throttle MAX PWR (not abruptly)

WARNING

The proper performance of the engine at full throttle should be checked early in the take-off procedure, so that the take-off can be abandoned if necessary.

A rough engine, sluggish RPM increase, or failure to reach take-off RPM (2680 ± 20 RPM) are reasons for abandoning the take-off. If the engine oil is cold, an oil pressure in the yellow sector is permissible.

4. Elevator neutral
5. Rudder maintain direction

NOTE

In strong crosswinds steering can be augmented by use of the toe brakes. It should be noted, however, that this method increases the take-off roll, and should not generally be used.

4A.3.7 TAKE-OFF

Item 7 of the checklist is amended to read:

- I 7. Airspeed 67 KIAS (1200 kg, 2646 lb)
66 KIAS (1150 kg, 2535 lb)
60 KIAS (below 1000 kg, 2205 lb)

6. Nose wheel lift-off at $v_R = 59$ KIAS
7. Airspeed 66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)

above a safe height:

8. RPM lever 2400 RPM
9. Electrical fuel pump OFF
10. Landing light OFF

4A.3.8 CLIMBProcedure for best rate of climb

Item 2 of the checklist is amended to read:

2. Airspeed 67 KIAS (1200 kg, 2646 lb)
66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)

Cruise climb

Item 2 of the checklist is amended to read:

2. Airspeed 76 KIAS (1200 kg, 2646 lb)
73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

4A.3.8 CLIMBProcedure for best rate of climb

1. Flaps T/O
2. Airspeed 66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)
3. RPM lever 2400 RPM
4. Throttle MAX PWR
5. Mixture control lever RICH,
above 5000 ft hold EGT constant
6. Engine instruments in green sector
7. Trim as required

CAUTION

When the fuel pressure warning light illuminates, or the fuel pressure indication is below the green sector, the electrical fuel pump must be switched ON.

Cruise climb

1. Flaps UP
2. Airspeed 73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)
3. RPM lever 2400 RPM
4. Throttle MAX PWR
5. Mixture control lever RICH,
above 5000 ft hold EGT constant
6. Engine instruments in green sector
7. Trim as required

4A.3.8 CLIMBProcedure for best rate of climb

Item 8 is added behind item 7

8. Electrical fuel pump ON at high altitudes

The CAUTION is replaced by the following CAUTION:

CAUTION

Operation at high altitudes with the electrical fuel pump OFF may cause vapor bubbles, resulting in intermittent low fuel pressure indications , sometimes followed by high fuel flow indications.

Cruise climb

Item 8 is added behind item 7

8. Electrical fuel pump ON at high altitudes

The following CAUTION is added:

CAUTION

Operation at high altitudes with the electrical fuel pump OFF may cause vapor bubbles, resulting in intermittent low fuel pressure indications , sometimes followed by high fuel flow indications.

4A.3.9 CRUISE

1. Flaps UP
2. Throttle set performance according to table
3. RPM lever 1800 - 2400 RPM

NOTE

Favorable combinations of manifold pressure and RPM are given in Chapter 5.

NOTE

To optimize engine life the cylinder head temperature (CHT) should lie between 150 °F and 400 °F in continuous operation, and not rise above 435 °F in fast cruise.

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NOTE

The oil temperature in continuous operation should lie between 165 °F and 220 °F. If possible, the oil temperature should not remain under 180 °F for long periods, so as to avoid accumulation of condensation water.

4. Mixture set in accordance with
4A3.10 MIXTURE ADJUSTMENT

5. Trim as required
6. Fuel tank selector as required
(max. difference 10 US gal)

NOTE

While switching from one tank to the other, the electrical fuel pump should be switched ON.

CAUTION

When the fuel pressure warning light illuminates, or the fuel pressure indication is below the green sector, the electrical fuel pump must be switched ON.

4A.3.10 MIXTURE ADJUSTMENT**CAUTION**

1. The maximum permissible cylinder head temperature (500 °F) must never be exceeded.
2. The mixture control lever should always be moved slowly.
3. Before selecting a higher power setting the mixture control lever should, on each occasion, be moved slowly to fully RICH.
4. Care should always be taken that the cylinders do not cool down too quickly. The cooling rate should not exceed 50 °F per minute.

Item 7 is added behind item 6

7. Electrical fuel pump ON at high altitudes

The first CAUTION on this page is replaced by the following CAUTION:

CAUTION

Operation at high altitudes with the electrical fuel pump OFF may cause vapor bubbles, resulting in intermittent low fuel pressure indications , sometimes followed by high fuel flow indications.

Best Economy Mixture

The best economy mixture setting may only be used up to a power setting of 75 %. In order to obtain the lowest specific fuel consumption at a particular power setting proceed as follows: Slowly pull the mixture control lever back towards LEAN until the engine starts to run roughly. Then push the mixture control lever forward just far enough to restore smooth running. At the same time the exhaust gas temperature (EGT) should reach a maximum.

The exact value of EGT can be obtained by pressing the far left button on the engine instrument unit VM 1000. In the "lean" mode one bar represents 10 °F.

Best Power Mixture

The mixture can be set for maximum performance at all power settings. The mixture should first be set as for 'best economy'. The mixture should then be enriched until the exhaust gas temperature is approximately 100 °F lower.

This mixture setting produces the maximum performance for a given manifold pressure and is mainly used for high power settings (approximately 75 %).

4A.3.11 DESCENT

1. Mixture control lever adjust as required for the altitude,
operate slowly
2. RPM lever 1800 - 2400 RPM
3. Throttle as required

CAUTION

When reducing power, the change in cylinder head temperature should not exceed 50 °F per minute. This is normally guaranteed by the 'self adapting inlet'. An excessive cooling rate may occur however, when the engine is very hot and the throttle is reduced abruptly in a fast descent. This will be indicated by a flashing cylinder head temperature indication.

CAUTION

When the fuel pressure warning light illuminates, or the fuel pressure indication is below the green sector, the electrical fuel pump must be switched ON.

4A.3.11 DESCENT

Item 4 is added behind item 3

4. Electrical fuel pump ON at high altitudes

The second CAUTION on this page is replaced by the following CAUTION:

CAUTION

Operation at high altitudes with the electrical fuel pump OFF may cause vapor bubbles, resulting in intermittent low fuel pressure indications , sometimes followed by high fuel flow indications.

4A.3.12 LANDING APPROACH

Item 13 of the checklist and the NOTE are amended to read:

	13. Approach speed	73 KIAS (1200 kg, 2646 lb)
		71 KIAS (1150 kg, 2535 lb)
		67 KIAS (1092 kg, 2407 lb)
		63 KIAS (1000 kg, 2205 lb)
		58 KIAS (850 kg, 1874 lb)

NOTE

|
|
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|
|

In case of airplanes with a maximum landing mass less than the maximum permitted flight mass, a landing with a higher mass constitutes an abnormal operating procedure. Refer to Sections 2.7 and 4B.6.

4A.3.12 LANDING APPROACH

[...]

before landing:

[...]

13. Approach speed 71 KIAS (1150 kg, 2535 lb)
67 KIAS (1092 kg, 2407 lb)
63 KIAS (1000 kg, 2205 lb)
58 KIAS (850 kg, 1874 lb)

NOTE

In case of airplanes with a maximum landing mass of 1092 kg (2407 lb), a landing with a higher mass constitutes an abnormal operating procedure. Refer to Sections 2.7 and 4.B6.

[...]

4A.3.13 GO-AROUND

Item 2 of the checklist is amended to read:

- 2. Airspeed 67 KIAS (1200 kg, 2646 lb)
66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)

above a safe height:

Item 5 of the checklist is amended to read:

- 5. Airspeed 76 KIAS (1200 kg, 2646 lb)
73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

4A.3.13 GO-AROUND

1. Throttle MAX PWR
2. Airspeed 66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)
3. Flaps T/O

above a safe height:

4. RPM lever 2400 RPM
5. Airspeed 73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)
6. Flaps UP
7. Electrical fuel pump OFF

4A.3.14 AFTER LANDING

1. Throttle IDLE
2. Brakes as required
3. Electrical fuel pump OFF
4. Transponder OFF / SBY
5. Pitot heating OFF
6. Avionics as required
7. Lights as required
8. Flaps UP

4A.3.15 ENGINE SHUT-DOWN

1. Parking brake set
2. Engine instruments check
3. Avionics master switch OFF
4. All electrical equipment OFF
5. Throttle 1000 RPM
6. Ignition check OFF until RPM drops noticeably,
then immediately BOTH again
7. Mixture control lever LEAN - shut engine off
8. Ignition switch OFF
9. Master switch OFF

4A.3.16 POST-FLIGHT INSPECTION

1. Ignition switch OFF, remove key
2. Master switch ON
3. Avionics master switch ON
4. ELT check activated:
listen on 121.5 MHz
5. Avionics master switch OFF
6. Master switch OFF
7. Parking brake release, use chocks
8. Airplane moor, if unsupervised for extended
period

NOTE

If the airplane is not operated for more than 5 days, the long-term parking procedure should be applied. If the airplane is not operated for more than 30 days, the storage procedure should be applied. Both procedures are described in the Airplane Maintenance Manual (Doc. No. 6.02.01) in Chapter 10.

4A.3.17 FLIGHT IN RAIN**NOTE**

Performance deteriorates in rain; this applies particularly to the take-off distance and to the maximum horizontal speed. The effect on the flight characteristics is minimal. Flight through very heavy rain should be avoided because of the associated visibility problems.

4A.3.18 REFUELING**CAUTION**

Before refueling, the airplane must be connected to electrical ground. Grounding points: unpainted areas on steps, left and right.

4A.3.19 FLIGHT AT HIGH ALTITUDE

At high altitudes the provision of oxygen for the occupants is necessary. Legal requirements for the provision of oxygen should be adhered to.

Also see Section 2.11 OPERATING ALTITUDE.

CHAPTER 4B

ABNORMAL OPERATING PROCEDURES

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4B.1 PRECAUTIONARY LANDING

Item 4 of the checklist is amended to read:

4. Airspeed 76 KIAS (1200 kg, 2646 lb)
73 KIAS (1150 kg, 2535 lb)
68 KIAS (1000 kg, 2205 lb)
60 KIAS (850 kg, 1874 lb)

on final approach:

6. Flaps LDG
7. Safety harnesses tighten
8. Touchdown with the lowest possible airspeed

CAUTION

If sufficient time is remaining, the risk in the event of a collision with obstacles can be reduced as follows:

- Fuel tank selector OFF
- Ignition switch OFF
- Master switch OFF

4B.2 INSTRUMENT INDICATIONS OUTSIDE OF GREEN RANGE

(a) High oil pressure when starting in low ambient temperatures

- Reduce RPM and re-check oil pressure at a higher oil temperature.
- If on reducing the RPM the indicated oil pressure does not change, it is probable that the fault lies in the oil pressure indication. Terminate flight preparation.

(b) High manifold pressure

If the manifold pressure indicator is clearly above the green range, the reading is faulty. In this case the performance settings should be undertaken by means of the lever settings. The airplane should be serviced.

(c) Oil temperature

A constant reading of the oil temperature of 26 °F or 317 °F suggests a faulty oil temperature sensor. The airplane should be serviced.

(d) Cylinder head temperature and exhaust gas temperature

A very low reading of CHT or EGT for a single cylinder may be the result of a loose sensor. In this case the reading will indicate the temperature of the engine compartment. The airplane should be serviced.

4B.3 FAILURES IN THE ELECTRICAL SYSTEM

(a) Low voltage caution (VOLT or LOW VOLTS)

This caution is indicated when the normal on-board voltage (28 V) drops below 24 V.

Possible reasons are:

- A fault in the power supply.
- RPM too low.

(i) 'Low Voltage' caution on the ground:

1. Engine speed 1200 RPM
2. Electrical equipment OFF
3. Ammeter check

If the caution light does not go out, and the ammeter flashes and reads zero:

- Terminate flight preparation.

(ii) 'Low Voltage' caution during flight:

1. Electrical equipment OFF if not needed
2. Ammeter check

If the caution light does not go out, and the ammeter flashes and reads zero:

- Follow procedure in 3.7.2 (b) Alternator failure.

(iii) 'Low Voltage' caution during landing:

- Follow (i) after landing.

(b) Electronic ignition control unit

If the electronic ignition control unit is installed but inoperative, the white status light for the ignition (IGN or IGNITION) will be illuminated, and the conventional magneto ignition will take over the ignition control.

The flight can be continued normally. However, fuel consumption will slightly increase, and engine starting will become difficult.

4B.4 TAKE-OFF FROM A SHORT GRASS STRIP

Item 8 of the checklist is amended to read:

8. Airspeed 67 KIAS (1200 kg, 2646 lb)
66 KIAS (1150 kg, 2535 lb)
60 KIAS (1000 kg, 2205 lb)
54 KIAS (850 kg, 1874 lb)

4B.4 TAKE-OFF FROM A SHORT GRASS STRIP

1. Brakes apply
2. Flaps T/O
3. Throttle MAX PWR
4. Elevator (control stick) fully aft
5. Brakes release
6. Hold direction using rudder

NOTE

In strong crosswinds steering can be augmented by use of the toe brakes. It should be noted, however, that this method increases the take-off roll, and should not generally be used.

7. Elevator (control stick) release slowly, when nose wheel has lifted
 Allow airplane to lift off as soon as possible and increase speed at low level. *5.000000*
8. Airspeed 66 KIAS (1150 kg, 2535 lb)
 60 KIAS (1000 kg, 2205 lb)
 54 KIAS (850 kg, 1874 lb)
9. RPM lever 2400 RPM, above safe altitude
10. Flaps UP, above safe altitude
11. Electrical fuel pump OFF, above safe altitude
12. Landing light as required

4B.5 FAILURES IN THE FLAP OPERATING SYSTEM

The following paragraph is amended to read:

Modified approach procedure depending on the available flap setting

(a) Only UP available:

Airspeed	76 KIAS (1200 kg, 2646 lb)
	73 KIAS (1150 kg, 2535 lb)
	68 KIAS (1000 kg, 2205 lb)
	60 KIAS (850 kg, 1874 lb)

Land at a flat approach angle, use throttle to control airplane speed and rate of descent.

(b) Only T/O available:

Airspeed	76 KIAS (1200 kg, 2646 lb)
	73 KIAS (1150 kg, 2535 lb)
	68 KIAS (1000 kg, 2205 lb)
	60 KIAS (850 kg, 1874 lb)

Land at a flat approach angle, use throttle to control airplane speed and rate of descent.

(c) Only LDG available:

Perform normal landing.

4B.5 FAILURES IN FLAP OPERATING SYSTEM

Failure in position indication or function

- Check flap position visually.
- Keep airspeed in white sector.
- Re-check all positions of the flap switch, as both of the flap end positions (T/O and LDG) are highly reliable.

Modified approach procedure depending on the available flap setting

(a) Only UP available:

SUPERCEDED

Airspeed ~~73 KIAS (1150 kg, 2535 lb)~~
~~68 KIAS (1000 kg, 2205 lb)~~
~~60 KIAS (850 kg, 1874 lb)~~

Land at a flat approach angle, use throttle to control airplane speed and rate of descent.

(b) Only T/O available:

Airspeed ~~73 KIAS (1150 kg, 2535 lb)~~
~~68 KIAS (1000 kg, 2205 lb)~~
~~60 KIAS (850 kg, 1874 lb)~~

Land at a flat approach angle, use throttle to control airplane speed and rate of descent.

(c) Only LDG available:

Perform normal landing.

4B.6 LANDING WITH HIGH LANDING MASS

This Section is amended to read:

NOTE

This Section only applies to airplanes with a maximum landing mass less than the maximum flight mass. All landings with a current flight mass not exceeding the maximum permissible landing mass constitute a normal operating procedure. Refer to Sections 2.7 and 4A.3.12.

NOTE

The maximum landing mass given in Chapter 2 is the highest mass for landing conditions at the maximum descent velocity. This velocity was used in the strength calculations to determine the landing gear loads during a particularly hard landing.

Perform landing approach and landing according to Chapter 4A, but maintain an increased airspeed during landing approach.

Approach speed 73 KIAS (1200 kg, 2646 lb)
71 KIAS (1150 kg, 2535 lb)

WARNING

Damage of the landing gear can result from a hard landing with a flight mass above the maximum landing mass.

4B.6 LANDING WITH HIGH LANDING MASS**NOTE**

The maximum landing mass given in Chapter 2 is the highest mass for landing conditions at the maximum descent velocity. This velocity was used in the strength calculations to determine the landing gear loads during a particularly hard landing.

Perform landing approach and landing according to Chapter 4A, but maintain an increased airspeed during landing approach.

Approach speed 71 KIAS (1150 kg, 2535 lb)

WARNING

Damage of the landing gear can result from a hard landing with a flight mass above the maximum landing mass.

4B.6 LANDING WITH HIGH LANDING MASS

The following NOTE is inserted in front of the existing text:

NOTE

This Section only applies to airplanes with a maximum landing mass of 1092 kg (2407 lb). In case of airplanes with a maximum landing mass of 1150 kg (2535 lb) a landing with a mass between 1092 kg and 1150 kg (2407 and 2535 lb) constitutes a normal operating procedure. Refer to Sections 2.7 and 4A.3.12.

CHAPTER 5

PERFORMANCE

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5.1 INTRODUCTION

The performance tables and diagrams on the following pages are presented so that, on the one hand, you can see what performance you can expect from your airplane, while on the other they allow comprehensive and sufficiently accurate flight planning. The values in the tables and the diagrams were obtained in the framework of the flight trials using an airplane and power-plant in good condition, and corrected to the conditions of the International Standard Atmosphere (ISA = 15 °C/59 °F and 1013.25 hPa/29.92 inHg at sea level).

- The performance diagrams do not take into account variations in pilot experience or a poorly maintained airplane. The performances given can be attained if the procedures quoted in this manual are applied, and the airplane has been well maintained.

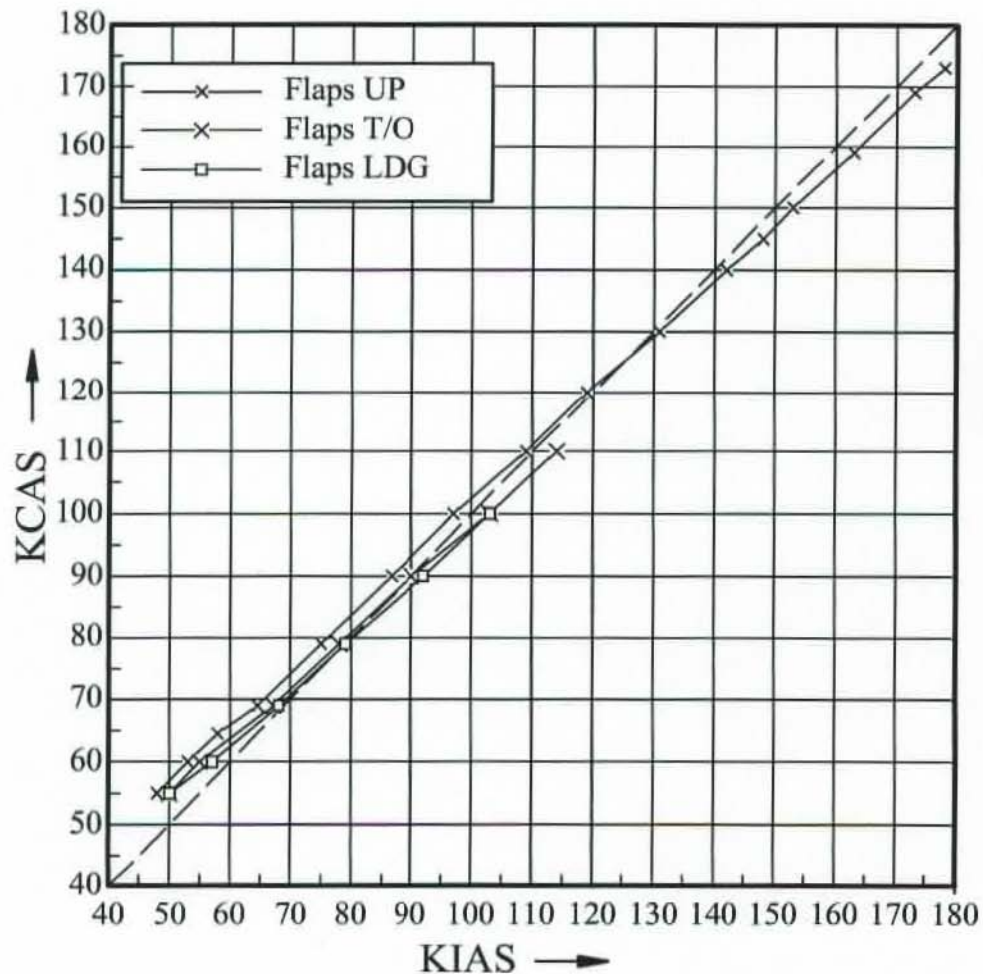
Where appropriate, any flight performance degradation resulting from the absence of wheel fairings is given as a percentage.

5.2 USE OF THE PERFORMANCE TABLES AND DIAGRAMS

- In order to illustrate the influence of a number of different variables, the performance data is reproduced in the form of tables or diagrams. These contain sufficiently detailed information so that conservative values can be selected and used for the determination
- of adequate performance data for the planned flight.

5.3 PERFORMANCE TABLES AND DIAGRAMS

5.3.1 AIRSPEED CALIBRATION



5.3.2 TABLE FOR SETTING ENGINE PERFORMANCE

			Engine power as % of max. take-off power						
			45 %				55 %		
	RPM		1800	2000	2200	2400	2000	2200	2400
Fuel Flow [US gal/h]	Best Economy		5.8	6	6.3	6.6	7	7.2	7.5
	Best Power		-	-	7.3	7.7	-	8.5	8.7
ISA	[°C]	[°F]	Manifold Pressure (MP) [inHg]						
MSL	15	59	22.7	21.3	20.2	19.0	23.9	22.4	21.2
1000	13	55	22.4	21.0	19.9	18.7	23.6	22.2	21.0
2000	11	52	22.1	20.7	19.6	18.4	23.3	21.9	20.7
3000	9	48	21.8	20.4	19.3	18.2	23.0	21.6	20.4
4000	7	45	21.5	20.2	19.0	17.9	22.7	21.2	20.1
5000	5	41	21.2	19.9	18.7	17.6	22.3	20.9	19.8
6000	3	38	20.9	19.6	18.4	17.4	22.0	20.6	19.5
7000	1	34	20.5	19.3	18.2	17.1	21.7	20.3	19.3
8000	-1	31	20.2	19.0	17.9	16.9	21.3	20.0	19.0
9000	-3	27	19.9	18.7	17.6	16.6	21.1	19.7	18.7
10000	-5	23	19.6	18.4	17.3	16.3	-	19.4	18.4
11000	-7	19	19.3	18.2	17.0	16.1		19.1	18.1
12000	-9	16	-	17.9	16.7	15.8		-	17.8
13000	-11	12		17.6	16.4	15.5			17.6
14000	-13	9		-	16.1	15.3			-
15000	-15	6			15.8	15.0			
16000	-17	2			15.5	14.7			
17000	-19	-2			-	14.5			

			Engine power as % of max. take-off power				
			65 %		75 %		
	RPM		2000	2200	2400	2200	2400
Fuel Flow [US gal/h]	Best Economy		7.9	8.2	8.5	9.2	9.5
	Best Power		-	9.5	9.8	10.7	11
ISA	[°C]	[°F]	Manifold Pressure (MP) [inHg]				
MSL	15	59	26.8	24.9	23.4	27.3	25.8
1000	13	55	26.4	24.5	23.2	26.8	25.5
2000	11	52	26.0	24.2	22.9	26.5	25.2
3000	9	48	25.7	23.8	22.6	26.1	24.8
4000	7	45	25.4	23.5	22.3	-	24.5
5000	5	41	-	23.1	22.0		24.1
6000	3	38		22.8	21.7		-
7000	1	34		22.4	21.4		
8000	-1	31		-	21.0		
9000	-3	27			20.7		
10000	-5	23			-		

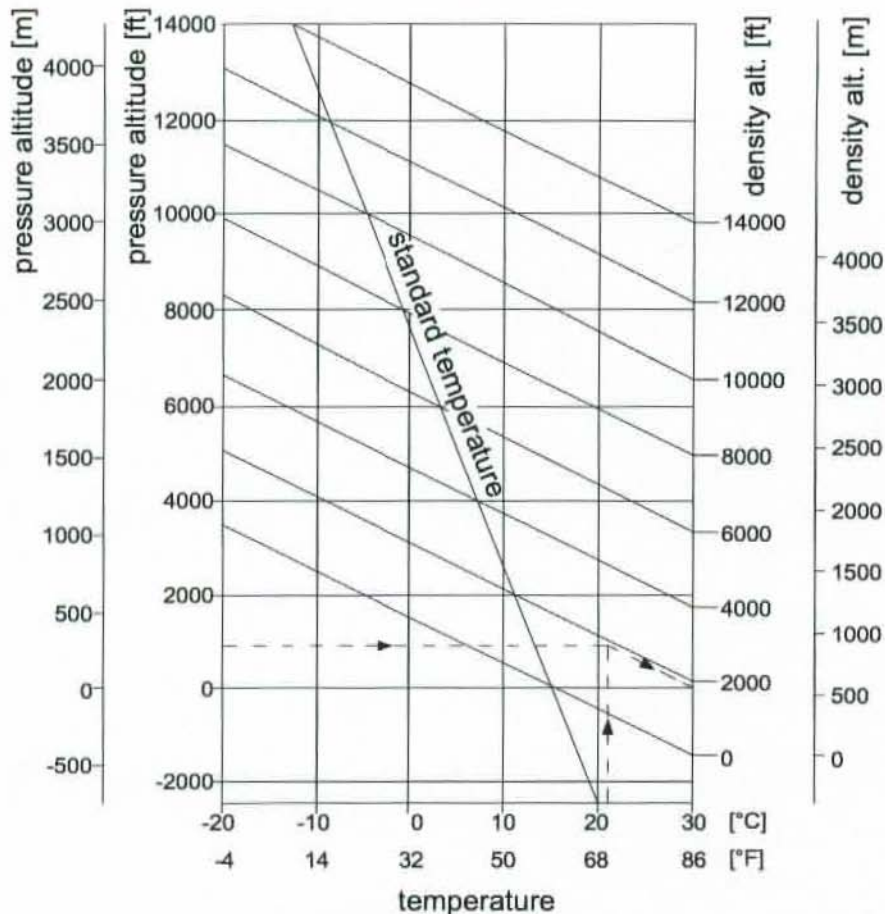
The areas shaded grey under each RPM heading are the recommended bands.

Correcting the table for variation from Standard Temperature

- At ISA + 15 °C (ISA + 27 °F) the performance values fall by approx. 3 % of the power selected according to the above table.
- At ISA - 15 °C (ISA - 27 °F) the performance values rise by approx. 3 % of the power selected according to the above table.

5.3.3 PRESSURE ALTITUDE - DENSITY ALTITUDE

Conversion from pressure altitude to density altitude.



- Example:
1. Set 1013.25 hPa on altimeter and read pressure altitude (900 ft).
 2. Establish ambient temperature (+21 °C).
 3. Read off density altitude (1800 ft).

Result: From a performance calculation standpoint the airplane is at 1800 ft.

5.3.4 STALLING SPEEDS

The following table is added:

Mass: 1200 kg (2646 lb) (if MAM 40-227 is carried out)

Airspeeds in KIAS

1200 kg		Bank Angle			
		0°	30°	45°	60°
Flaps	UP	53	58	68	83
	T/O	52	57	67	81
	LDG	52	57	66	80

5.3.4 STALLING SPEEDS

Mass: 980 kg (2161 lb)

Airspeeds in KIAS

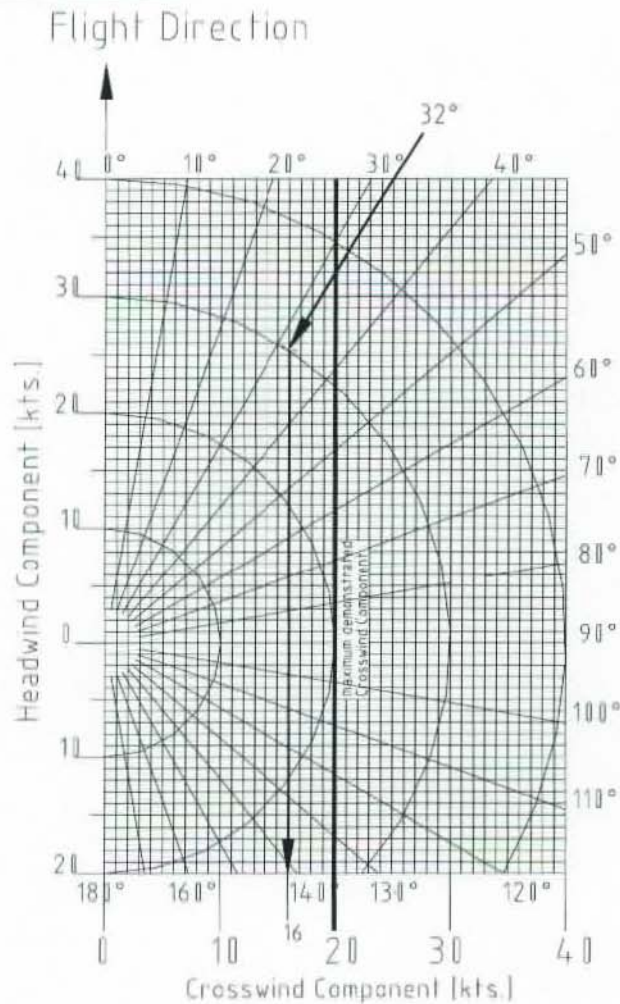
980 kg		Bank Angle			
		0°	30°	45°	60°
Flaps	UP	47	52	58	73
	T/O	44	51	58	72
	LDG	42	49	57	71

Mass: 1150 kg (2535 lb)

Airspeeds in KIAS

1150 kg		Bank Angle			
		0°	30°	45°	60°
Flaps	UP	52	57	66	79
	T/O	51	55	64	78
	LDG	49	55	62	76

5.3.5 WIND COMPONENTS



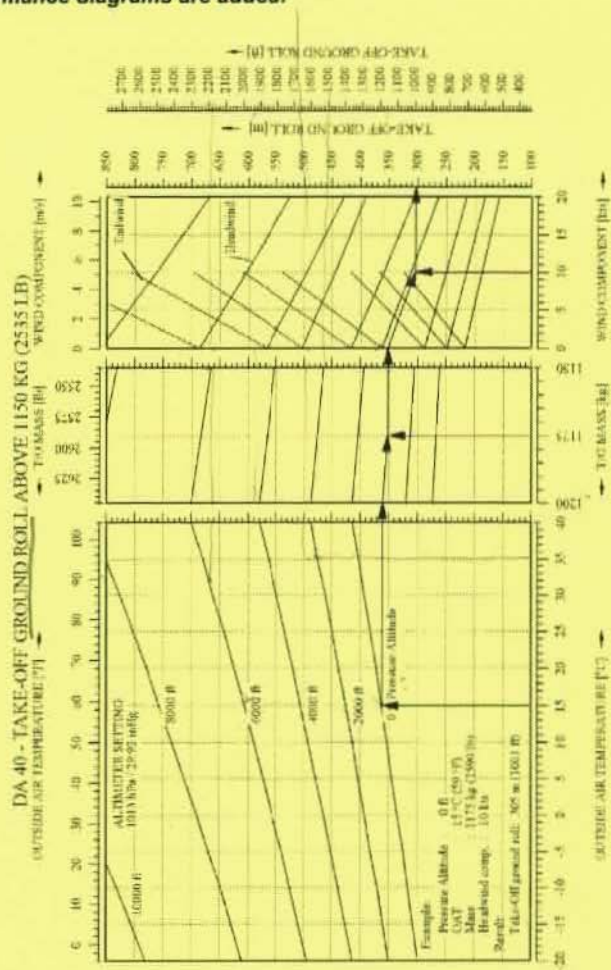
Example: Flight direction : 360°
 Wind : 32°/30 kts
 Result: Crosswind component : 16 kts
 Max. demonstrated crosswind component : 20 kts

5.3.6 TAKE-OFF DISTANCE

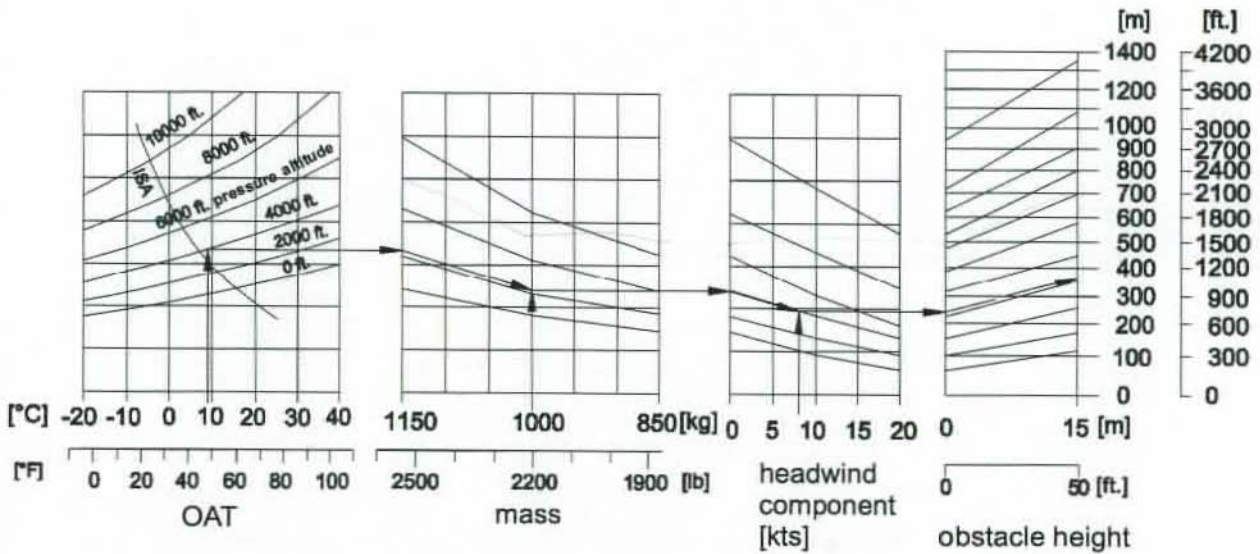
The checklist is amended to read:

- Conditions:
- Throttle MAX PWR
 - RPM lever 2700 RPM
 - Flaps T/O
 - Lift-off speed approx. 59 KIAS
 - Climb-out speed 67 KIAS (1200 kg, 2646 lb)
66 KIAS (1150 kg, 2535 lb)
60 KIAS (below 1000 kg, 2205 lb)
 - Runway level, asphalt surface

The performance diagrams are added:



Take-off distance



Example:

Pressure altitude : 4000 ft
 OAT : 8 °C (46 °F)
 Mass : 1000 kg (2205 lb)
 Headwind comp. : 8 kts

Result:

T/O roll : approx. 240 m (787 ft)
 T/O distance over 50 ft obstacle : approx. 370 m (1214 ft)

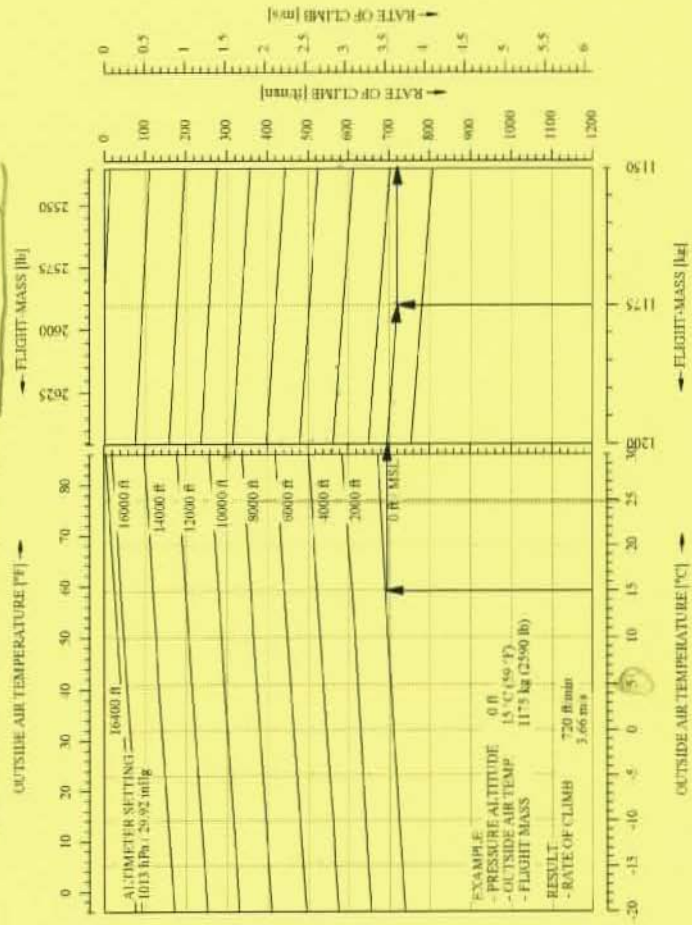
5.3.7 CLIMB PERFORMANCE- TAKE-OFF CLIMB

The checklist is amended to read:

<u>Conditions:</u>	- Throttle	MAX PWR
	- RPM lever	2400 RPM
	- Flaps	T/O
	- Airspeed	67 KIAS (1200 kg, 2646 lb)
		66 KIAS (1150 kg, 2535 lb)
		60 KIAS (1000 kg, 2205 lb)
		54 KIAS (850 kg, 1874 lb)

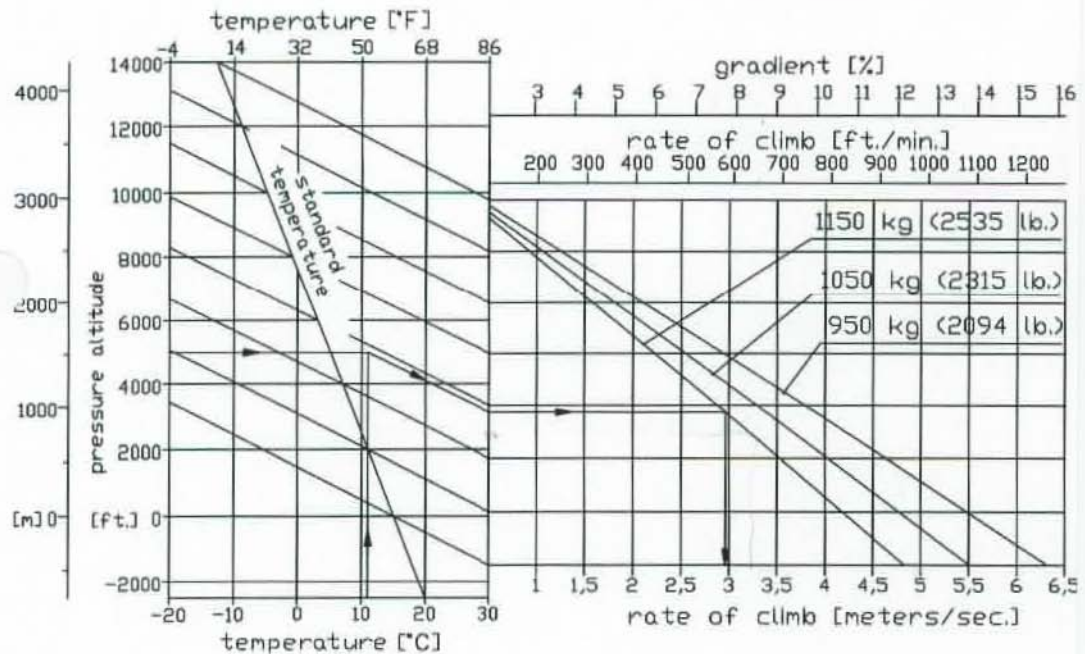
The performance diagram is added:

DA 40 - TAKE-OFF CLIMB PERFORMANCE ABOVE 1150 KG (2535/LB)



5.3.7 CLIMB PERFORMANCE - TAKE-OFF CLIMB

- Conditions:
- Throttle MAX PWR
 - RPM lever 2400 RPM
 - Flaps T/O
 - Airspeed 66 KIAS (1150 kg, 2535 lb)
 60 KIAS (1000 kg, 2205 lb)
 54 KIAS (850 kg, 1874 lb)



Example: Pressure altitude 5000 ft
 Temperature 12 °C (54 °F)
 Take-off mass 1150 kg (2535 lb)

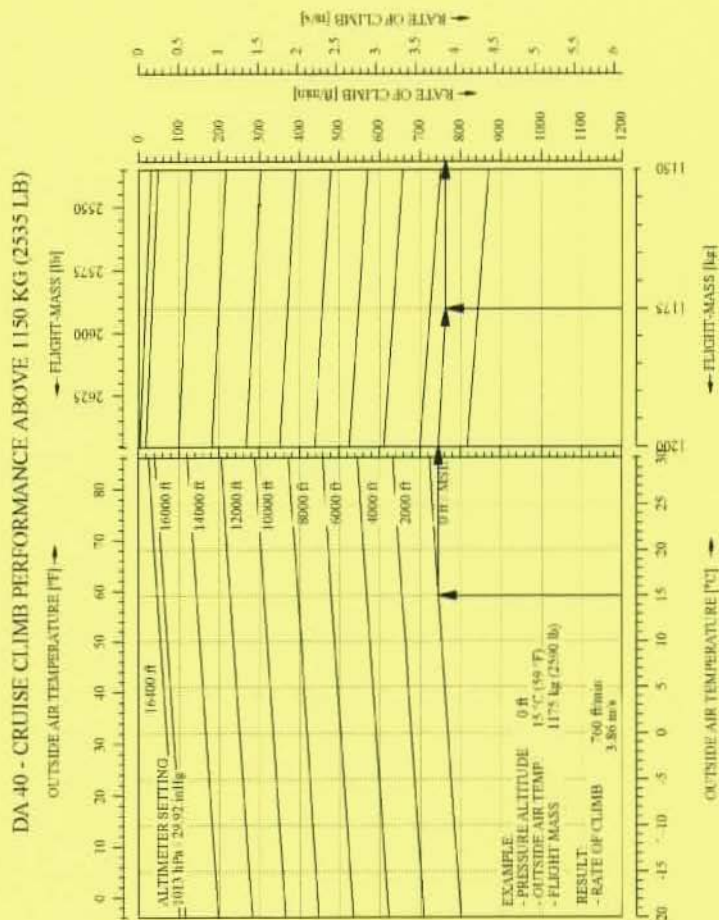
Result: Rate of climb 3.0 m/s (591 ft/min)

5.3.8 CLIMB PERFORMANCE - CRUISE CLIMB

The checklist is amended to read:

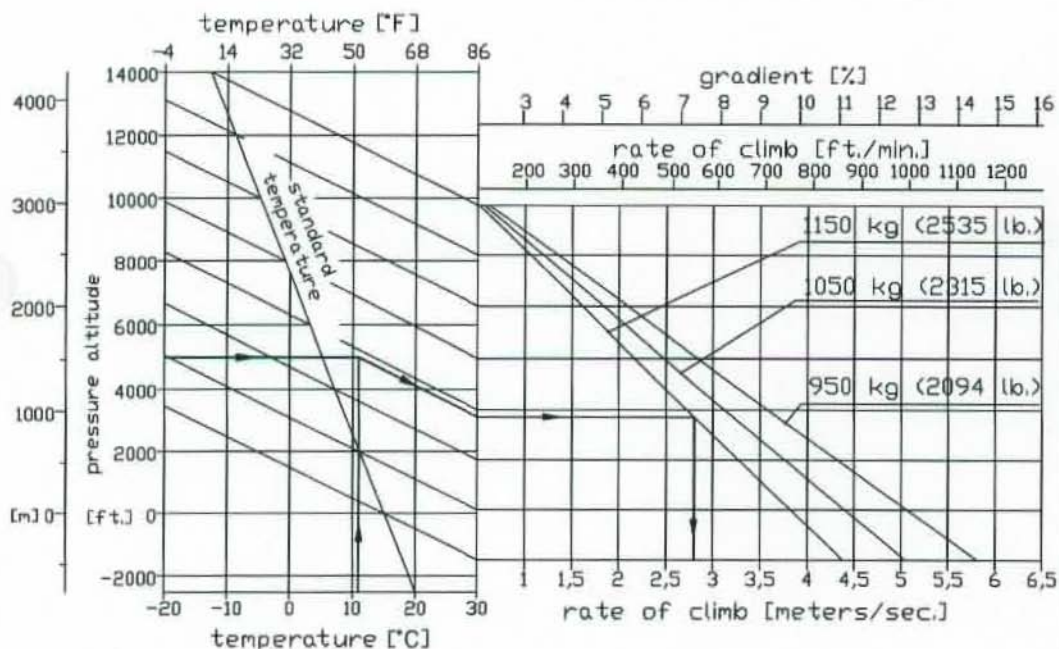
Conditions:	- Throttle	MAX PWR
	- RPM lever	2400 RPM
	- Flaps	UP
	- Airspeed	76 KIAS (1200 kg, 2646 lb)
		73 KIAS (1150 kg, 2535 lb)
		68 KIAS (1000 kg, 2205 lb)
		60 KIAS (850 kg, 1874 lb)

The performance diagram is added:



5.3.8 CLIMB PERFORMANCE - CRUISE CLIMB

- Conditions:
- Throttle MAX PWR
 - RPM lever 2400 RPM
 - Flaps UP
 - Airspeed 73 KIAS (1150 kg, 2535 lb)
 68 KIAS (1000 kg, 2205 lb)
 60 KIAS (850 kg, 1874 lb)

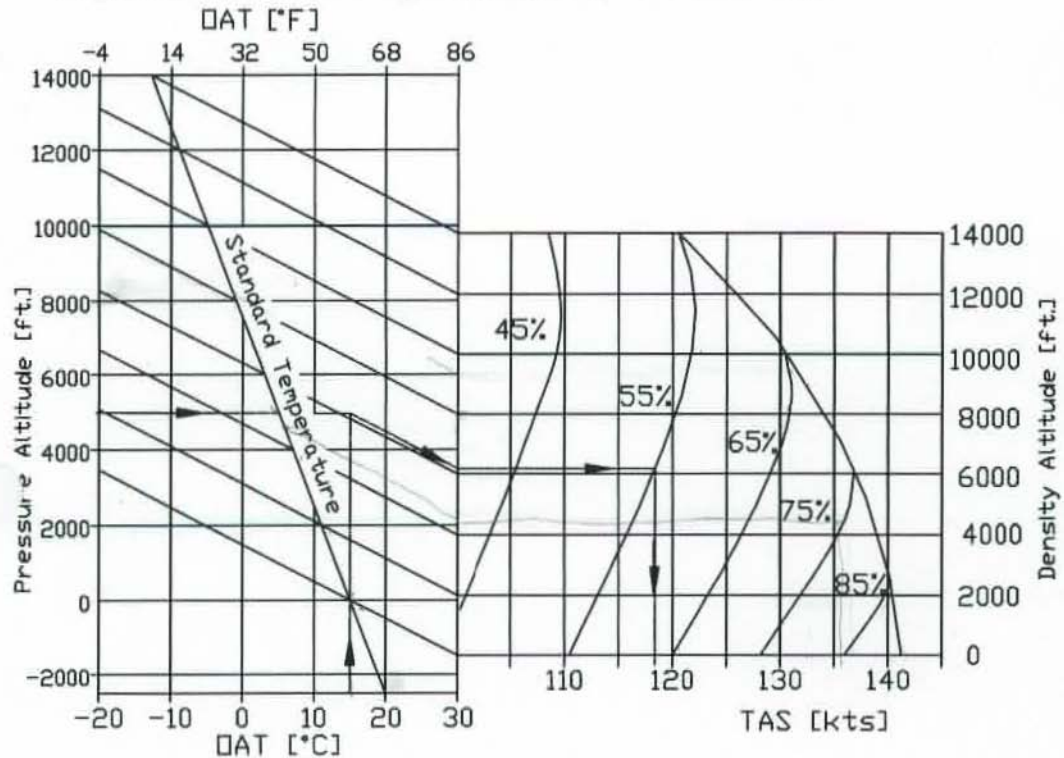


Example: Pressure altitude 5000 ft
 Temperature 12 °C (54 °F)
 Take-off mass 1150 kg (2535 lb)

Result: Rate of climb 2.8 m/s (551 ft/min)

5.3.9 CRUISING (TRUE AIRSPEED TAS)

Diagram to establish True Airspeed (TAS) at a given power setting.



Example: Pressure altitude 5000 ft
 Temperature 15 °C (59 °F)
 Power setting 55 %

Result: True Airspeed (TAS) 118 kts

CAUTION

In case of operation without wheel fairings the cruising speed reduces by approximately 5 %.

5.3.10 LANDING DISTANCE - FLAPS LDG**The checklist is amended to read:**

- Conditions:
- Throttle IDLE
 - RPM lever HIGH RPM
 - Flaps LDG
 - Approach speed 73 KIAS (1200 kg, 2646 lb)
71 KIAS (1150 kg, 2535 lb)
63 KIAS (1000 kg, 2205 lb)
58 KIAS (850 kg, 1874 lb)
 - Runway level, asphalt surface

The NOTE is added in front of the existing table:**NOTE**

A landing mass above 1150 kg (2535 lb) up to 1200 kg (2646 lb) will increase the landing distance over a 50 ft (15 m) obstacle and the landing ground roll distance up to 6%.

5.3.10 LANDING DISTANCE - FLAPS LDG

- Conditions:
- Throttle IDLE
 - RPM lever HIGH RPM
 - Flaps LDG
 - Approach speed 71 KIAS (1150 kg, 2535 lb)
63 KIAS (1000 kg, 2205 lb)
58 KIAS (850 kg, 1874 lb)
 - Runway level, asphalt surface

values for ISA and MSL, at 1150 kg (2535 lb)	
Landing distance over a 50 ft (15 m) obstacle	approx. 638 m (2093 ft)
Ground roll	approx. 352 m (1155 ft)

CAUTION

On grass strips, depending on the characteristics of the surface (particularly moisture), allowance must be made for a ground roll which is up to 50 % longer.

A poorly maintained airplane, deviation from the prescribed procedures and unfavorable external factors (high temperature, rain, unfavorable wind effects, etc.) can considerably increase the landing distance.

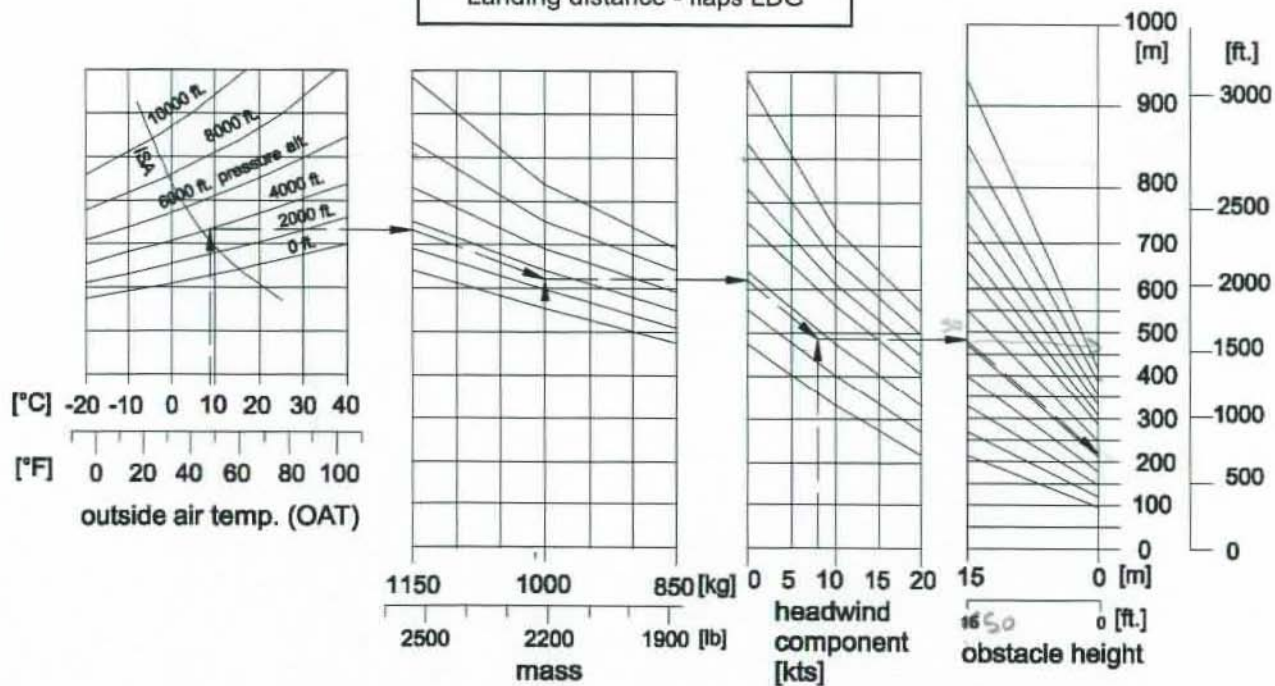
CAUTION

A descending ground slope of 2 % (2 m per 100 m, or 2 ft per 100 ft) results in an increase in the landing distance of approximately 10 %. The effect on the ground roll can be greater.

WARNING

For a safe landing the available runway length must be at least equal to the landing distance over a 50 ft (15 m) obstacle.

Landing distance - flaps LDG

*Example:*

Pressure altitude : 4000 ft
 OAT : 8 °C (46 °F)
 Mass : 1000 kg (2205 lb)
 Headwind comp. : 8 kts

Result:

Landing distance over 50 ft obstacle : approx. 480 m (1575 ft)
 Ground roll : approx. 230 m (755 ft)

5.3.11 LANDING DISTANCE - FLAPS UP

The checklist is amended to read:

- Conditions:
- Throttle IDLE
 - RPM lever HIGH RPM
 - Flaps LDG
 - Approach speed 73 KIAS (1200 kg, 2646 lb)
71 KIAS (1150 kg, 2535 lb)
63 KIAS (1000 kg, 2205 lb)
58 KIAS (850 kg, 1874 lb)
 - Runway level, asphalt surface

The NOTE is added in front of the existing table:

NOTE

A landing mass above 1150 kg (2535 lb) up to 1200 kg (2646 lb) will increase the landing distance over a 50 ft (15 m) obstacle and the landing ground roll distance up to 6%.

5.3.11 LANDING DISTANCE - FLAPS UP

- Conditions:
- Throttle IDLE
 - RPM lever HIGH RPM
 - Flaps UP
 - Approach speed 71 KIAS (1150 kg, 2535 lb)
63 KIAS (1000 kg, 2205 lb)
58 KIAS (850 kg, 1874 lb)
 - Runway level, asphalt surface

values for ISA and MSL, at 1150 kg (2535 lb)

Landing distance over a 50 ft (15 m) obstacle	approx. 775 m (2543 ft)
Ground roll	approx. 471 m (1545 ft)

CAUTION

On grass strips, depending on the characteristics of the surface (particularly moisture), allowance must be made for a ground roll which is up to 50 % longer.

A poorly maintained airplane, deviation from the prescribed procedures and unfavorable external factors (high temperature, rain, unfavorable wind effects, etc.) can considerably increase the landing distance.

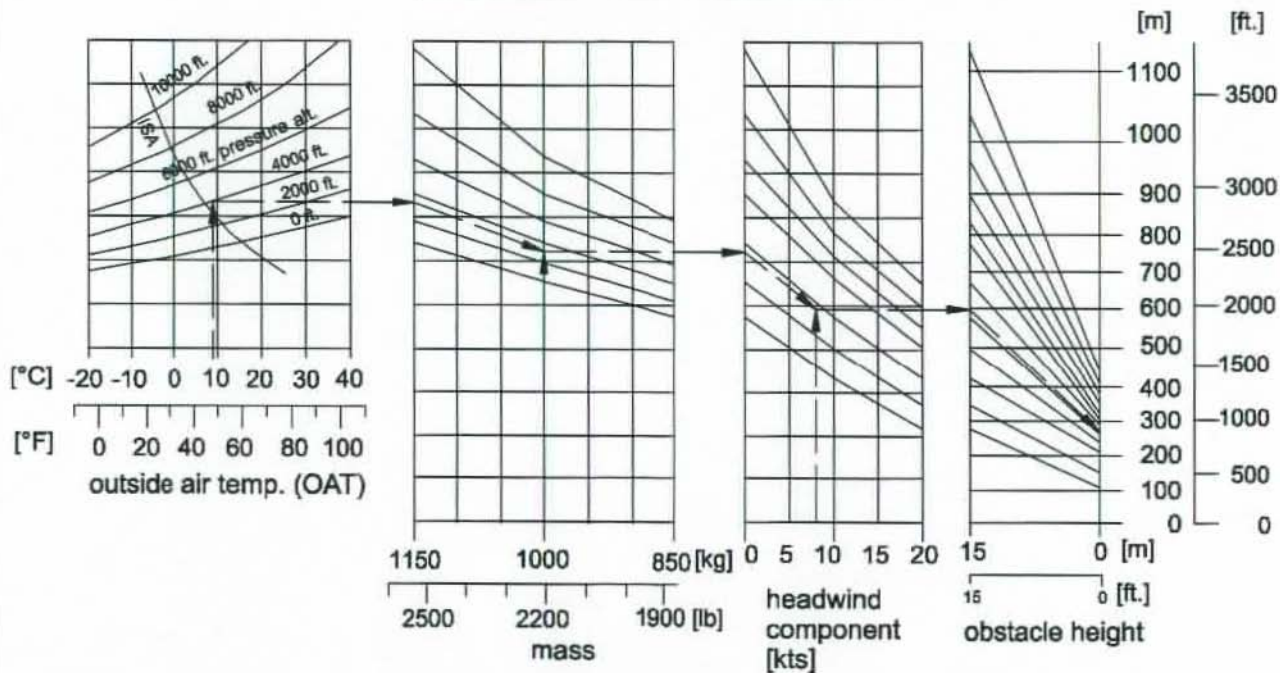
CAUTION

A descending ground slope of 2 % (2 m per 100 m, or 2 ft per 100 ft) results in an increase in the landing distance of approximately 10 %. The effect on the ground roll can be greater.

WARNING

For a safe landing the available runway length must be at least equal to the landing distance over a 50 ft (15 m) obstacle.

Landing distance - flaps UP

*Example:*

Pressure altitude : 4000 ft
 OAT : 8 °C (46 °F)
 Mass : 1000 kg (2205 lb)
 Headwind comp. : 8 kts

Result:

Landing distance over 50 ft obstacle : approx. 580 m (1903 ft)
 Ground roll : approx. 270 m (886 ft)

5.3.12 GRADIENT OF CLIMB ON GO-AROUND

The following is added behind the existing text:

■ If MÄM 40-227 is carried out:

■ The DA 40 reaches a constant gradient of climb of 7 % in the following condition:

- - Mass max. flight mass
(1200 kg, 2646 lb)
- - Power setting Take-off
- - Flaps LDG
- - Airspeed 73 KIAS
- - ISA, MSL

5.3.13 APPROVED NOISE DATA

The following is added:

■ if MÄM 40-227 is carried out:

■ ICAO Annex 16 Chapter X

 : 78.4 dB(A)

■ JAR-36 Subpart C : 78.4 dB(A)

5.3.12 GRADIENT OF CLIMB ON GO-AROUND

The DA 40 reaches a constant gradient of climb of 7.0 % in the following condition:

- Mass max. flight mass (1150 kg, 2535 lb)
- Power setting Take-off
- Flaps LDG
- Airspeed 70 KIAS
- ISA, MSL

5.3.13 APPROVED NOISE DATA

ICAO Annex 16 Chapter X : 69.28 dB(A)
JAR-36 Subpart C : 69.28 dB(A)

CHAPTER 6

MASS AND BALANCE

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6.1 INTRODUCTION

In order to achieve the performance and flight characteristics described in this Airplane Flight Manual and for safe flight operation, the airplane must be operated within the permissible mass and balance envelope.

The pilot is responsible for adhering to the permissible values for loading and center of gravity (CG). In this, he should note the movement of the CG due to fuel consumption. The permissible CG range during flight is given in Chapter 2.

The procedure for determining the flight mass CG position at any point in time is described in this Chapter. Over and above this there is a comprehensive list of the equipment approved for this airplane (Equipment List), as also a list of that equipment installed when the airplane was weighed (Equipment Inventory).

Before the airplane is delivered the empty mass and the corresponding CG position are determined, and entered in Section 6.3 MASS AND BALANCE REPORT.

NOTE

Following equipment changes the new empty mass and the corresponding CG position must be determined by calculation or by weighing.

Following repairs or repainting the new empty mass and the corresponding CG position must be determined by weighing.

Empty mass, empty mass CG position, and the empty mass moment must be certified in the Mass and Balance Report by an authorized person.

NOTE

Refer to Section 1.6 UNITS OF MEASUREMENT for conversion of SI units to US units and vice versa.

6.2 DATUM PLANE

The Datum Plane (DP) is a plane which is normal to the airplane's longitudinal axis and in front of the airplane as seen from the direction of flight. The airplane's longitudinal axis is parallel with the upper surface of a 600:31 wedge which is placed on top of the rear fuselage in front of the vertical stabilizer. When the upper surface of the wedge is aligned horizontally, the Datum Plane is vertical. The Datum Plane is located 2.194 meters (86.38 in) forward of the most forward point of the root rib on the stub wing.

6.3 MASS AND BALANCE REPORT

The empty mass and the corresponding CG position established before delivery are the first entries in the Mass and Balance Report. Every change in permanently installed equipment, and every repair to the airplane which affects the empty mass or the empty mass CG must be recorded in the Mass and Balance Report.

For the calculation of flight mass and corresponding CG position (or moment), the *current* empty mass and the corresponding CG position (or moment) in accordance with the Mass and Balance Report must always be used.

Condition of the airplane for establishing the empty mass:

- Equipment as per Equipment Inventory (see Section 6.5)
- Including brake fluid, lubricant (7.6 liters = 8 qts), plus unusable fuel (4 liters = approx. 1 US gal).

6.4 FLIGHT MASS AND CENTER OF GRAVITY

The following information enables you to operate your DA 40 within the permissible mass and balance limits. For the calculation of the flight mass and the corresponding CG position the following tables and diagrams are required:

- 6.4.1 MOMENT ARMS
- 6.4.2 LOADING DIAGRAM
- 6.4.3 CALCULATION OF LOADING CONDITION
- 6.4.4 PERMISSIBLE CENTER OF GRAVITY RANGE
- 6.4.5 PERMISSIBLE MOMENT RANGE

The diagrams should be used as follows:

1. Take the empty mass and the empty mass moment of your airplane from the Mass and Balance Report, and enter the figures in the appropriate boxes under the column marked 'Your DA 40' in Table 6.4.3 - 'CALCULATION OF LOADING CONDITION'.
2. Read the fuel quantity indicators to determine the fuel quantity. If an indicator shows 15 US gal, up to 20 US gal can be in the tank. In this case, the exact quantity must be determined with the fuel quantity measuring device.

The difference between the actual amount of oil in the engine (check with dipstick) and the maximum oil quantity is called 'Oil not added'; this mass and its related moment are counted as negative. The empty mass of the airplane is established with the maximum amount of oil in the engine, thus the 'missing' oil must be subtracted. If the airplane is flown with maximum oil, the 'Oil not added' entry should be zero.

In our example 6.0 qts have been measured on the dip-stick. We are thus 2.0 qts short of the maximum, which equates to 1.9 liters. Multiplying this quantity by the mass density of 0.89 kilograms per liter gives a mass of 'Oil not added' of 1.7 kg. (In U.S. units: 2.0 qts multiplied by the mass density of 1.86 lb/qts gives a mass of 3.7 lb.)

3. Multiply the individual masses by the moment arms quoted to obtain the moment for every item of loading and enter these moments in the appropriate boxes in Table 6.4.3 - 'CALCULATION OF LOADING CONDITION'.

4. Add up the masses and moments in the respective columns. The total moments may be rounded to whole numbers. The CG position is calculated by dividing the total moment by the total mass (using row 6 for the condition with empty fuel tanks, and row 8 for the pre take-off condition). The resulting CG position must be inside the limits.

As an illustration the total mass and the CG position are entered on Diagram 6.4.4 'PERMISSIBLE CENTER OF GRAVITY RANGE'. This checks graphically that the current configuration of the airplane is within the permissible range.

5. Graphical method:

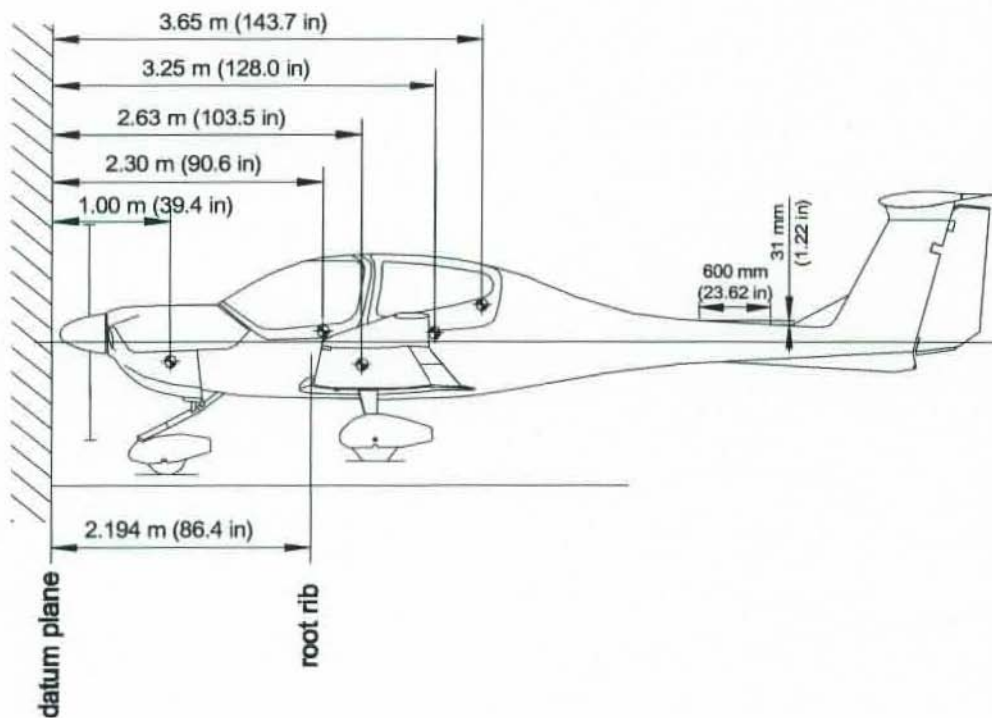
Diagram 6.4.2 'LOADING DIAGRAM' is used to determine the moments. The masses and moments for the individual items of loading are added. Then Diagram 6.4.5 'PERMISSIBLE MOMENT RANGE' is used to check whether the total moment associated with the total mass is in the admissible range.

The result found with the graphical method is however inaccurate. In doubtful cases the result must be verified using the exact method given above.

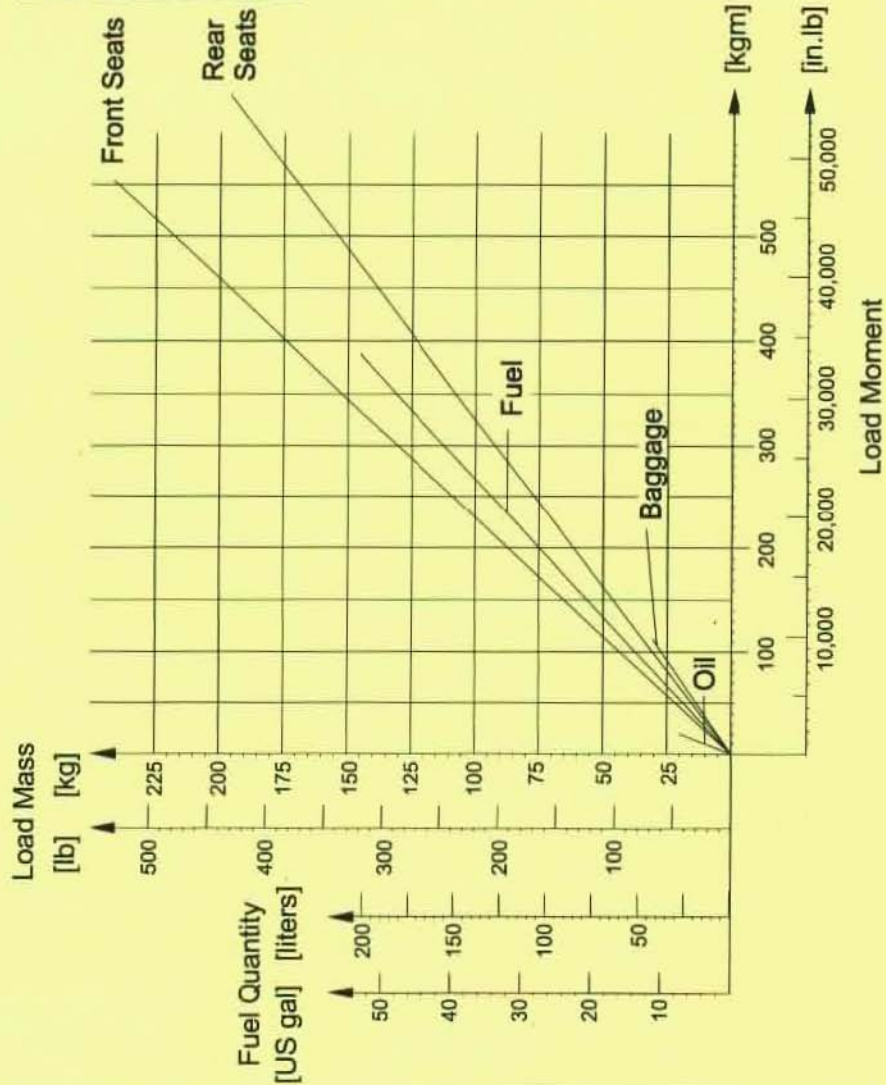
6.4.1 MOMENT ARMS

The most important lever arms aft of the Datum Plane:

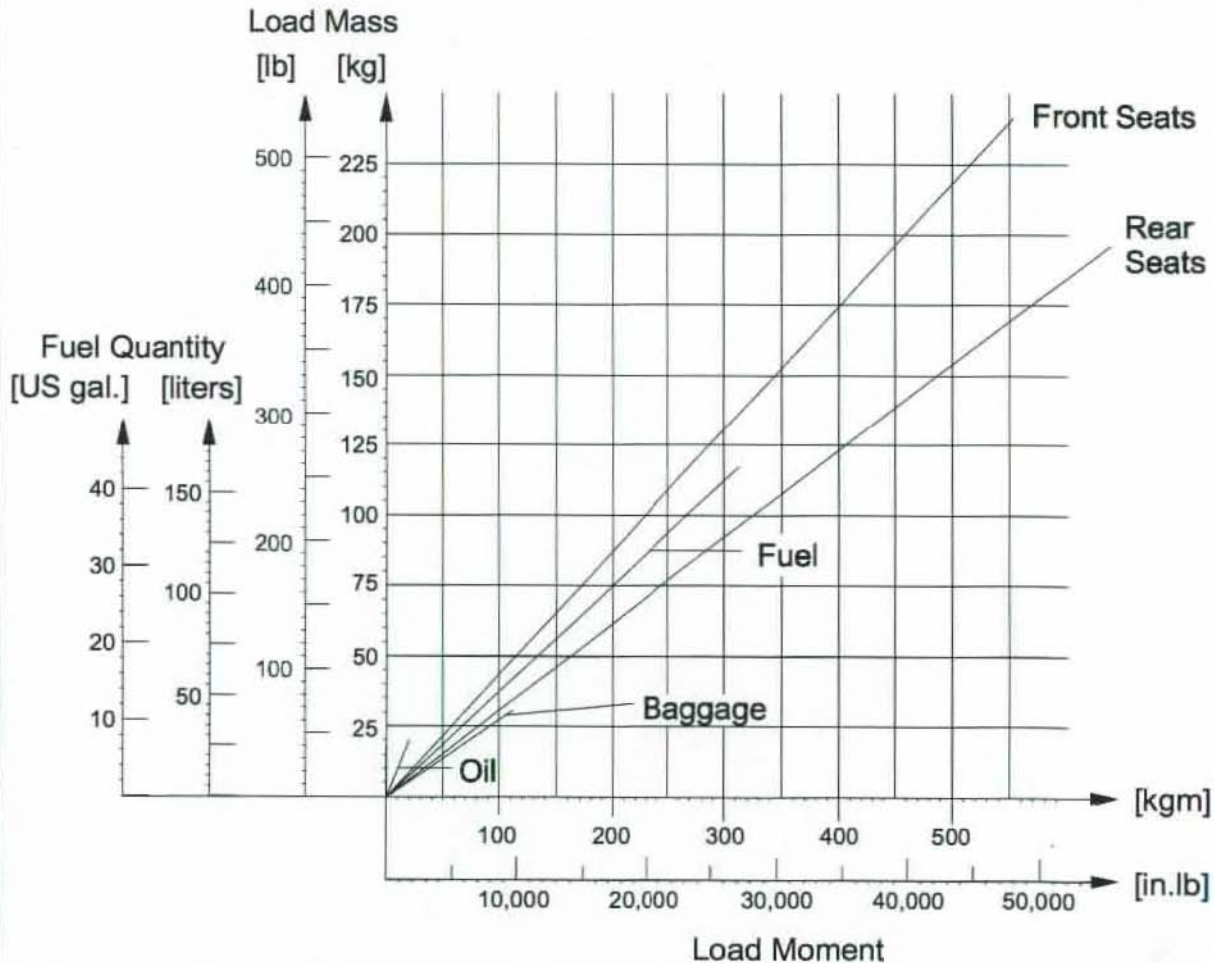
- Oil	:	1.00 m	39.4 in
- Front seats	:	2.30 m	90.6 in
- Rear seats	:	3.25 m	128.0 in
- Wing tank	:	2.63 m	103.5 in
- Baggage	:	3.65 m	143.7 in



6.4.2 LOADING DIAGRAM



6.4.2 LOADING DIAGRAM



6.4.3 CALCULATION OF LOADING CONDITION

CALCULATION OF LOADING CONDITION	DA 40 (Example)		Your DA 40	
	Mass [kg] [lb]	Moment [kgm] [in.lb]	Mass [kg] [lb]	Moment [kgm] [in.lb]
1. Empty mass (from Mass and Balance Report)	735 1620	1820 158,000		
2. Oil not added Lever arm: 1.00 m (39.4 in)	-1.7 -3.7	-1.7 -146		
3. Front seats Lever arm: 2.30 m (90.6 in)	150 331	345 29,989		
4. Rear seats Lever arm: 3.25 m (128.0 in)	75 165.5	244 21,184		
5. Baggage Lever arm: 3.65 m (143.7 in)	0 0	0 0		
6. Total mass and total moment with empty fuel tanks (Total of 1.-5.)	958.3 2112.8	2407 209,027		
7. On-board usable fuel (0.72 kg/liter) (6.01 lb/US gal) Lever arm: 2.63 m (103.5 in)	134 295.4	352 30,574		
8. Total mass and total moment with full fuel tanks (Total 6. plus 7.)	1092.3 2408.2	2759 239,601		
<p>9. The total moments from rows 6 and 8 (2407 and 2759 kgm) (209,027 and 239,601 in.lb) must be divided by the related total mass (958.3 and 1092.3 kg respectively) (2112.8 and 2408.2 lb) and then located in Diagram 6.4.4 'PERMISSIBLE CENTER OF GRAVITY RANGE'.</p> <p>As in our example CG positions (2.512 m and 2.526 m respectively) (98.93 and 99.49 in) and masses fall into the permitted area, this loading condition is allowable.</p>				

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6.4.3 CALCULATION OF LOADING CONDITION

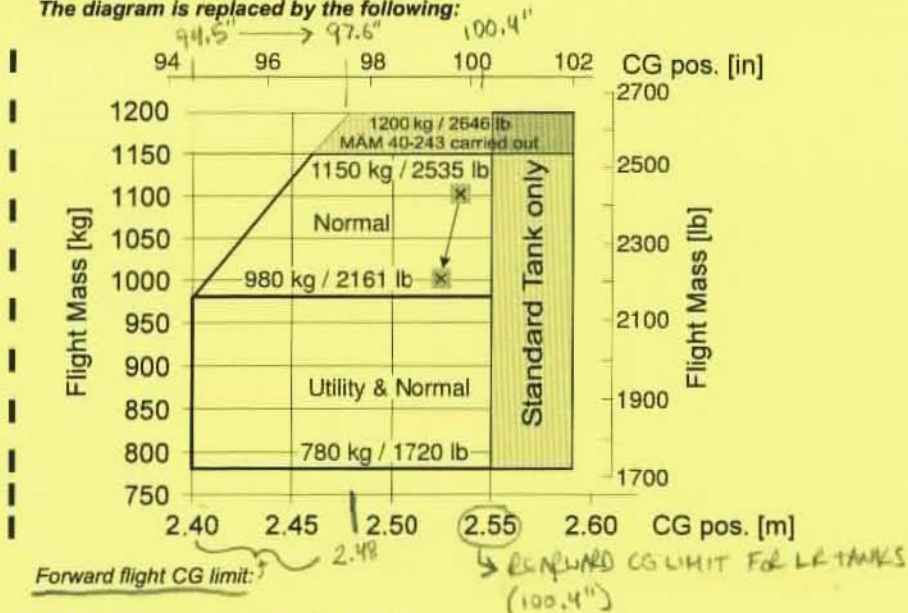
For N566DS

CALCULATION OF LOADING CONDITION	DA 40 (Example)		Your DA 40	
	Mass [kg] [lb]	Moment [kgm] [in.lb]	Mass [kg] [lb]	Moment [kgm] [in.lb]
1. Empty mass (from Mass and Balance Report)	735 1620	1820 158,000	1757	1699
2. Oil not added Lever arm: 1.00 m (39.4 in)	-1.7 -3.7	-1.7 -146		
3. Front seats Lever arm: 2.30 m (90.6 in)	150 331	345 29,989		
4. Rear seats Lever arm: 3.25 m (128.0 in)	150 331	487.5 42,368		
5. Baggage Lever arm: 3.65 m (143.7 in)	0 0	0 0		
6. Total mass and total moment with empty fuel tanks (Total of 1.-5.)	1033.3 2278.3	2650.8 230,211		
7. On-board usable fuel (0.72 kg/liter) (6.01 lb/US gal) Lever arm: 2.63 m (103.5 in)	109.4 242	287.7 25,047		
8. Total mass and total moment with full fuel tanks (Total 6. plus 7.)	1142.7 2520.3	2938.5 255,258		
9. The total moments from rows 6 and 8 (2650.8 and 2938.5 kgm) (230,211 and 255,258 in.lb) must be divided by the related total mass (1033.3 and 1142.7 kg respectively) (2278.3 and 2520.3 lb) and then located in Diagram 6.4.4 'PERMISSIBLE CENTER OF GRAVITY RANGE'. As in our example CG positions (2.565 m and 2.572 m respectively) (101.04 and 101.28 in) and masses fall into the permitted area, this loading condition is allowable.			MAX: 2646	

FRONT CG: 94.5" (216.1LB) - 97.6" (2646) REAR CG: 100.4"

6.4.4 PERMISSIBLE CENTER OF GRAVITY RANGE

The diagram is replaced by the following:



The paragraph is amended to read:

2.40 m (94.5 in) aft of DP from 780 kg to 980 kg (1720 lb to 2161 lb)

2.46 m (96.9 in) aft of DP at 1150 kg (2535 lb)

linear variation between these values

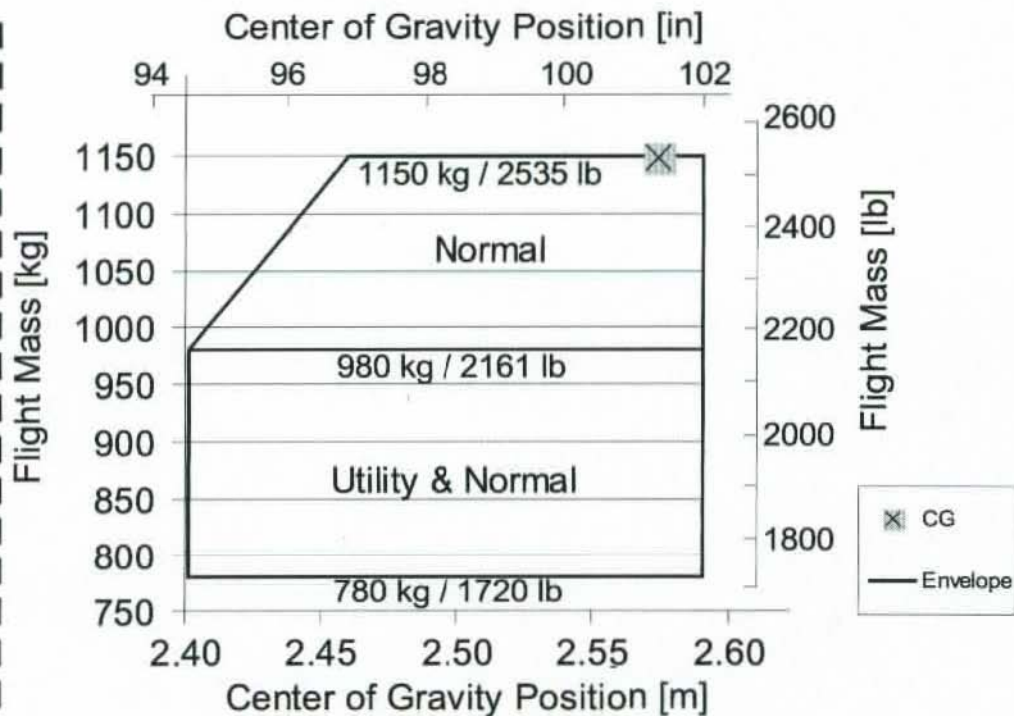
If MAM 40-227 is carried out: 15 IS (100 LB WT INCREASE)

2.40 m (94.5 in) aft of DP from 780 kg to 980 kg (1720 lb to 2161 lb)

2.48 m (97.6 in) aft of DP at 1200 kg (2646 lb)

linear variation between these values

6.4.4 PERMISSIBLE CENTER OF GRAVITY RANGE



The CG shown in the diagram is that from the example in Table 6.4.3 'CALCULATION OF LOADING CONDITION', row 8 (pre take-off condition).

The flight CG position must lie within the following limits:

Most forward flight CG:

2.40 m (94.5 in) aft of Datum Plane at 780 to 980 kg (1720 to 2161 lb)

2.46 m (96.9 in) aft of Datum Plane at 1150 kg (2535 lb)

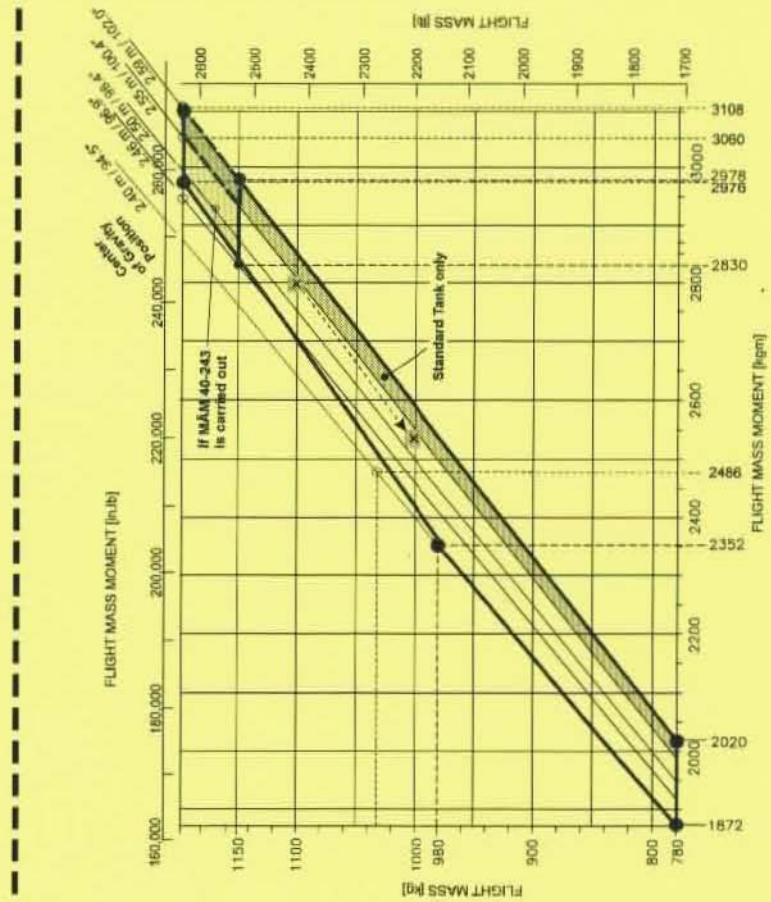
linear variation between these values

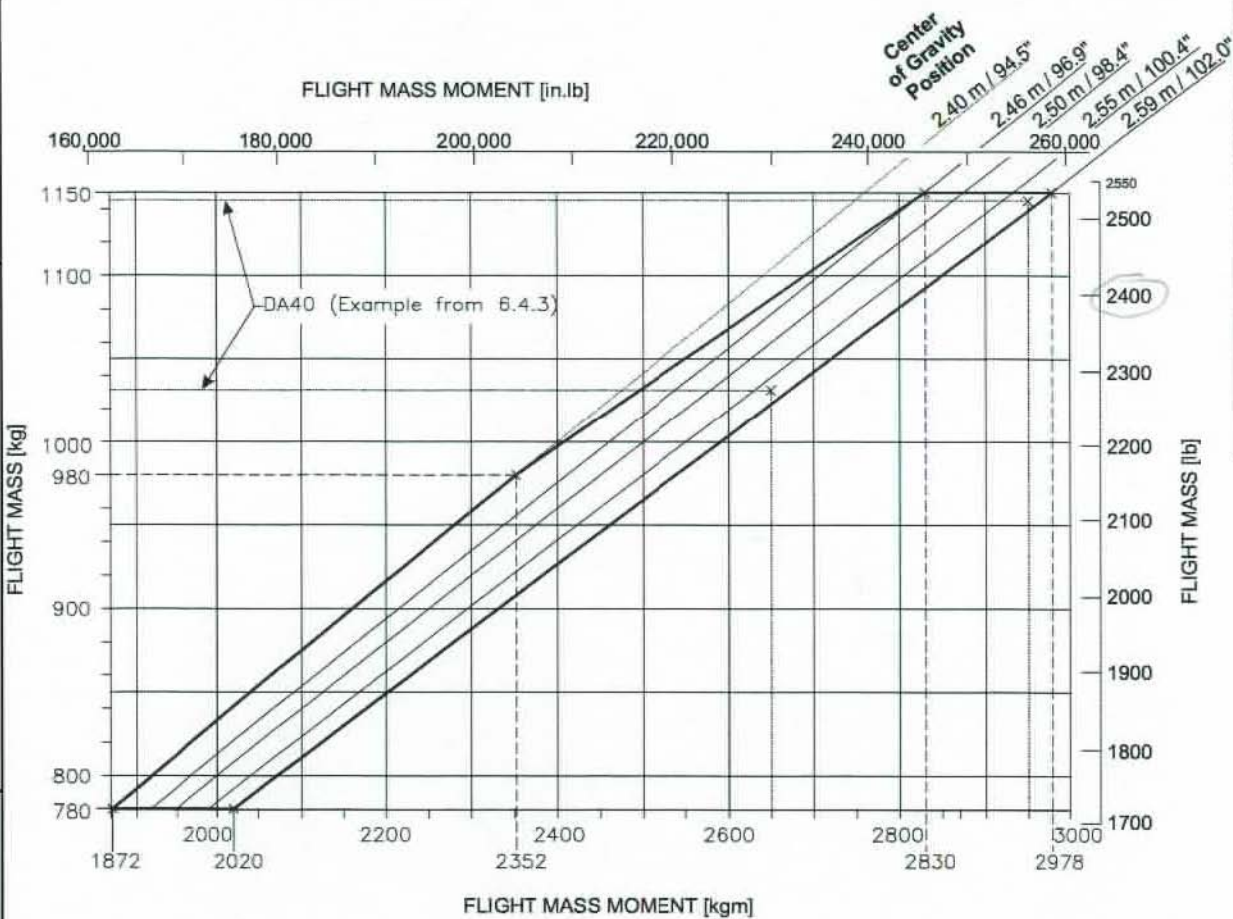
Most rearward flight CG:

2.59 m (102.0 in) aft of Datum Plane

6.4.5 PERMISSIBLE MOMENT RANGE

The diagram is replaced by the following:





6.4.5 PERMISSIBLE MOMENT RANGE

6.5 EQUIPMENT LIST AND EQUIPMENT INVENTORY

All equipment that is approved for installation in the DA 40 is shown in the *Equipment List* below.

The items of equipment installed in your particular airplane are indicated in the appropriate column. The set of items marked as 'installed' constitutes the *Equipment Inventory*.

Airplane Serial No.: 40646		Registration: N566DS		Date: FEB 23, 2000	
Description	Type	Part No.	Manufacturer	S/N	in- stalled
AVIONICS COOLING					
Cooling fan	Cyclone 21-3 Port	CRB122253	Lone Star Aviation		
COMMUNICATION					
COMM 1 antenna	DMC63-1/A		DM	N/A	X
COMM 2 antenna	DMC63-2		DM	N/A	X
COMM #1 ¹	KX125	069-01028-1101	Bendix/King		
COMM #1 ¹	KX 155A	069-01032-0201	Bendix/King		
COMM #1 ¹	KX 165A	069-01033-0101	Bendix/King		
COMM #2	KX 155A	069-01032-0201	Bendix/King		
Audio Panel / Marker / ICS ¹	KMA 28	066-01176-0101	Bendix/King		
ICS ²	PM1000 II	11922	PS Engineering		
Headset, pilot	Echelon 100		Telex		
Headset, co-pilot	Echelon 100		Telex		
Headset, LH pax	Echelon 100		Telex		
Headset, RH pax	Echelon 100		Telex		
Speaker	FRS8 /4 Ohms		Visaton	N/A	X
Handmic	100TRA	62800-001	Telex	N/A	X
AUTOPILOT SYSTEM:					
KAP 140	KAP 140		Bendix/King	N/A	X
Flight computer	KC 140	065-00176-5402	Bendix/King		
Pitch servo	KS 270 C	065-00178-2500	Bendix/King	8771	X
Pitch servo mount	KM 275	065-00030-0000	Bendix/King	N/A	X
Roll servo	KS 271 C	065-00179-0300	Bendix/King	11567	X
Roll servo mount	KM 275	065-00030-0000	Bendix/King	N/A	X
Trim servo	KS 272 C	065-00180-3500	Bendix/King	8712	X
Trim servo mount	KM 277	065-00041-0000	Bendix/King	N/A	X
Configuration module	KCM 100	071-00073-5000	Bendix/King	14917	X
Sonalert	SC	SC 628	Mallory	N/A	X
Control stick		DA4-2213-12-90	DAI	N/A	X
CWS switch		031-00514-0000	Bendix/King	N/A	X
AP-Disc switch		031-00428-0000	Bendix/King	N/A	X
Trim switch assy.		200-09187-0000	Bendix/King	N/A	X
Flight computer	KC140	065-00176-7703	Bendix/King	9771	X

6.5 EQUIPMENT LIST AND EQUIPMENT INVENTORY

The following items must be added to the Equipment List:

Airplane Serial No.: <i>40.646</i>		Registration: <i>N526DS</i>		Date: <i>FEB 23, 2006</i>	
Description	Type	Part No.:	Manufacturer	S/N	in- stalled
EQUIPMENT					
Winter Baffle		DAI-2157-00-00	DAI	<i>N/A</i>	X

Airplane Serial No.: 40646		Registration: N566DS		Date: FEB 23 2006	
Description	Type	Part No.	Manufacturer	S/N	in- stalled
ELECTRICAL POWER					
Battery	CB24-11M (G243)		Concorde (Gill)	N/A	X
Emergency Battery (28 pcs.)	MN 1500 AA		Duracell		
Ammeter	VM1000	4010050	Vision Microsyst.		
Ammeter current sensor	VM1000	3010022	Vision Microsyst.		
Voltmeter	VM1000	4010050	Vision Microsyst.		
Voltage regulator		VR2000-28-1 (D)	Electrosyst., Inc.		
External power connector			DAI	N/A	X
Alternator	ALU-8521LS	ALU-8521LS	Electrosyst., Inc.	N/A	X
EQUIPMENT					
Safety belts, pilot	5-01-1C0701	000918	Schroth	N/A	X
Safety belts, co-pilot	5-01-1C5701	000918	Schroth	N/A	X
Safety belts, RH pax	5-01-1B0701	000918	Schroth	N/A	X
Safety belts, LH pax	5-01-1B5701	000918	Schroth	N/A	X
ELT unit		E-01	ACK	54189	X
ELT remote switch		ED105	ACK	N/A	X
ELT antenna		ED109	ACK	N/A	X
FLIGHT CONTROLS					
Stall horn		DAI-9031-00-00	DAI	N/A	X
Flaps control unit (inst. panel)	430550		DAI	N/A	X
Flaps actuator assy	430555		DAI	N/A	X
SAFETY EQUIPMENT					
Fire extinguisher, portable		HAL1	AIR Total		
First aid kit					
FUEL					
Fuel qty indicator	VM1000	4010028	Vision Microsyst.		
Fuel qty sensor LH	VM1000	30100-11	Vision Microsyst.		
Fuel qty sensor RH	VM1000	30100-11	Vision Microsyst.		
HYDRAULIC					
Master cylinder		10-54A	Cleveland	N/A	X
Parking valve		60-5B	Cleveland	N/A	X
Brake assembly		30-239A	Cleveland	N/A	X
INDICATING / REC. SYSTEMS					
Digital chronometer	LC-2	AT420100	AstroTech		
Flight timer ³		85000-12	Hobbs	N/A	X
Flight timer ¹		85094-12	Hobbs		
Annunciator panel (system) ⁴			DAI		
Annunciator panel ⁴	WW-IDC 001		White Wire		

Airplane Serial No.:		Registration:		Date:	
40,146		N566DS		FEB 23 2006	
Description	Type	Part No.	Manufacturer	S/N	Installed
LIGHTS					
Map/Reading light assy crew		W1461.0.010	Rivoret	N/A	X
Cabin light		W1461.0.010	Rivoret	N/A	X
Instr./Radio lights dimmer assy		WW-LCM-002	White Wire		
Glareshield lamp assy		DA4-3311-10-01	DAI	N/A	X
Glareshield light inverter		APVL328-8-3-L-18QF	Quantaflux	N/A	X
Strobe/Pos. light assy LH	A600-PR-D-28	01-0790006-05	Wheelen	N/A	X
Strobe/Pos. light assy RH	A600-PG-D-28	01-0790006-07	Wheelen	N/A	X
Strobe light power supply LH/RH	A490ATS-CF-14/28	01-0770062-05	Wheelen	N/A	X
Taxi light	70346	01-0770346-05	Wheelen	N/A	X
Landing light	70346	01-0770346-03	Wheelen	N/A	X
Electroluminescent lamps	Quantaflux 1600		Quantaflux	N/A	X
NAVIGATION					
Pitot/Static probe, heated		DAI-9034-57-00	DAI	N/A	X
P/S probe HTR fail sensor		DA4-3031-01-00	DAI	N/A	X
Altimeter inHg/mbar, primary ⁵		5934PD-3	United In.		
Altimeter inHg/mbar, primary ⁵	LUN 1128	1128-14B6	Mikrotechna		
Altimeter inHg/mbar, secondary ⁵		5934PD-3	United In.	452007	X
Altimeter inHg/mbar, secondary ⁵	LUN 1128	1128-14B6	Mikrotechna		
Vertical speed indicator ⁶		7000	United In.		
Vertical speed indicator ⁶	LUN 1144	1144-A4B4	Mikrotechna		
Airspeed indicator ⁷		8025	United In.	191850	X
Airspeed indicator ⁷	LUN 1116	1116-B4B3	Mikrotechna		
Outside air temp. indic.		301F (C)	Davtron		
Magnetic compass		C2400L4P	Airpath		
Compass system C/O ⁸	KCS 55A		Bendix/King		
Slave gyro	KG 102 A	060-00015-0000	Bendix/King		
HSI	KI 525A	066-03046-0007	Bendix/King		
Slaving unit	KA 51B	071-01242-0001	Bendix/King		
Flux valve	KMT 112	071-01052-0000	Bendix/King		
Dir. gyro, free ⁸	AIM2051BLD	505-0031-931	BF-Goodrich		
Attitude indicator	AIM1100-28L(OF)	504-0111-936	BF-Goodrich		
Attitude indicator	AIM1100-28LK(OF)	504-0111-938	BF-Goodrich		
Turn coordinator w/o AP pickup	1394T100-(3Z)		Mid Continent Instr.		
Turn coordinator	1394T100-(12RZ)		Mid Continent Instr.		
Marker antenna	C1102		Comant	104228	X
DME	KN 62A	066-01068-0004	Bendix/King		
DME antenna	KA60	071-01174-0000	Bendix/King		
Transponder ⁹	KT 76A	066-1062-10	Bendix/King		
Transponder ⁹	KT 76C	066-01156-0101	Bendix/King		
XPDR antenna	KA60	071-01591-0001	Bendix/King		
Altitude digitizer		D120-P2-T	TCI		
ADF	KR87	066-01072-0004	Bendix/King		
ADF antenna	KA44B	071-01234-0000	Bendix/King		
ADF indicator	KI227	066-03063-0001	Bendix/King		

Airplane Serial No.: <i>40 646</i>		Registration: <i>N566DS</i>		Date: <i>FEB 23, 2006</i>	
Description	Type	Part No.	Manufacturer	S/N	in- stalled
NAV antenna coupler	CI505		Comant		
VOR/LOC/GS antenna		CI157P	Comant		
NAV/COM #1 ¹	KX125	069-01028-1101	Bendix/King		
NAV/COM #1 volt conv. ¹	KA39	071-01041-0001	Bendix/King		
NAV #1 ¹	KX155A	069-01032-0201	Bendix/King		
NAV #1 ¹	KX155A	069-01033-0101	Bendix/King		
NAV #2	KX155A	069-01032-0201	Bendix/King		
CDI, VOR/LOC #1 ⁴	KI 208	066-03056-0000	Bendix/King		
CDI, VOR/LOC #2	KI 208	066-03056-0000	Bendix/King		
GPS ¹⁰	KLN 89B	066-01148-0102	Bendix/King		
GPS ¹⁰	KLN 94	069-01034-0101	Bendix/King		
GPS antenna	KA 92	050-03318-0000	Bendix/King		
GPS/AP switch assy	MD41-528		Mid Continent		
ENGINE	IO-360 M1A		Textron Lycoming	<i>L-32677-51A</i>	X
ENGINE FUEL CONTROL					
Fuel flow transmitter	VM1000	3010032	Vision Microsyst.		
Fuel pressure transmitter	VM1000	3010017	Vision Microsyst.		
ENGINE IGNITION SYSTEM ¹¹					
SlickSTART booster	SS1001		Unison	<i>N/A</i>	X
Laser ignition controller	LC-1002-03	LC-1002-03	Unison		
Laser ignition harness	LH-1004-43		Unison		
Magneto RH/LH	4370/4347		Slick	<i>N/A</i>	X
Magneto RH/LH	4770/4771		Slick		
ENGINE INDICATING					
RPM sensor	VM1000	3010005	Vision Microsyst.		
Manifold pressure sensor	VM1000	3010016	Vision Microsyst.		
Cyl. head temp. probes (4 each)	VM1000	1020061	Vision Microsyst.		
EGT probes	VM1000	1020060	Vision Microsyst.		
Data processing unit ¹²	DPU	4010067	Vision Microsyst.		
Data processing unit ¹²	DPU	4010081	Vision Microsyst.		
Intergr. engine data display	VM1000	4010050	Vision Microsyst.		
I/O board assy. ¹²		3020003	Vision Microsyst.		
I/O board assy. ¹²		3020018	Vision Microsyst.		
ENGINE OIL					
Oil temperature sensor	VM1000	3010021	Vision Microsyst.		
Oil pressure transducer	VM1000	3010018	Vision Microsyst.		
ENGINE STARTING					
Starter	149-24LS		SKYTEC	<i>N/A</i>	X
<i>Airfield indicator</i>		<i>524-011-944</i>	<i>BF-Geotrich</i>	<i>25266</i>	X

Airplane Serial No.: <u>40 646</u>		Registration: <u>N566DS</u>		Date: <u>FEB 23, 2006</u>	
Description	Type	Part No.	Manufacturer	S/N	Installed
PROPELLER	MTV-12-B/180-17		mt-Propeller		
GOVERNOR	C-210776		Woodward	<u>N/A</u>	<u>X</u>
AIRPLANE FLIGHT MANUAL		Doc. No 6.01.01-E	DAI	<u>N/A</u>	<u>X</u>

Place: CF XUDate: FEB 23, 2006Signature: 

 DA
 Q 42

1. One of the following COM #1 units may be installed:
 - KX 125 (including KA 39) or KX 155A (OÄM 40-085) or KX 165A (OÄM 40-083).
2. One of the following intercoms may be installed:
 - KMA 28 (OÄM 40-067) or PM 1000 II.
3. One of the following flight timers may be installed:
 - Part No. 85094-12 (MÄM 40-029) or Part No. 85000-12.
4. One of the following Annunciator Panels may be installed:
 - DAI Annunciator Panel or White Wire WW-IDC-001(OÄM 40-060).
5. One of the following altimeters may be installed as primary altimeter:
 - United Instruments 5934PD-3 or Mikrotechna 1128-14B6.
 - One of the following altimeters may be installed as secondary altimeter:
 - United Instruments 5934PD-3 or Mikrotechna 1128-14B6.
6. One of the following vertical speed indicators may be installed:
 - United Instruments 7000 or Mikrotechna 1144-A4B4.
7. One of the following airspeed indicators may be installed:
 - United Instruments 8025 or Mikrotechna 1116-B4B3.

8. One of the following systems may be installed:
Compass System KCS 55A (OÄM 40-067) or
Directional Gyro AIM 205-1BLD (including CDI #1 KI 208; OÄM 40-055).
9. One of the following transponders may be installed:
KT 76A or KT 76C (OÄM 40-067).
10. One of the following GPS units may be installed:
KLN 89 or KLN 94 (VFR: OÄM 40-065; IFR: OÄM 40-067).
11. One of the following ignition systems may be installed:
SlickSTART booster with Slick 4370/4347 magnetos (OÄM 40-073) or
LASAR ignition controller & harness with Slick 4770/4771 magnetos.
12. One of the following combinations of DPU and I/O board assy. may be installed:
DPU 4010067 with I/O board assy. 3020003 or
DPU 4010081 with I/O board assy. 3020018 (MÄM 40-039/a)

A & M MAINTENANCE SERVICES, INC.
 134 S. CLOW INTERNATIONAL PARKWAY
 SUITE B
 BOLINGBROOK, ILLINOIS 60490
 630-759-8130



Weight & Balance Equipment List Revision

<i>Aircraft Make</i>	<i>Aircraft Model</i>
<i>Diamond Aircraft</i>	<i>DA-40-180</i>
<i>Serial No.</i>	<i>Aircraft Registration</i>
<i>40.646</i>	<i>N566DS</i>

Present Weight & Balance Data

<i>Old Date</i>	<i>Max. Gros</i>	<i>Useful Load</i>	<i>E.W.</i>	<i>C.G.</i>	<i>Moment</i>
<i>02/23/06</i>	<i>2535.00 lbs</i>	<i>784.00 lbs</i>	<i>1751.00 lbs</i>	<i>96.745 in</i>	<i>169400.08 in lbs</i>
<i>02/23/06</i>	<i>1150.00 kg</i>	<i>355.80 kg</i>	<i>794.20 kg</i>	<i>2.457 m</i>	<i>1951.70 m kg</i>

Removed Equipment

<i>Mfgr.</i>	<i>Description</i>	<i>Part/Serial #</i>	<i>Wt. Change</i>	<i>Arm</i>	<i>Moment</i>
<i>NONE</i>					

Installed Equipment

<i>Mfgr.</i>	<i>Description</i>	<i>Part/Serial #</i>	<i>Wt. Change</i>	<i>Arm</i>	<i>Moment</i>
	<i>OSB-40-057</i>				
	<i>Max Take-Off</i>				
	<i>Increase</i>				

<i>New Date</i>	<i>Max Gross</i>	<i>Useful Load</i>	<i>E.W.</i>	<i>C.G.</i>	<i>Moment</i>
<i>04/11/08</i>	<i>2646.00 lbs</i>	<i>895.00 lbs</i>	<i>1751.00 lbs</i>	<i>96.745 in</i>	<i>169400.08 in lbs</i>
<i>04/11/08</i>	<i>1200.00 kg</i>	<i>405.8 kg</i>	<i>794.20 kg</i>	<i>2.457 m</i>	<i>1951.70 m kg</i>

<i>Signature:</i> <i>Marli Agone</i>	<i>Date:</i> <i>04/11/2008</i>
Marilina Angarone AP 344688075	

Model: DA 40



Aircraft Serial No.:	40.646
Registration:	N566DS
Issue Date:	23.02.06

Aircraft Specific Weighting Report

Data with reference to the Type Certificate Data Sheet and the Flight Manual

Reference Plane: Vertical plane 2194 mm(86.38 in) in front of wing leading edge at root rib.
 Horizontal reference line: Wedge 600:31 (2.96°), 2910mm (114.57 in) aft of step in the cockpit rim.
 Equipment List dated: 23.02.06
 Cause for Weighing:

ORIGINAL

Weight and Balance Calculations

Weight Condition:

Include brake fluid, engine oil and unusable fuel (2 liters/0.5 US gal on each side).

Empty Weight:

Finding Arm: (Measured)

Support	Gross		Tare		Net Weight	
	kg	lbs	kg	lbs	kg	lbs
Front, G ₂	121.1	267.0	0	0	121.1	267.0
Main, G _{1LH}	333.4	735.0	0	0	333.4	735.0
Main, G _{1RH}	339.7	749.0	0	0	339.7	749.0
EMPTY WEIGHT, G					794.2	1751.0

	Lever Arm	
	mm	in.
X ₂	933	36.73
X _{1LH}	2724	107.24
X _{1RH}	2739	107.83

Empty Weight Centre of Gravity, X_{CG}:

Empty Weight CG Formula

$$X_{CG} = \frac{(G_{1LH} \cdot X_{1LH} + G_{1RH} \cdot X_{1RH}) + (G_2 \cdot X_2)}{G_2 + G_{1LH} + G_{1RH}} = \begin{matrix} 2.457 \text{ m} \\ 96.745 \text{ in.} \end{matrix}$$

Empty Weight Moment, M:

Empty Weight Moment Formula

$$M = G \times X_{CG} = \begin{matrix} 1951.70 \text{ kg}\cdot\text{m} \\ 169400.08 \text{ in}\cdot\text{lbs} \end{matrix}$$

Maximum Permitted Useful Load:

Maximum Ramp Weight	1150.0 kg	2535 lbs
Empty Weight	794.2 kg	1751.0 lbs
Maximum Useful Load	355.8 kg	784.0 lbs

Data to be entered into the Flight Manual:

Empty Weight, G:	Empty Weight Moment, M:
794.2 kg	1951.70 kg·m
1751.0 lbs	169400.08 in·lbs

Place: D.A.I.C. - CYXU Date: 23.02.06	Authorizing Stamp DA Q 42	Authorizing Signature
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CHAPTER 7

DESCRIPTION OF THE AIRPLANE AND ITS SYSTEMS

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7.1 INTRODUCTION

Chapter 7 contains a description of the airplane and its systems, together with operating instructions.

For details about optional equipment see Chapter 9.

7.2 AIRFRAME

Fuselage

The GFRP fuselage is of semi monocoque molded construction. The fire protection on the firewall is of a special fire-resistant matting, which is covered on the engine side by stainless steel cladding. The two main bulkheads are GFRP/CFRP items.

Wings

The wings have a front and rear spar; each wing has a top shell and a bottom shell - a 'fail-safe' concept. The wings, as well as the aileron and flaps, are made of GFRP/CFRP, and are principally of sandwich construction. An aluminum fuel tank is installed in each of the wings.

Empennage

The airplane has a 'T' tail of GFRP semi monocoque construction. Both the stabilizers have twin spars and a skin with no sandwich. Rudder and elevator are of sandwich construction.

7.3 FLIGHT CONTROLS

The ailerons, elevator and wing flaps are operated through control rods, while the rudder is controlled by cable. The flaps are electrically operated. Elevator forces can be balanced by a trim tab on the elevator, which is operated by a Bowden cable.

Ailerons

Construction: GFRP/CFRP composite sandwich

Hinges: There are 4 hinges, which are hinge pins mounted in an aluminum bracket. They are secured in position by a roll pin. The absence of this roll pin can lead to the loss of the hinge pin and a consequent loss of flight safety.

Operation: A rod-end bearing is screwed into a steel push rod and locked by means of a nut which has locking varnish applied to it. Damage to this varnish can indicate a twisting and thus a change to the adjustment. The connection between the rod-end bearing and the control horn is a bolt, the nut of which is likewise sealed with locking varnish.

The aluminum control horn is attached to the aileron with 3 screws.

Flaps

Construction: GFRP/CFRP composite sandwich

Hinges: There are 6 hinges, which are hinge pins mounted in an aluminum bracket. They are secured in position by a roll pin. The absence of this roll pin can lead to the loss of the hinge pin and a consequent loss of flight safety. Another aluminum fitting is located at the fuselage and is attached to a torsion tube. The torsion tube is located in the fuselage, creating a connection between the left and right flaps.

Operation: A rod-end bearing is screwed into a steel push rod and locked by means of a nut which has locking varnish applied to it. Damage to this varnish can indicate a twisting and thus a change to the adjustment. The connection between the rod-end bearing and the control horn is a bolt, the nut of which is likewise sealed with locking varnish.

The flap control horn is attached to the flap with 3 screws.

The flaps are driven by an electric motor and have 3 settings:

- Cruise (UP), totally retracted
- Take-off (T/O), and
- Landing (LDG).

The flaps are operated by means of a 3-position flap selector switch on the instrument panel. The positions of the switch correspond to the positions of the flaps, the Cruise position of the switch being at the top. If the switch is moved to another position, the flaps continue to travel automatically until they have reached the position selected on the switch. The UP and LDG positions are additionally protected by a limit switch to guard against over-running the end positions.

The electrical flap drive has an automatic circuit breaker which can also be operated manually.

Flap position indicator:

The current flap position is indicated by means of three lights beside the flap selector switch.

When the upper light (green) is illuminated, the flaps are in the Cruise position (UP); when the center light (white) is illuminated, the flaps are in Take-off position (T/O); when the lower light (white) is illuminated, the flaps are in Landing position (LDG).

When two lights are illuminated simultaneously, the flaps are between the two indicated positions. This is the case only when the flaps are traveling.

Elevator

Construction: GFRP sandwich

Hinges: 5 hinges

Operation: Steel push-rods;

Two of the bellcrank bearings are accessible to visual inspection next to the lower hinge of the rudder. The elevator horn and its bearing, as well as the connection to the push-rod, can be visually inspected at the upper end of the rudder.

Rudder

Construction: GFRP sandwich

Hinges: Upper hinge: One bolt.

Lower hinge: Bearing bracket including rudder stops, held by 4 screws to the rear web of the vertical stabilizer. The mating part on the rudder is a bracket which is attached to the rudder by 2 bolts. The bolts and nuts are accessible to visual inspection.

Operation: Steel cables, the eyes of which are connected to the bolts on the bracket.

Elevator Trim

The trim control is a black wheel in the center console to the rear of the engine controls. To guard against over-rotating, the trim wheel incorporates a friction device. A mark on the wheel shows the take-off (T/O) position.

Turn wheel to the front = nose down

Turn wheel to the rear = nose up

Pedal adjustment**NOTE**

The pedals may only be adjusted on the ground!

The pedals are unlocked by pulling the black handle which is located behind the rear attachment.

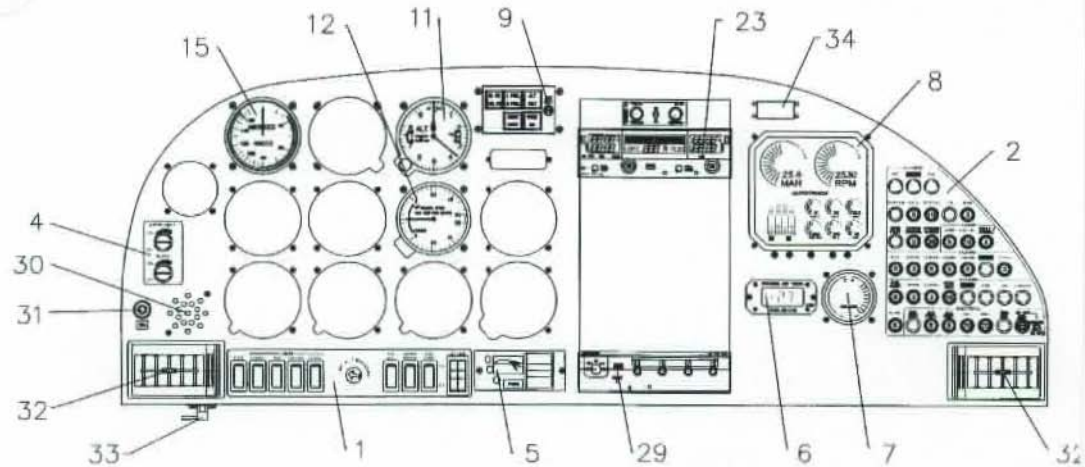
Forward adjustment:

Whilst keeping the handle pulled, push the pedals forward with your feet. Release the handle and allow the pedals to lock into place.

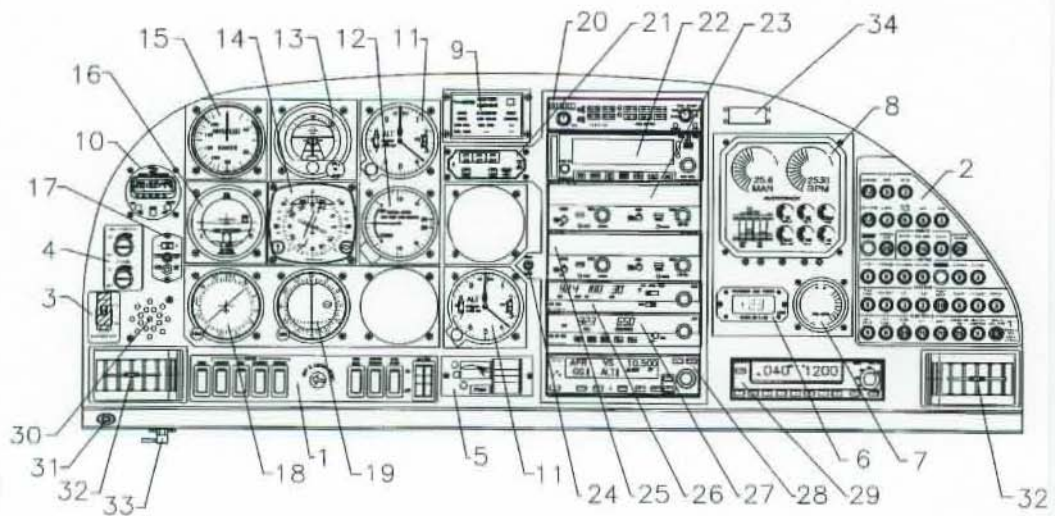
Rearward adjustment:

Using the unlocking handle, pull the pedals back to the desired position. Release the handle and push the pedals forward with your feet until they lock into place.

7.4 INSTRUMENT PANEL



VFR instrument panel



IFR instrument panel

Major instruments and controls	
--------------------------------	--

1 Electrical switches, ignition switch	18 ADF indicator
2 Circuit breakers*	19 Course deviation indicator (CDI)
3 Emergency switch	20 Audio amplifier / intercom / marker beacon receiver
4 Rotary buttons for instrument lighting and flood light	21 GPS Annunciation Control Unit
5 Flap selector switch	22 GPS
6 OAT indicator	23 Radio / VOR, No. 1
7 Fuel quantity indicator	24 Remote DME switch
8 Engine instruments	25 Radio / VOR, No. 2
9 Lights (Annunciator Panel)	26 DME
10 Chronometer	27 ADF receiver
11 Altimeter	28 Autopilot control unit (optional)
12 Vertical speed indicator (VSI)	29 Transponder
13 Attitude gyro (artificial horizon)	30 Stall warning horn
14 Horizontal situation indicator (HSI)	31 Microphone socket
15 Airspeed indicator	32 Ventilation nozzles
16 Turn & bank indicator	33 Alternate Static Valve (optional for VFR version)
17 Slaving meter	34 ELT operating unit (RCPI)

*) Designations and abbreviations used to identify the circuit breakers are explained in Section 1.5 DEFINITIONS AND ABBREVIATIONS.

Cockpit ventilation

Ventilation in the front is provided by the movable ventilation nozzles (17) in the instrument panel. Furthermore there are spherical nozzles in the roll bar on the left and right side next to the front seats as well as on the central console above the passengers' heads. The spherical nozzles are opened and closed by twisting.

7.5 LANDING GEAR

The landing gear consists of a main landing gear of sprung steel struts, and a free-castering nose wheel which is sprung by an elastomer package.

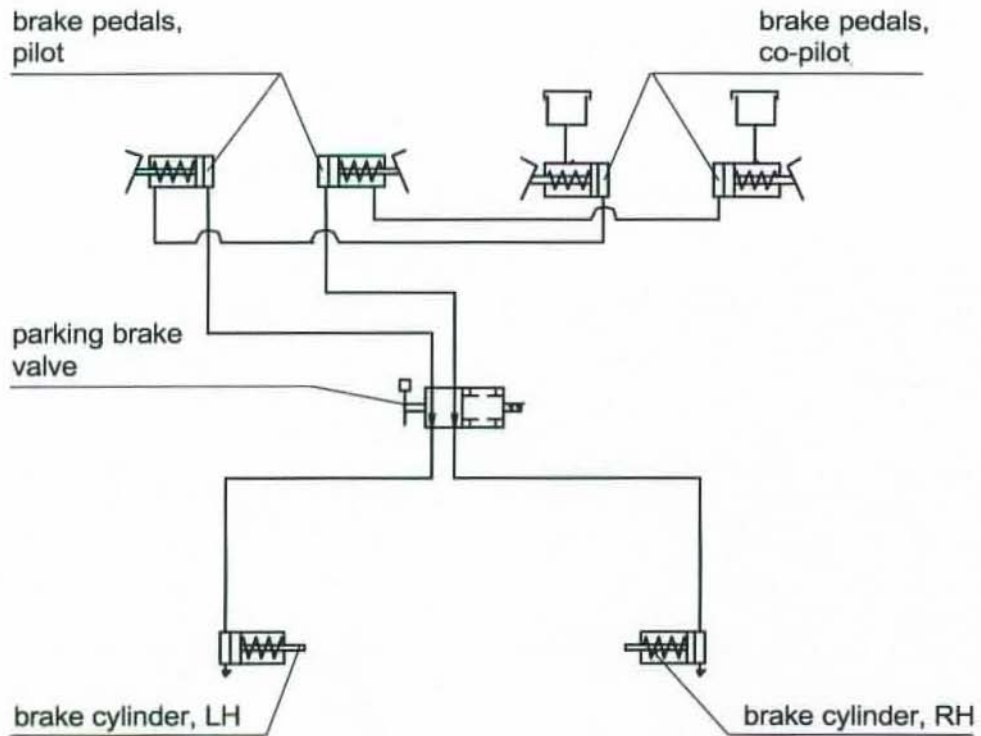
The wheel fairings are removable. When flying without wheel fairings, it should be noted that there is a reduction in some areas of performance (see Chapter 5).

Wheel brakes

Hydraulically operating disk brakes act on the wheels of the main landing gear. The wheel brakes are individually operated by means of toe pedals.

Parking brake

The lever is located on the small center console under the instrument panel, and is in the upper position when the brakes are released. To operate the parking brake pull the lever downwards until it catches. Brake pressure is built up by multiple operation of the toe brake pedals, and is maintained until the parking brake is released. To release, the lever is pushed upwards.

Hydraulic system schematic

7.6 SEATS AND SAFETY HARNESSSES

To increase passive safety, the seats are constructed using a carbon fiber/Kevlar hybrid material and GFRP. The seats are removable to facilitate the maintenance and inspection of the underlying controls. Covers on the control sticks prevent loose objects from falling into the area of the controls.

The seats have removable furnishings and are equipped with energy-absorbing foam elements.

The seats are fitted with three-part safety harnesses. The harnesses are fastened by inserting the end of the belts in the belt lock, and are opened by pressing the red release on the belt lock.

The backs of the rear seats can be laid forward after pulling upwards on the knob of the locking bolt.

7.7 BAGGAGE COMPARTMENT

The baggage compartment is behind the seat backs of the rear seats. Without a baggage net, no baggage may be loaded.

7.8 CANOPY, REAR DOOR, AND CABIN INTERIOR

Front canopy

The front canopy is closed by pulling down on the canopy frame, following which it is locked by means of a handle on the left hand side of the frame. On locking, steel bolts lock into mating holes in polyethylene blocks.

"Cooling Gap" position: A second setting allows the bolts to lock in, leaving a gap under the forward canopy.

- The canopy can be blocked by a locking device (optional) on the left side near the canopy opening lever by turning the key clockwise. The closed and blocked canopy can be
- opened from inside by pulling the lever inside the opening handle.

WARNING

The airplane may be operated with the front canopy in the "cooling gap" position on the ground only. Before take-off the front canopy must be completely closed and locked, but not blocked with the locking device.

- A window on the left hand side of the canopy can be opened for additional ventilation
- or as an emergency window. Some serial numbers have another window on the right
- hand side of the canopy.

7.9 POWER PLANT

7.9.1 ENGINE, GENERAL

Lycoming IO-360-M1A: Air-cooled four-cylinder four-stroke engine. Horizontally-opposed direct-drive engine with fuel injection and underslung exhaust.

Displacement: 5916 cm³ (361 in³)

Max. power: 180 HP (134.2 kW) at 2700 RPM at sea level and ISA

Max. continuous power: 160 HP (119.3 kW) at 2400 RPM at sea level and ISA

The principal engine accessories at the front of the engine are the propeller governor, the starter motor, and the alternator. The ignition (optionally controlled by an electronic control unit), the twin magneto system and the mechanical fuel pump are at the rear of the engine. Fuel is supplied via a fuel injection system.

Further information should be obtained from the engine operating manual.

The engine instruments are on the right hand side of the instrument panel.

The ignition switch is designed as a key-operated lock. The ignition is switched on by moving the switch to the right from the OFF position to the L-R-BOTH positions. A further turn to the right to the START position will operate the starter motor.

7.9.2 OPERATING CONTROLS

The engine performance is controlled by means of three levers: throttle, RPM lever and mixture control lever, situated together as a group on the large center console (also referred to as the throttle quadrant). 'Front' and 'rear' are defined in relation to the direction of flight.

Throttle

- left hand lever with large, black knob

This lever is used to set the manifold pressure (MP). When the throttle is furthest forward, the engine is being provided with extra fuel for high performance settings.

Lever forward (MAX PWR) = Full throttle, higher MP

Lever to rear (IDLE) = Idle, low MP

High manifold pressure means that a large quantity of fuel-air mixture is being supplied to the engine, while low manifold pressure means a lesser quantity of fuel-air mixture is being supplied.

RPM lever

- central lever with blue handle

Lever forward (HIGH RPM) = High RPM, fine pitch

Lever to rear (LOW RPM) = Low RPM, coarse pitch

By means of this lever the propeller governor controls the propeller pitch and thus engine RPM (= propeller RPM). A selected RPM is held constant by the governor independent of the airspeed and the throttle setting ('Constant Speed').

The propeller governor is flanged onto the front of the engine. It regulates the supply of engine oil to the propeller. The propeller governor oil circulation is an integral part of the engine oil circulation system. Following a defect in governor or oil system, the blades go the finest possible pitch (maximum RPM), thus allowing continuation of the flight.

CAUTION

Following failure of the governor or a serious drop in oil pressure, the RPM should be adjusted using the throttle. Every effort should be made not to exceed 2700 RPM.

CAUTION

The throttle and RPM lever should be moved slowly, in order to avoid over-speeding and excessively rapid RPM changes. The light wooden propeller blades produce more rapid RPM changes than metal blades.

Mixture control lever

- right hand lever with red handle and lock to avoid inadvertent operation.

This lever is used to set the proportions in the fuel-air mixture which is supplied to the engine.

Lever forward (RICH) = Mixture rich (in fuel)

Lever to rear (LEAN) = Mixture lean (in fuel)

If the lever is at the forward stop, extra fuel is being supplied to the engine which at higher performance settings contributes to engine cooling.

In cruise, the mixture should be made leaner in order to reach the appropriate fuel-air mixture. The leaning procedure is given in Chapter 4.

To shut off the engine the mixture control lever is pulled to the rear stop. Air without fuel is thus drawn into the cylinders and the engine dies. When the engine is stationary there is thus no fuel in the cylinders.

Alternate Air

In the event of the loss of manifold pressure because of icing or blocking of the air filter, there is the possibility of drawing air from the engine compartment. The operating lever for Alternate Air is located under the instrument panel to the left of the center console. To open Alternate Air the lever is pulled to the rear. Normally, Alternate Air is closed, with the lever in the forward position.

Placard on the lever, forward position:

ALTERNATE AIR

Placard on the lever, visible when lever is in the rearward position:

**ALTERNATE AIR
ON**

7.9.3 PROPELLER

An mt-Propeller MTV 12 B/180-17 hydraulically-regulated 3-bladed constant speed propeller is installed. It has wood-composite blades with fiber-reinforced plastic coating and stainless steel edge cladding; in the region of the propeller hub the leading edge is coated with adhesive PU foil. These blades combine the lowest weight whilst minimizing the chance of vibration.

CAUTION

Operation on the ground at high RPM should be avoided as far as possible, as the blades could suffer stone damage. For this reason a suitable site for engine runs (magneto and propeller checks) should be selected, where there are no loose stones or similar items.

WARNING

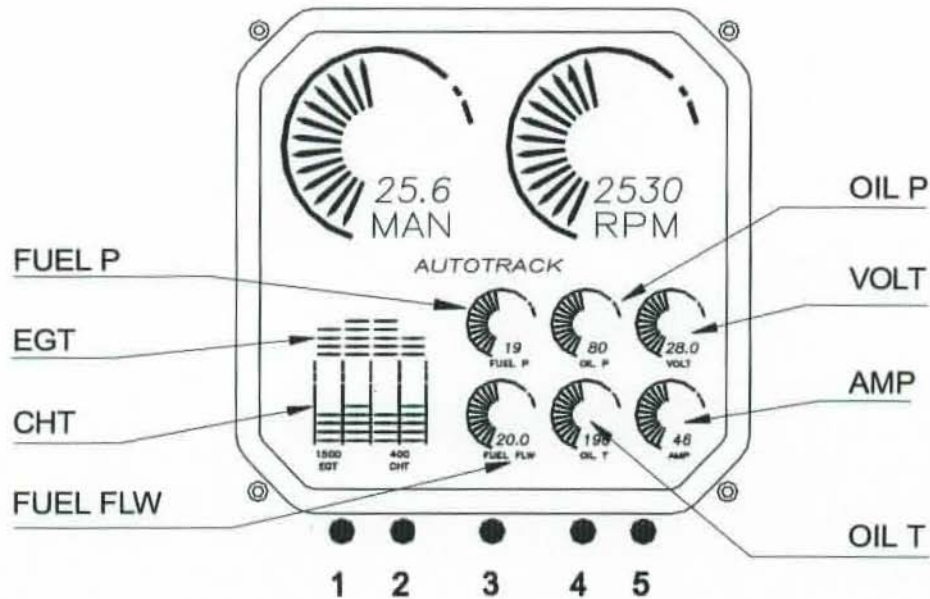
Never move the propeller by hand while the ignition is switched ON, as it may result in serious personal injury.

Never try to start the engine by hand.

Governor

A Woodward C-210776 governor is installed.

7.9.4 ENGINE INSTRUMENTS



Button 1: 'Lean' mode

Button 2: Digital exhaust gas / cylinder head temperature mode

Button 3: Switch in autotrack

Button 4: Fuel computer mode

Button 5: Engine data recorder

Button 3 has an additional function on switch-on: Display mode

Sweep mode or pointer mode

If on switch-on button 3 is kept pressed until the display transfers from activating all bars/pointers to indicating the actual values, the type of presentation can be selected. In the one case the circular instruments show the values with a pointer as in conventional analog instruments, whilst in the other case the circular instruments fill with pointers/bars up to the current value. It remains for the pilot to select his preferred presentation.

Indications on the Vision Microsystems VM 1000 Engine Instrument

Designation	Indication	Unit
MAN	Manifold pressure	inHg
RPM	RPM	RPM
EGT	Exhaust gas temperature	°F
CHT	Cylinder head temperature	°F
FUEL P	Fuel pressure	psi
FUEL FLW	Fuel flow	US gal/hr
OIL P	Oil pressure	psi
OIL T	Oil temperature	°F
VOLT	Voltage	V
AMP	Intensity of current	A

Button 1 - Lean mode

Upon powering up the unit the normal mode is shown. Between the colored sector markings the cylinder head temperatures of the individual cylinders are shown by bars. Above those are bars showing the exhaust gas temperatures of the individual cylinders.

In the event of the failure of a sensor the relevant indication remains empty. A flashing cylinder head temperature indication means either that the cylinder is too hot, or that it is being cooled too rapidly (shock-cooling).

The operation of button 1 causes the display to move to 'lean' mode. This is confirmed by 2 half-bars appearing to the left and right of the bar blocks. In this mode all bars which previously showed cylinder head and exhaust gas temperature are used for exhaust gas temperature only. One bar represents 10 °F. If the columns are completely filled with bars before the mixture is lean, button 1 should be pressed twice so that the bars start again at the base of the indicator.

A flashing bar column indicates that the relevant cylinder has reached the hottest exhaust gas temperature. This point will be marked with a single bar, which can be used as a reference for enriching the mixture. As an option, the numerical indication can be used additionally for this purpose.

Button 2: Digital exhaust gas / cylinder head temperature mode

Using this button, the numerical indication for exhaust gas and cylinder head temperature underneath the graphical representation of these figures is set. Following each sequential operation of the button the exhaust gas and cylinder head temperatures of an individual cylinder are displayed. In this, the display jumps automatically from the number of the current cylinder to its current temperature. After the fourth cylinder the display goes into the automatic mode, which gives both the number of the cylinder with the highest exhaust gas temperature as well as (beside it) the number of the hottest cylinder. Alternating with this, the associated temperatures are displayed.

Button 3: Switch in autotrack

In the autotrack mode changes in the engine values are shown. If button 3 is operated in flight, variations from the current values will be displayed, in that the relevant circular instrument and the annotation AUTOTRACK will start to flash.

In order to leave the mode, button 3 must be operated. The mode is left automatically if there is a critical value to be indicated.

Button 4 - Fuel computer mode

By operating button 4 the display is switched from fuel flow (FUEL FLW) to a numerical indication underneath it. There are 4 modes, which are called up by pressing button 4 in sequence. The modes are:

REM: The remaining fuel is shown in US gal. The steps in this are 0.1 US gal. This mode is only available if the "ADD" mode - add up fuel - has previously been activated.

HRS: This mode shows the remaining flight time (in hours) on the basis of the current fuel flow. The steps in this indication are tenths of hours. This mode is also only available if the "ADD" mode - add up fuel - has previously been activated.

BRN: This mode shows the amount of fuel used (in US gal) since the equipment was switched on. The steps in this are 0.1 US gal.

ADD: This mode can be used after refueling to bring the fuel quantity, which the equipment uses for its calculations, up to date. In order to utilize the REM and HRS modes, the computer needs to be told how much fuel has been taken on. 10 US gal are added by pressing button 3, while pressing button 5 adds one US gal to the total. The quantity is confirmed by pressing button 4. In doing this, the quantity which has been entered in ADD is added to the previous total under REM. To check the fuel quantity button 4 should be pressed until REM is shown.

If too much has been added, button 4 should not be pressed for confirmation. After approx. 20 seconds the computer automatically leaves the ADD mode.

CAUTION

Incorrect use of the computer in the fuel-computer mode will result in false statements in the "REM - remaining fuel" and the "HRS - remaining flight time" modes. Before using the fuel computer mode in flight the pilot must be certain that he has understood the operation and use of the equipment. Beyond this, use of the fuel computer must not be regarded as a substitute for fuel planning for a flight.

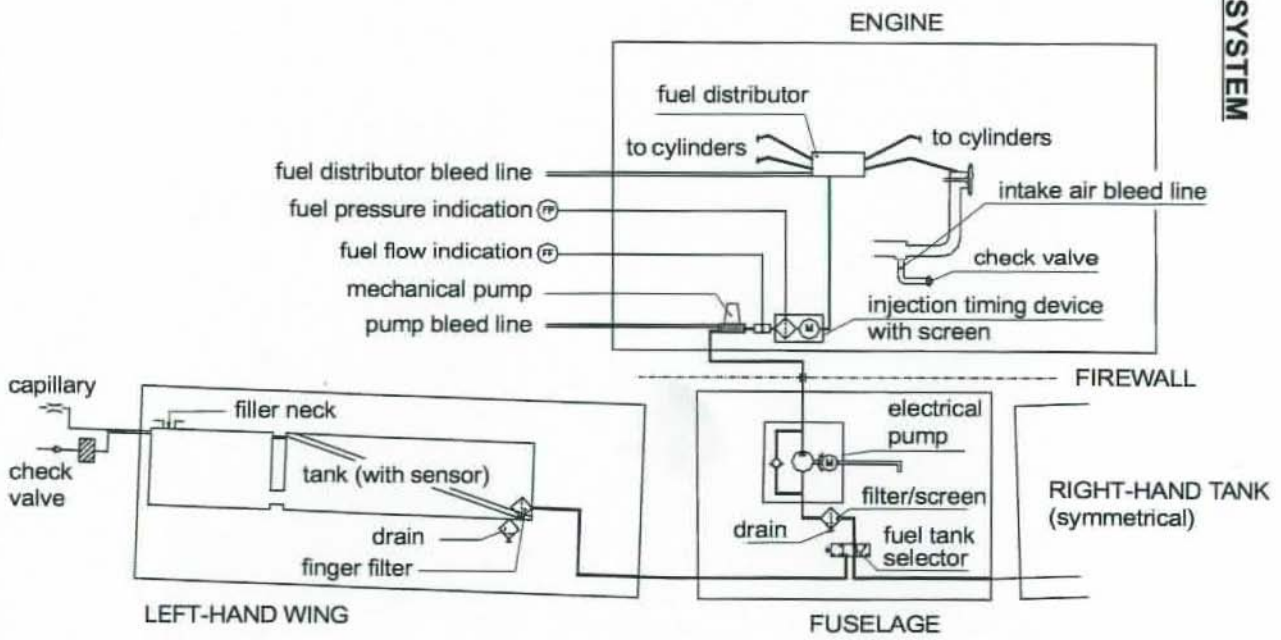
Button 5 - Engine Data Recorder

Operating button 5 will activate the engine data recorder. The digital values shown are the minimum values recorded by the engine instrument unit during operation, such as lowest voltage, lowest fuel pressure, etc. The numerical RPM indicator will indicate the total operating hours.

Pressing button 5 again will show the maximum values encountered. Pressing button 5 still another time will turn off the engine data recorder and the display will return to the original mode. If button 5 is not pressed for approximately 20 seconds, the display will automatically return to the original mode.

Data of the engine data recorder can be called during or immediately after flight only. With each new flight the old data will be overwritten.

7.10 FUEL SYSTEM



Fuel pumps

The fuel system is equipped with a mechanical and an electrical fuel pump. The mechanical pump provides for the normal fuel supply.

The electrical fuel pump is provided as an auxiliary and emergency pump, which does not operate under normal circumstances. It is operated with the FUEL PUMP switch on the row of switches on the instrument panel. It is checked during engine start, and is used as a safety back-up during take-off and landing, as well as when switching fuel tanks. It is also switched on for safety in the event of a decrease in fuel pressure.

Fuel tank selector

The fuel tank selector is situated on the center console. Its positions are LEFT (tank), RIGHT (tank) and OFF. The OFF position is reached by turning the selector to the right while pulling up the safety catch of the fuel tank selector. This is to ensure that an OFF selection is not made unintentionally.



Fuel tanks

Each of the two wing tanks consists of two aluminum chambers which are joined by a piece of flexible hose and two independent vent hoses. There are two separate vents per tank. The hose terminations are situated on the underside of the wing, approx. 2 meters (7 ft) from the wing tip. One vent acts as a capillary, both to equalize the air pressure, and to provide a safety factor in the event of a failure of the other vent. The second vent is a check valve, to allow air to enter the tank, but prevent flow to the outside.

A coarse filter (finger filter) is fitted before the outlet. To allow draining of the tank, there is an outlet valve at its lowest point. A gascolator sits at the lowest point in the fuel system. A drain valve is fitted to the gascolator, which can be used to remove water and sediment which has collected in the fuel system. This valve is fitted centrally on the underside of the fuselage, approximately 30 cm (1 ft) forward of the wing leading edge.

A capacity probe ascertains fuel quantity in the tank. When the fuel quantity indicator reads zero, only the unusable fuel remains in the tank. The total capacity of each tank is 20 US gal, the maximum quantity that can be indicated is 15 US gal. Up to an actual quantity of 15 US gal the indication is correct. At an actual quantity above 15 US gal the indication remains at 15 US gal.

NOTE

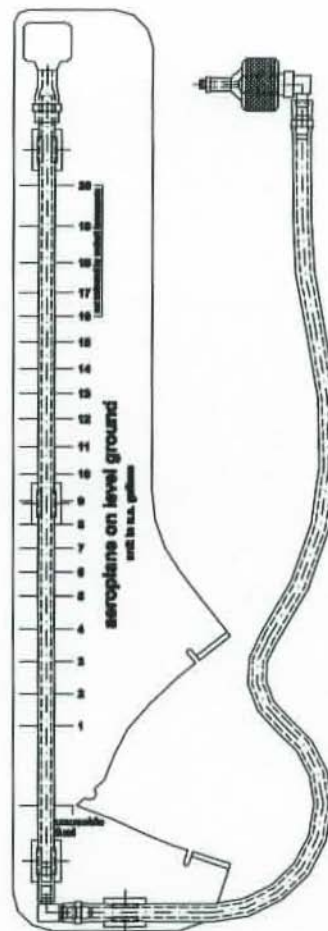
When the fuel quantity indicator reads 15 US gal, the correct fuel quantity must be determined with the fuel quantity measuring device. If this measurement is not carried out, the fuel quantity available for flight planning is 15 US gal.

Fuel quantity measuring device

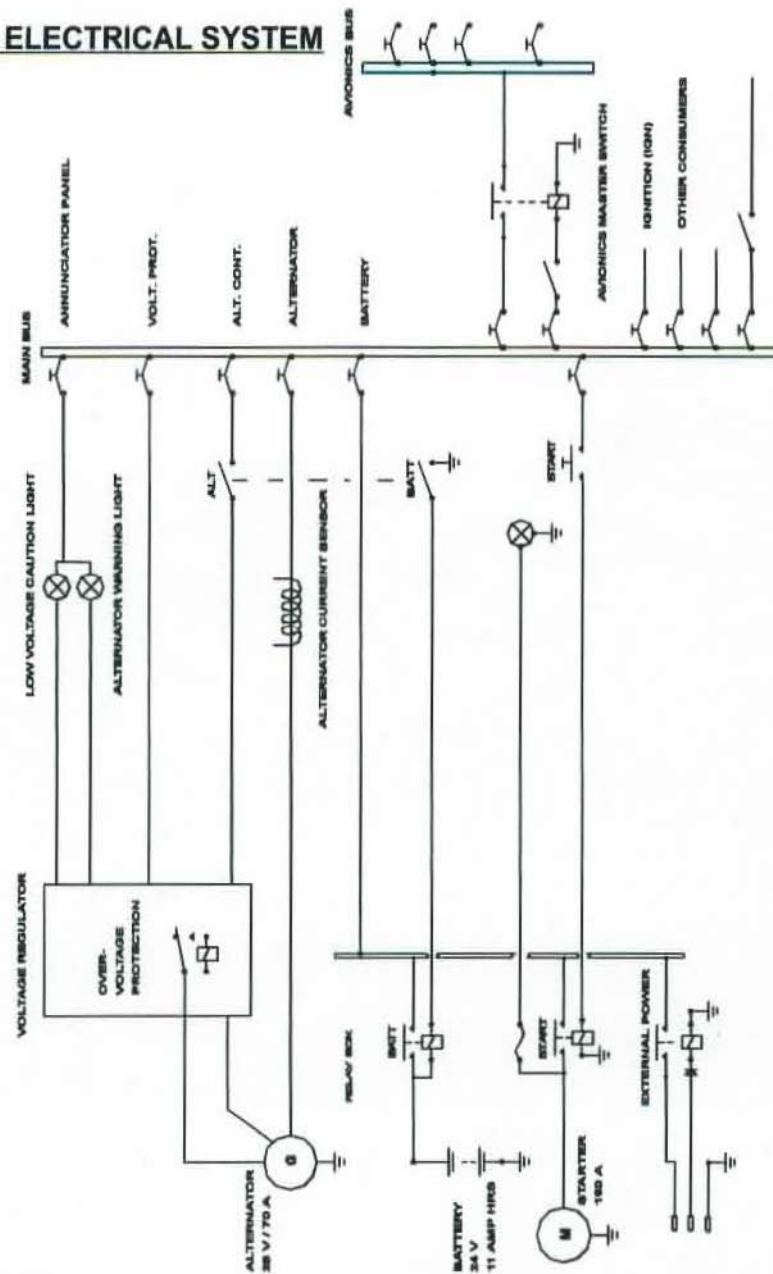
The fuel quantity measuring device allows the fuel quantity in the tank to be determined during the pre-flight inspection. It functions according to the principle of communicating containers. The fuel quantity measuring device has a recess which fits the airfoil of the wing. With this recess the device is held against the stall strip at the leading edge of the wing. The exact position is marked by a bore in the stall strip. Then the metal connector is pressed against the drain of the tank. The amount of fuel in the tank can now be read off from the vertical ascending pipe.

For an exact indication the airplane must stand on a horizontal ground.

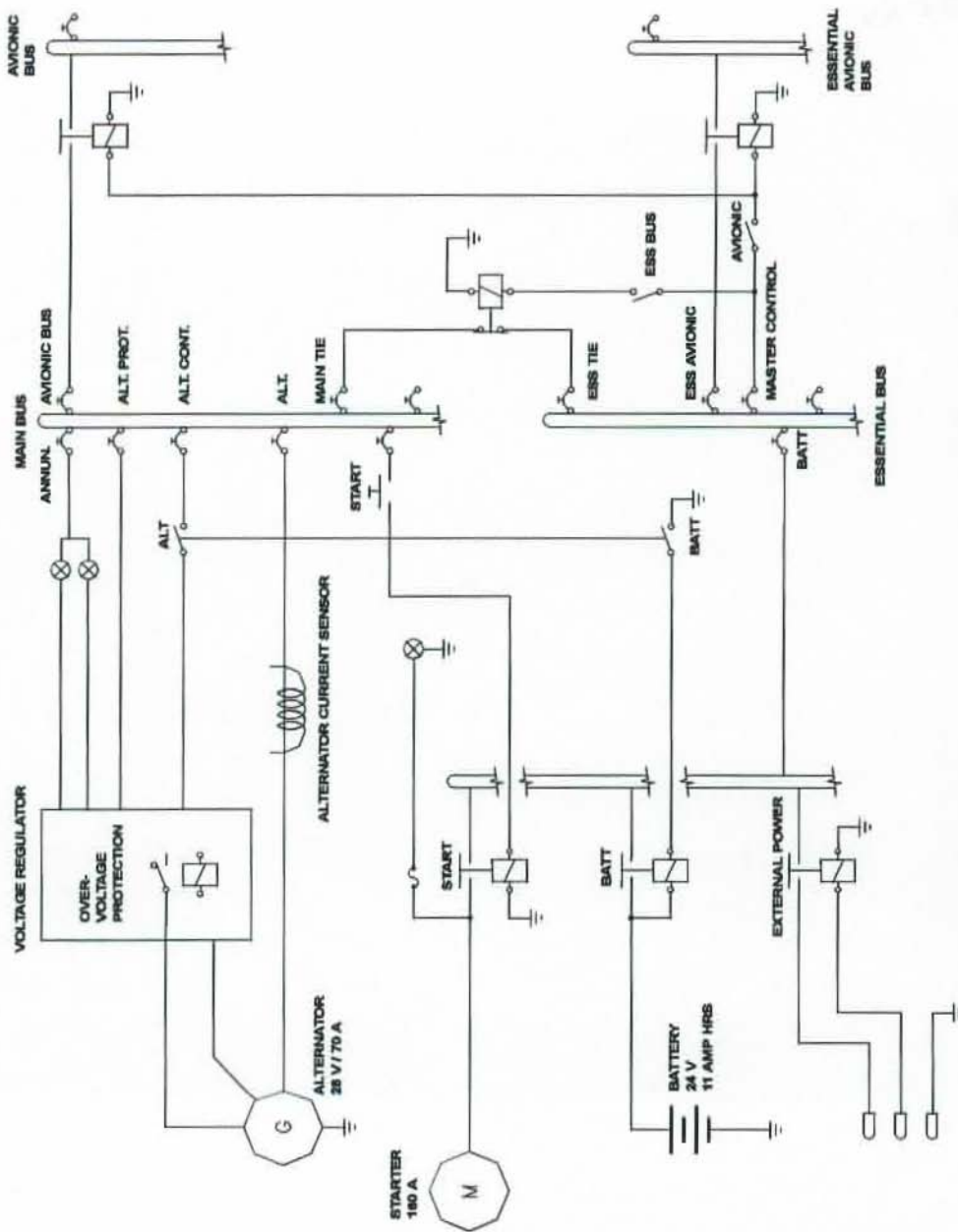
The designated place for the fuel quantity measuring device is the bag on the rear side of the pilot seat.



7.11 ELECTRICAL SYSTEM



Simplified wiring diagram for serial numbers without Essential Bus



Simplified wiring diagram for serial numbers with Essential Bus

7.11.1 GENERAL

The DA 40 has 28 Volt DC system, which can be sub-divided into:

- Power generation
- Storage
- Distribution
- Consumers

Power generation

The 70 ampère alternator (generator) is mounted on the front of the engine. It is driven by a V-belt, and charges the battery. In the event of alternator failure, the battery provides the system with electrical energy. Given the provision of these two independent sources of electrical power, the complete failure of the electrical system is extremely unlikely.

Storage

Power is stored in an 11 ampère-hour lead-acid battery, which is mounted in the right-hand side of the engine compartment. The battery is connected to the airplane electrical system via the main (70 ampère) circuit breaker.

In addition, a non-rechargeable dry battery is installed in the IFR model as a further source of power for the attitude gyro (artificial horizon) and the flood light. When the emergency switch is set to ON, these two systems are supplied with power for 1 hour and 30 minutes, independent of all other electrical consumers. During each 100 hour inspection, this battery is checked for proper functioning. Every 2 years or after use (broken seal on the switch) the battery cells must be replaced.

Distribution

Electrical power is distributed via the 'Main Bus' and - if installed - the 'Essential Bus'.

Master switch (ALT/BAT)

The 'Master Switch' is divided into a 'Master Switch (ALT)' on the left and a 'Master Switch (BAT)' on the right. Both switches together are known as the 'Master Switch'.

Consumers

The individual consumers (e.g. radio, electrical fuel pump, position lights, etc.) are connected to the main bus via automatic circuit breakers.

Designations and abbreviations used to identify the circuit breakers are explained in Section 1.5 DEFINITIONS AND ABBREVIATIONS.

Ignition

The basic version of the DA 40 is equipped with the electric start boost system SlickSTART. This system improves the start characteristics by delivering more spark energy during the engine start sequence. After engine starting the ignition is controlled by the conventional retard breaker magneto system.

As an option, the LASAR electronic ignition control unit can be installed instead of the SlickSTART system. This unit measures manifold pressure and RPM and uses these parameters to optimize the ignition timing. This provides for smooth engine running and improved starting behavior. If the electronic ignition control is not in operation, the status light for the ignition illuminates and the conventional magneto ignition takes over the ignition control. Also, during operation of the engine on only one magneto, for example during the magneto check, the ignition is not controlled electronically and the status light for the ignition should illuminate. For engine restart in flight without the electronic ignition control being operative an engine speed of more than 500 RPM is necessary. The magneto ignition is independent of the electrical network, therefore providing safe engine operation even in the event of a power failure.

Voltmeter

The voltmeter displays the potential on the main bus. If the alternator is operating, the alternator voltage is shown, otherwise it is that provided by the battery.

Ammeter

The ammeter displays the current with which the alternator is being loaded.

Landing and taxi lights

Landing and taxi lights are built into the left wing, and are each operated by means of a switch (LANDING, TAXI) on the row of switches on the instrument panel.

Position and strobe lights

Combined position and strobe lights (anti collision lights) are installed on both wing tips. Each system is operated by a switch (POSITION, STROBE) on the row of switches on the instrument panel.

Flood light

A two-dimensional light emitter is mounted above the instrument panel. It illuminates the instrument panel as well as all levers, switches, etc. With a rotary button (FLOOD) in the left-hand section of the instrument panel the flood light is switched on and its brightness is adjusted.

Instrument lighting

With a rotary button (INSTRUMENT) in the left-hand section of the instrument panel the internal lighting of the instruments is switched on and its brightness is adjusted.

Pitot heating

The Pitot probe, which provides measurement for the Pitot-static system, is electrically heated. The heating is activated with a switch (PITOT) on the row of switches on the instrument panel. The temperature is automatically kept constant by means of a thermal switch on the Pitot probe, and as an additional safety measure a thermal fuse is built in. If this thermal fuse is activated, the Pitot heating can no longer be switched on, and the Pitot heating caution will be displayed. In this case the system should be serviced.

NOTE

The Pitot heating caution will also be displayed whenever the Pitot heating system is switched OFF.

7.11.2 DAI ANNUNCIATOR PANEL

(WARNING, CAUTION AND STATUS LIGHTS)

There are two variants of the annunciator panel ('DAI' and 'White Wire'). The 'DAI' variant, which is described below, can be identified by the lights in the shape of a square.

Testing the annunciator panel

In the process of the pre-flight check the lights of the annunciator panel must be checked by operating the test switch. This is to check that the lights have not failed. All lights must be serviceable.

Alternator warning light (ALT)

The alternator warning light illuminates on alternator failure. The only remaining source of electrical power is the battery. The color is red.

The procedure to be followed upon alternator warning is given in 3.7.2 FAILURES IN THE ELECTRICAL SYSTEM.

Low voltage caution light (VOLT)

This caution light illuminates when the on-board voltage drops below 24 volts. It goes out again when the voltage exceeds 25 volts. The color is amber.

The procedure to be followed upon low voltage caution is given in 4B.3 FAILURES IN THE ELECTRICAL SYSTEM.

Fuel pressure warning light (FUEL PR)

The fuel pressure warning light illuminates when the fuel pressure drops below 14 psi. The color is red.

Low fuel caution lights (L FUEL and R FUEL)

Each tank has its own caution light. It starts to flash when the fuel quantity becomes low, and illuminates permanently when the quantity of usable fuel in the respective tank drops below 3 US gal (± 1 US gal). The indication is calibrated for straight and level flight. The light may illuminate during turns which are flown with slip, or while taxiing in curves. The color is amber.

Oil pressure warning light (OIL PR)

The oil pressure warning light illuminates when the oil pressure drops below 25 psi. The color is red.

The procedure to be followed upon oil pressure warning is given in 3.2.3 ENGINE PROBLEMS IN FLIGHT.

Door warning light (DOOR)

The door warning light illuminates when the front canopy and/or the rear door is not closed and locked. The color is red.

Status light for the ignition (IGN)

This light is only used when the electronic ignition control unit is installed.

The status light for the ignition illuminates when the electronic ignition control is not operating. In this case the conventional magneto ignition will be in use. The color is white.

The procedure to be followed upon illumination of the ignition status light is given in 4B.3 FAILURES IN THE ELECTRICAL SYSTEM.

Starter warning light (START)

The starter warning light illuminates when the starter is being operated or when the connection between the starter motor and the engine has not been broken. This occurs when the pinion of the starter motor remains engaged with the propeller flywheel. The color is red.

The procedure to be followed upon starter warning is given in 3.7.2 FAILURES IN THE ELECTRICAL SYSTEM.

Pitot heating caution light (PITOT)

The Pitot heating caution light is illuminated when the Pitot heating is not switched on, or when there is a failure of the Pitot heating system. The color is amber.

Prolonged operation of the Pitot heating on the ground can also cause the Pitot heating caution light to illuminate. In this case it indicates the activation of the thermal switch, which prevents overheating of the Pitot heating system on the ground. This is a normal function of the system. After a cooling period, the heating system will be switched on again automatically.

7.11.3 WHITE WIRE ANNUNCIATOR PANEL (WARNING, CAUTION AND STATUS LIGHTS)

There are two variants of the annunciator panel ('DAI' and 'White Wire'). The 'White Wire' variant, which is described below, can be identified by the flat front panel and the 'White Wire' logo on the display in the upper left corner.

Testing the annunciator panel

In the process of the pre-flight check, proper functioning of the annunciator panel must be verified. This functional check is automatically started after switching the battery master switch ON. All lights are flashed, and the aural alert is muted. By pressing the 'acknowledge' button, the lights are extinguished, and a momentary aural alert is sounded. This test verifies functionality of the microprocessor, the lights, and the aural signal.

The pilot may initiate additional system tests by holding the 'acknowledge' button for 2 seconds. All lights will begin flashing, and the aural alert will sound continuously.

Warning messages

A warning is indicated by a continuous aural alert (sounded in the airplane's intercomm system), flashing of the red WARNING light, and flashing of the red warning light associated with the affected system.

By pressing the 'acknowledge' button, which is now illuminated green, the aural alert will be terminated, and the WARNING light will be extinguished. The warning light associated with the affected system will change from flashing to solid illumination.

Caution messages

A caution is indicated by a momentary aural alert (sounded in the airplane's intercomm system), flashing of the amber CAUTION light, and flashing of the amber caution light associated with the affected system.

By pressing the 'acknowledge' button, which is now illuminated green, the CAUTION light will be extinguished. The caution light associated with the affected system will change from flashing to solid illumination.

The LOW FUEL caution message is displayed in a slightly different manner (extended functionality), which is described below.

Alternator warning message (ALTERNATOR)

The alternator warning message is displayed on alternator failure. The only remaining source of electrical power is the battery.

The procedure to be followed upon alternator warning is given in 3.7.2 FAILURES IN THE ELECTRICAL SYSTEM.

Low voltage caution message (LOW VOLTS)

The low voltage caution message is displayed when the on-board voltage drops below 24 volts. It is terminated when the voltage exceeds 25 volts again.

The procedure to be followed upon low voltage caution is given in 4B.3 FAILURES IN THE ELECTRICAL SYSTEM.

Fuel pressure warning message (FUEL PRESS)

The fuel pressure warning message is displayed when the fuel pressure drops below 14 psi.

Low fuel caution message (LOW FUEL)

As soon as the amount of usable fuel *in one tank* is less than 3 US gal (± 1 US gal), a caution message is displayed in the usual manner (momentary aural alert, flashing CAUTION light, flashing LOW FUEL caution light). Termination of the message is also done as usual ('acknowledge', CAUTION light is extinguished, LOW FUEL caution light changes to solid illumination).

As soon as the amount of usable fuel *in the second tank* is also less than 3 US gal (± 1 US gal), a caution message is displayed in a different manner. A *continuous* aural alert is sounded in the airplane's intercomm system, the amber CAUTION light is flashed, and the amber LOW FUEL caution light is flashed.

By pressing the 'acknowledge' button, which is now illuminated green, the aural alert will be terminated, and the CAUTION light will be extinguished. The LOW FUEL caution light will continue to be flashed.

The indication is calibrated for straight and level flight. The caution message may be triggered during turns which are flown with slip, or while taxiing in curves.

Oil pressure warning message (OIL PRESS)

The oil pressure warning message is displayed when the oil pressure drops below 25 psi.

The procedure to be followed upon oil pressure warning is given in 3.2.3 ENGINE PROBLEMS IN FLIGHT.

Door warning message (DOORS)

The door warning message is displayed when the front canopy and/or the rear door is not closed and locked.

Status light for the ignition (IGN)

This light is only used when the electronic ignition control unit is installed.

The status light for the ignition is illuminated when the electronic ignition control is not operating. In this case the conventional magneto ignition will be in use. The color is white.

The WARNING light, the CAUTION light, and the aural alert will *not* be activated.

The procedure to be followed upon illumination of the ignition status light is given in 4B.3 FAILURES IN THE ELECTRICAL SYSTEM.

Starter warning message (START)

The starter warning message is displayed when the connection between the starter motor and the engine has not been broken. This occurs when the pinion of the starter motor remains engaged with the propeller flywheel.

Furthermore, the START warning light is illuminated continuously as long as the starter is being operated. In this case the WARNING light and the aural alert will not be activated.

The procedure to be followed upon starter warning is given in 3.7.2 FAILURES IN THE ELECTRICAL SYSTEM.

Pitot heating caution message (PITOT)

The Pitot heating caution message is displayed when the Pitot heating is not switched on, or when there is a failure of the Pitot heating system.

Prolonged operation of the Pitot heating on the ground can also cause the Pitot heating caution message to be displayed. In this case it indicates the activation of the thermal switch, which prevents overheating of the Pitot heating system on the ground. This is a normal function of the system. After a cooling period, the heating system will be switched on again automatically.

Trim failure warning message (TRIM FAIL)

The White Wire annunciator panel is prepared for the installation of an autopilot in the DA 40. When the autopilot is installed and ready for operation, this warning message indicates a failure of the automatic trim system of the autopilot. For further details, refer to the Supplement to the AFM for the autopilot (if installed).

Unused lights

The White Wire annunciator panel has two lights for possible future use. These lights are currently unused.

7.12 PITOT-STATIC SYSTEM

Total pressure is measured at the leading edge of a Pitot probe under the left wing. Static pressure is measured at two orifices at lower and rear edges of the same probe. To protect against dirt and condensation there are filters in the system, which are accessible from the wing root. The Pitot probe is electrically heated.

In addition, some serial numbers have an alternate static valve installed on the underside of the instrument panel. With this valve, the static pressure in the cabin can be used as static pressure source in the event of a failure of the Pitot-static system.

7.13 STALL WARNING

If airspeed drops below approximately 1.1 times the stalling speed, the stall warning horn, located in the instrument panel, will sound. The horn becomes progressively louder the closer one gets to stalling speed. Suction at an orifice on the left wing leading edge activates the horn via a hose. The orifice for the stall warning in the left wing is marked by a red ring.

7.14 AVIONICS

The radio and navigation equipment is located in the central part of the instrument panel. A transmit switch for the radio is mounted on the end of each control stick. There are connection facilities for up to 4 headsets between the front seats.

CHAPTER 8

AIRPLANE HANDLING, CARE AND MAINTENANCE

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8.1 INTRODUCTION

Chapter 8 contains the manufacturer's recommended procedures for proper ground handling and servicing of the airplane. The Airplane Maintenance Manual (Doc. No. 6.02.01) lists certain inspection and maintenance requirements which must be followed if the airplane is to retain a new plane performance and reliability.

8.2 AIRPLANE INSPECTION INTERVALS

For maintenance work on engine and propeller, the currently effective Operator's Manuals, Service Instructions, Service Letters and Service Bulletins of Lycoming and mt-Propeller must be followed. For airframe inspections, the currently effective checklists/manuals of the manufacturer must be followed.

CAUTION

Unscheduled maintenance checks are required after:

- hard landings
- propeller strike
- engine fire
- lighting strike
- occurrence of other malfunctions and damage

Unscheduled maintenance checks are described in the Airplane Maintenance Manual (Doc. No. 6.02.01; Section 05-50).

8.3 AIRPLANE ALTERATIONS OR REPAIRS

Alterations or repairs of the airplane may be carried out only according to the Airplane Maintenance Manual, Doc. No. 6.02.01, and only by authorized personnel.

8.4 GROUND HANDLING / ROAD TRANSPORT

8.4.1 GROUND HANDLING WITHOUT TOW BAR

During forward traversing the nose wheel will follow the movement of the airplane. Change in direction is achieved by pulling on the propeller near the spinner. To traverse in the rear direction, the tail section of the airplane should be pushed down until the nose wheel is clear of the ground. This method can also be used to turn the airplane around its main landing gear.

8.4.2 GROUND HANDLING WITH TOW BAR

For pushing or pulling the airplane on the ground, it is recommended to use the tow bar which is available from the manufacturer. The tow bar is bent apart and engaged in the appropriate holes in the nose wheel fairing as shown on the picture below. The arresting knob must be fully engaged.



WARNING

The tow bar must be removed before starting the engine.

CAUTION

The tow bar may only be used for moving the airplane on the ground by hand. After moving the airplane, the tow bar must be removed.

NOTE

When moving the airplane rearward, the tow bar must be held firmly to prevent abrupt sideward deflection of the nose wheel.

8.4.3 PARKING

For short term parking, the airplane must be positioned into the wind, the parking brake must be engaged and the wing flaps must be in the retracted position. For extended and unattended parking, as well as in unpredictable wind conditions, the airplane must be anchored to the ground or placed in a hangar. Parking in a hangar is recommended.

Control surfaces gust lock

The manufacturer offers a control surfaces gust lock which can be used to block the primary controls. It is recommended that the control surfaces gust lock be used when parking outdoors, because otherwise the control surfaces can hit the stops in strong tail wind. This can lead to excessive wear or damage.

WARNING

The control surfaces gust lock must be removed before flight.

The control surfaces gust lock is installed as follows:

1. Move the rudder pedals fully rearward.
2. Engage the control surfaces gust lock with the pedals.
3. Engage the stick, wrap straps around stick once.
4. Attach the locks and tighten the straps.

For removal, reverse the sequence.



8.4.4 MOORING

The tail fin of the airplane has a hole which can be used to tie-down the airplane to the ground. Also on each wing near the wing tip, an eyelet with a metric M8 thread can be installed and used as tie-down points.

8.4.5 JACKING

The DA 40 can be jacked at the two jackpoints located on the lower side of the fuselage's LH and RH root ribs as well as at the tail fin.

8.4.6 ALIGNMENT

For alignment push down on the tail section at the fuselage/vertical tail junction until the nose wheel is clear of the ground. With the nose wheel free, the DA 40 can be turned around the main landing gear. After turning the airplane into the correct position, release the tail section until the nose wheel is back on the ground.

8.4.7 ROAD TRANSPORT

For transporting the airplane on the road it is recommended that an open trailer be used. All airplane components must be stored on a cushioned surface and secured to avoid any movement during transportation.

1. Fuselage:

The fuselage should stand on the main and nose landing gear. It must be ensured that the fuselage will not move in a forward, backward or upward direction. Furthermore, it must be ensured that the propeller has sufficient clearance so that it cannot be damaged due to fuselage movement during transportation.

2. Wings:

For transportation, both wings must be removed from the fuselage. To avoid any damage, the wings are stored in an upright position on the leading edge with the root rib area positioned on an upholstered profiled surface with a width of at least 400 mm (1.3 ft). The outside wing area (approximately 3 m (10 ft) from the root rib area) is placed on an upholstered profiled surface with a minimum width of 300 mm (1 ft).

The wings must be secured to avoid any sliding movement to the rear.

3. Horizontal stabilizer:

The horizontal stabilizer is stored flat on the trailer and secured with straps, or in an upright position sitting on the leading edge on a profiled surface. All storing surfaces must be upholstered with felt or cellular rubber.

8.5 CLEANING AND CARE

CAUTION

The airplane must be kept clean. The bright surface prevents the structure from overheating.

CAUTION

Excessive dirt deteriorates the flight performance.

8.5.1 PAINTED SURFACES

The entire surface of the airplane is painted with a white weatherproof two component paint. Nevertheless, the airplane should be protected against moisture and dampness. The airplane should not be stored outdoors for long periods of time. Moisture that has penetrated must be removed by storing the affected parts in a dry place and turning them over several times.

Dirt, insects, etc. can be removed with water alone and if necessary with a mild detergent. An automotive paint cleaner can be used for stubborn spots. For best results, clean the airplane after the day's flying is ended, so that the dirt will not become ingrained.

Oil stains, exhaust stains, etc. on the lower fuselage skin can be removed with a cold detergent. Before starting, ensure that the detergent does not affect the surface finish.

Use commercial automotive preservatives without silicone additives to conserve the paint finish.

8.5.2 CANOPY AND REAR DOOR

The canopy and rear door should be cleaned with 'Plexiklar' or any other acrylic glass detergent if available; otherwise use lukewarm water. Final cleaning should be done with a clean piece of chamois-leather or soft cloth. Never rub or polish dry acrylic glass.

8.5.3 PROPELLER

Damage and malfunctions during operation must be inspected by authorized personnel.

Surface

The manufacturer uses PU paint or acrylic paint which is resistant to almost any solvent. The blades may be treated with commercial automotive cleaning agents or preservatives. The penetration of moisture into the wooden core must be avoided by all means. Should doubts arise, an appropriately rated inspector must be consulted.

8.5.4 ENGINE

Engine cleaning is part of the scheduled inspections.

8.5.5 INTERIOR SURFACES

The interior should be cleaned using a vacuum cleaner. All loose items (pens, bags etc.) should be removed or properly stored and secured.

All instruments can be cleaned using a soft dry cloth, plastic surfaces should be wiped clean using a damp cloth without any cleaning agents.

