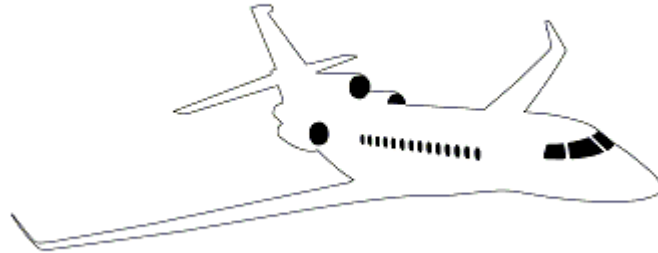


# ***FALCON SERVICE BULLETIN***

## **FALCON 7X**



**No 056-R1**

**JULY 20, 2016**

**MANDATORY**

**ATA 20**

**STANDARD PRACTICES  
AIRFRAME WIRING  
PROTECTION IMPROVEMENT OF VARIOUS WIRING  
ASSEMBLIES**

Copyright © 2016 by Dassault Aviation  
All rights reserved. No part of this work may be reproduced or copied  
in any form or by any means without written permission  
of Dassault Aviation.

**FALCON 7X**

**7X -056**

**STANDARD PRACTICES  
AIRFRAME WIRING  
PROTECTION IMPROVEMENT OF VARIOUS WIRING ASSEMBLIES**

---

**Initial issuance**

October 03, 2014

List of effective pages :  
1 to 36

---

**Revision 1**

July 20, 2016

List of effective pages :  
1 to 34  
List of modified pages :  
1, 3, 4, 6, 7, 10, 11, 12, 13, 14, 15, 16, 26, 29, 34.

**REASON:-** Modification of section 7X-056-4 effectivity:

Aircraft that have already implemented the initial issuance of this Service Bulletin have to check whether their S/N is in the new effectivity list of section 7X-056-4.

If so, the related procedure must be applied.

- Various improvements.

**STANDARD PRACTICES  
 AIRFRAME WIRING  
 PROTECTION IMPROVEMENT OF VARIOUS WIRING ASSEMBLIES**

**1. PLANNING INFORMATION**
**A. EFFECTIVITY**

This Service Bulletin is applicable to FALCON 7X aircraft with serial numbers 2 and subsequent as per the following table:

	Applicable to aircraft
Section 7X-056-1	without M876
Section 7X-056-2	without M897
Section 7X-056-3	without M900
Section 7X-056-4	Aircraft S/N 002 through 131
Section 7X-056-5	without M954
Section 7X-056-6	without M980
Section 7X-056-7	without M1021
Section 7X-056-8	Aircraft S/N 002 to 215

**B. REASON**

The purpose of this Service Bulletin is to improve the protection of various wiring assemblies in order to remove any risks of interference or contact that could lead to a premature damage of the electrical wiring.

**C. DESCRIPTION**

The operation consists in inspecting various wiring assemblies and improving the layout robustness by:

- modifying the clamping and/or routing,
- adding new brackets, clamps, cable protections,
- improving connection with lock wire.

**D. COMPLIANCE**

Mandatory:

- It is anticipated that this Service Bulletin will be rendered mandatory by an EASA Airworthiness Directive to be released.
- As prescribed by the Airworthiness Authorities of the country of registration of the aircraft.

**E. APPROVAL**

This Service Bulletin covers DASSAULT AVIATION modifications FALCON 7X M876, M897, M900, M937, M954, M980, M1021 and M1470 that have been approved under the authority of DOA nr. EASA.21J.051.

**FALCON 7X**

**7X -056**

The technical content of this document is approved under the authority of DOA nr. EASA.21J.051.

F. LABOR

Estimated labor-hours: Refer to Service Bulletin Commercial Summary.

NOTE: These labor-hours only concern the work described in this Service Bulletin and do not include other maintenance work that may be performed on this occasion.

G. MATERIAL - PRICE AND AVAILABILITY

The modification kit may be obtained from either address listed below:

Western hemisphere: DASSAULT FALCON JET CORP.  
SPARES DISTRIBUTION CENTER  
200 RISER ROAD  
LITTLE FERRY, NJ 07643 U.S.A.  
Telephone:  
• CANADA and U.S.A.: 1-800-800-4036  
• MEXICO: 001-800-800-4036  
• Other countries: 1-201-541-4809  
Fax:  
• CANADA and U.S.A.: 1-800-800-4817  
• MEXICO: 001-800-800-4817  
• Other countries: 1-201-440-7021

Other continents: DASSAULT AVIATION  
Falcon Spares  
BOITE POSTALE N°101  
AEROPORT DU BOURGET  
93350 - LE BOURGET Cedex (FRANCE)  
Please contact your Dassault Aviation  
Account Representative  
Telephone: 33 (0)1.48.35.56.78  
Fax: 33 (0)1.48.35.56.00

Price and availability on request.

H. TOOLING - PRICE AND AVAILABILITY

Normal maintenance tooling.

I. WEIGHT AND BALANCE

Change in weight: None.

Change in balance with respect to 25% MAC: None.

J. REFERENCES

Supplemental Maintenance Manual:



**FALCON 7X****7X -056**

- REMOVAL/INSTALLATION OF THE FORWARD LAVATORY HEADLINER

Structural Repair Manual:

- DRILLING (Refer to [RPI 51-40-03](#))
- RIVETING (Refer to [RPI 51-40-04](#))
- COUNTERSINKING (Refer to [RPI 51-40-06](#))

Aircraft Maintenance Manual:

- ◆
- GENERAL MAINTENANCE AND SAFETY PRECAUTIONS (Refer to [TASK 20-00-00-910-801](#)).
- BASIC SEALING METHODS (Refer to [TASK 20-32-00-390-801](#)).
- INSPECTION / CHECK OF ELECTRICAL AND ELECTRONIC SYSTEMS (Refer to [TASK 20-90-00-200-801](#))
- INSERTION AND EXTRACTION OF A WIRE IN A CONNECTOR (Refer to [TASK 20-95-00-910-801](#)).
- OPERATIONAL TEST OF THE HF COMMUNICATION SYSTEM (Refer to [TASK 23-12-00-710-801](#))
- OPERATIONAL TEST OF THE "BKUP" AUDIO BUTTON (Refer to [TASK 23-50-00-710-802](#))
- ENERGIZATION / DE-ENERGIZATION OF THE AIRCRAFT (Refer to [TASK 24-00-00-860-801](#)).
- REMOVAL / INSTALLATION OF THE REAR SECONDARY-POWER DISTRIBUTION-BOXES (SPDB) (Refer to [TASK 24-62-09-900-801](#)).
- REMOVAL / INSTALLATION OF THE COCKPIT BULKHEADS (Refer to [TASK 25-10-01-900-801](#))
- REMOVAL / INSTALLATION OF THE COCKPIT LININGS (Refer to [TASK 25-13-13-900-801](#)).
- REMOVAL / INSTALLATION OF THE BAGGAGE COMSECTIONMENT LININGS (Refer to [TASK 25-52-01-900-801](#)).
- REMOVAL / INSTALLATION OF THE LH FRONT FCS RACK (Refer to [TASK 27-90-05-900-801](#)).
- REMOVAL / INSTALLATION OF THE LH REAR FCS RACK (Refer to [TASK 27-90-05-900-802](#)).
- REMOVAL / INSTALLATION OF THE RH REAR FCS RACK (Refer to [TASK 27-90-05-900-804](#)).
- REMOVAL / INSTALLATION OF THE T34 BODY FAIRINGS (Refer to [TASK 53-60-00-900-801](#)).
- REMOVAL / INSTALLATION OF THE T5 BODY FAIRINGS (Refer to [TASK 53-70-05-900-801](#)).

**K. OTHER PUBLICATIONS AFFECTED**

The following publication will be revised at a later date to account for the changes introduced by this Service Bulletin:

- Illustrated Parts Catalog.

**2. ACCOMPLISHMENT INSTRUCTIONS**

**A. PRELIMINARY STEPS**

- (1) The aircraft must be in the maintenance configuration (Refer to [TASK 20-00-00-910-801](#)).
- (2) Make sure that the aircraft electrical systems are de-energized, if not, de-energize the aircraft systems (Refer to [TASK 24-00-00-860-801](#)).

**B. GENERAL ELECTRICAL INSTRUCTIONS**

- (1) When mechanical retrofit and/or modification operations are performed on the aircraft, take special care to protect the wiring in order to prevent metal chips (or burrs) from penetrating into the wire bundles, since they could eventually damage the insulation sheathing of the wires. Whenever possible, it is recommended that the wiring be removed from the work area. If that is not possible, it is absolutely necessary to protect the wiring.
- (2) Due to the risk of damaging the wire insulation, it is prohibited to blindly drill holes through a box structure containing wiring.
- (3) New wires that are installed alongside existing wires should be secured by the existing clamp.
- (4) Check that all connections are properly protected to ensure proper insulation, and that markings are visible and appropriate.
- (5) Connector wires should be inserted and extracted as per the recommendations in Standard practices, (Refer to [TASK 20-95-00-910-801](#)).
- (6) Electric and electronic equipment should be removed and installed as per the recommendations in Standard Practices, (Refer to [TASK 20-00-00-910-801](#)).
- (7) Visually inspect the work area for particles or debris, then vacuum-clean. The blowing of compressed air is prohibited since it could move metal chips to inaccessible areas.

**C. GENERAL STRUCTURAL WORK INSTRUCTIONS**

- (1) Obey the Structural Repair Manual and Standard Practices procedures for the following operations:
  - drilling (Refer to [RPI 51-40-03](#)),
  - riveting (Refer to [RPI 51-40-04](#)),
  - countersinking (Refer to [RPI 51-40-06](#)),
  - basic sealing methods (Refer to [TASK 20-32-00-390-801](#)).

**D. SECTION 7X-056-1 (M876) (Mechanical servicing compartment)**

Refer to **fig. 1**

**NOTE:** On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to [TASK 20-90-00-200-801](#)).

- (1) Unclamp the wiring bundle.

**NOTE:** If necessary, unclamp the surrounding pipes for a better access.

- (2) Discard the stud support (A-fig. 1).

- (3) Install the bracket (1-fig. 1).
- (4) Counterdrill the bracket and the panel at Fr 44 at dia. 2.5 mm (0.1 in).
- (5) Enlarge the holes to dia. 5.2 mm (0.2 in).
- (6) Install the new electrical bracket (1-fig. 1), using attaching parts (2, 3 and 4-fig. 1).
- (7) Clamp the electrical bundle and the surrounding pipes (if unclamped).

E. SECTION 7X-056-2 (M897, point 2) (Baggage compartment)

Refer to **fig. 2**

**NOTE:** On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to [TASK 20-90-00-200-801](#)).

**REMARK:** OTHER SIMILAR PTFE PROTECTIVE SHEETS ARE INSTALLED ON THE UPPER GUTTERS IN SECTION 7X-056-6 (M980).

- (1) Remove:
  - (a) The LH rear FCS rack ([2003CZ](#)) (Refer to [TASK 27-90-05-900-802](#)).
  - (b) The RH rear FCS rack ([2004CZ](#)) (Refer to [TASK 27-90-05-900-804](#)).
- (2) On each side, unclamp the wiring bundles routed on the elbowed gutters to be modified.
- (3) Move the bundles away from the gutter and bond the new PTFE protective sheets (5 and 6-fig. 2) over the rivet heads, using **interlaying sealant** (Refer to [TASK 20-32-00-390-801](#)).
- (4) Allow to cure.
- (5) Secure the electrical bundle.
- (6) Install:
  - (a) The LH rear FCS rack ([2003CZ](#)) (Refer to [TASK 27-90-05-900-802](#)).
  - (b) The RH rear FCS rack ([2004CZ](#)) (Refer to [TASK 27-90-05-900-804](#)).

F. SECTION 7X-056-3 (M900, points 1, 2, 3 and 4)

**NOTE:** On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to [TASK 20-90-00-200-801](#)).

- (1) Point 1 (Exterior access, under fairing)

Refer to **fig. 3**

  - (a) Remove the rear fuel tank rear fairing ([168AR](#)) (Refer to [TASK 53-60-00-900-801](#)).
  - (b) Secure the electrical bundle with new clamp (7-fig. 3), replace the existing screw with a new one (8-fig. 3).
  - (c) Install the rear fuel tank rear fairing ([168AR](#)) (Refer to [TASK 53-60-00-900-801](#)).
- (2) Point 2 (Baggage compartment)

Refer to **fig. 4**

  - (a) Remove the FR. 36 - FR. 39 oblique lining ([272KZ](#)) (Refer to [TASK 25-52-01-900-801](#)).

- (b) Position the new bracket (12-fig. 4) on the stiffener.
  - (c) Drill the stiffener to a dia. 2.5 mm (0.01 in.).
  - (d) Counterdrill the new bracket (12-fig. 4) and the stiffener for new rivets (13-fig. 4).
  - (e) Install the bracket (12-fig. 4), using rivets (13-fig. 4).
  - (f) Identify the stiffener with its new P/N F7XC535136214A2.
  - (g) Secure the electrical feeders on the bracket, using new clamp (9-fig. 4) and nut (10-fig. 4).
  - (h) Install the FR. 36 - FR. 39 oblique lining (**272KZ**) (Refer to **TASK 25-52-01-900-801**).
- (3) Point 3 (Cockpit, LH electrical cabinet FR 8 - 9)
- Refer to **fig. 5** and **fig. 6**
- (a) Remove the LH front FCS rack (**2001CZ**) (Refer to **TASK 27-90-05-900-801**).
  - (b) Unclamp the LH main electrical bundle, the LH base feeder and the Cockpit fuse harness from the three brackets to be removed.
  - (c) Remove the three brackets from the web (one detempel bracket and two stud supports).
  - (d) Install new rivet (27-fig. 6) to plug unused hole.
  - (e) Counterdrill the new bracket (15-fig. 6) according to the web holes.
  - (f) Install the new bracket (15-fig. 6) on the web, using rivets (25 and 26-fig. 6).
  - (g) Form the LH main electrical bundle for the new clamp (16-fig. 6).
  - (h) Attach the clamp (16-fig. 6) on the new bracket (15-fig. 6), using screws and washers (19 and 20-fig. 6).
  - (i) Secure the LH base feeder and the Cockpit fuse harness on the new bracket (15-fig. 6), using clamps (17 and 18-fig. 6) and nut (21-fig. 6).
  - (j) Install the LH front FCS rack (**2001CZ**) (Refer to **TASK 27-90-05-900-801**).
- (4) Point 4
- Refer to **fig. 7**
- (a) Remove:
    - The cockpit lateral lining No.5 (**222XZ**) (Refer to **TASK 25-13-13-900-801**).
    - The cockpit lateral lining No.7 (**222ZZ**) (Refer to **TASK 25-10-01-900-801**).
  - (b) Unclamp the RH main electrical bundle and move away from the RH cabinet structure.
  - (c) Bond the new protective sheet (28-fig. 7) to the crossbeam, using **interlaying sealant** (Refer to **TASK 20-32-00-390-801**).
  - (d) Allow to cure.
  - (e) Secure the RH main electrical bundle.
  - (f) Install (Refer to **TASK 25-13-13-900-801**):
    - The cockpit lateral lining No.5 (**222XZ**) (Refer to **TASK 25-13-13-900-801**).
    - The cockpit lateral lining No.7 (**222ZZ**) (Refer to **TASK 25-10-01-900-801**).
  - (g) Perform the operational test of the "BKUP" audio button (Refer to **TASK 23-50-00-710-802**).

(h) Perform the operational test of the HF communication system (Refer to [TASK 23-12-00-710-801](#)).

G. SECTION 7X-056-4 (M937, points 2, 3 and 4) (Baggage compartment)

NOTE: On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to [TASK 20-90-00-200-801](#)).

(1) Point 2

Refer to [fig. 8](#) and [fig. 9](#)

- (a) Remove the LH rear SPDB ([L2000PM](#)) (Refer to [TASK 24-62-09-900-801](#)).
- (b) Replace the existing protector with a new one (30-fig. 9)
- (c) Install the LH rear SPDB ([L2000PM](#)) (Refer to [TASK 24-62-09-900-801](#)).

(2) Point 3

Refer to [fig. 8](#) and [fig. 10](#)

- (a) Remove the LH rear SPDB ([L2000PM](#)) (Refer to [TASK 24-62-09-900-801](#)).
- (b) Unclamp the four electrical feeders.
- (c) Remove the existing bracket from the FR40.
- (d) Counterdrill the new bracket (31-fig. 10) according to the FR40 holes.
- (e) Install the the new bracket (31-fig. 10) on the FR40, using four rivets (32-fig. 10).
- (f) Secure the four electrical feeders on the new bracket, using existing attaching parts and new nut (33-fig. 10).
- (g) Install the LH rear SPDB ([L2000PM](#)) (Refer to [TASK 24-62-09-900-801](#)).

(3) Point 4

Refer to [fig. 8](#) and [fig. 11](#)

- (a) Remove the rear under-pylon fairing ([173FL](#)) (Refer to [TASK 53-70-05-900-801](#)).
- (b) Remove the LH rear FCS rack ([2003CZ](#)) (Refer to [TASK 27-90-05-900-802](#)).
- (c) Unclamp the wiring bundle from the existing bracket to be replaced.
- (d) Remove the bracket.

NOTE: Recover the metal chips from the exterior of the aircraft.

- (e) Install the new bracket (34-fig. 11) with [interlaying sealant](#), using rivets (35-fig. 11) with [interlaying sealant](#) under the rivet head (Refer to [TASK 20-32-00-390-801](#)).

(f) Position [ROUNDIT](#) type sheath as indicated on [fig. 11](#) and maintain it in position using white bindings.

NOTE: Local procurement.

- (g) Secure the wiring bundle to the new support with [cable grip](#).
- (h) Install the LH rear FCS rack ([2003CZ](#)) (Refer to [TASK 27-90-05-900-802](#)).
- (i) Install the rear under-pylon fairing ([173FL](#)) (Refer to [TASK 53-70-05-900-801](#)).

H. SECTION 7X-056-5 (M954, points 1, 2 and 3)

NOTE: On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to [TASK 20-90-00-200-801](#)).

(1) Point 1 (Mechanical servicing compartment)

Refer to **fig. 12**

- (a) Locate the negative electrical power lead connection of the hydraulic backup pump (**1002GP**).
- (b) Unscrew the negative electrical power lead from the aircraft structural ground.
- (c) Connect the negative electrical power lead with new locking plate (40-fig. 12) and new screw (41-fig. 12), torque value: 41 in.lbf ± 4 in.lbf (0.45 m.daN ± 0.04 m.daN).
- (d) Secure the screw with lock wire.
- (e) Apply **vinyl varnish** on the fastener assembly.

(2) Point 2 (Rear toilet compartment, RH side)

Refer to **fig. 13**

- (a) Remove the interior accommodation in order to gain access to the negative DC power lead connection of the RAT TRU (**2000PN**).
- (b) Unscrew the negative DC power lead from the aircraft structural ground.
- (c) Drill the ground support to a dia. 2.5 mm (0.01 in.).
- (d) Connect the negative DC power lead to the structural ground using new attaching parts (42, 43 and 44-fig. 13), torque value: 77.88 in.lbf ± 7.7 in.lbf (0.88 m.daN ± 0.08 m.daN).
- (e) Apply **vinyl varnish** on fastener assembly.
- (f) Secure the screw with lock wire.

(3) Point 3 (Cabin, FR 14 RH side)

Refer to **fig. 14**

NOTE: Only aircraft with M373 applied are concerned by point 3.

- (a) Remove the interior accommodation (RH main galley/Crew rest/partition/valance panel) in order to gain access to the routing of electrical feeders at FR14.
- (b) Unclamp, protect and move the electrical feeders away from the modification area (bracket (A-fig. 14)).
- (c) Remove the two rivets from the bracket (A-fig. 14).
- (d) Counterdrill the new bracket (45-fig. 14) according to the FR14 existing holes.
- (e) Install the new bracket (45-fig. 14), using rivets (46-fig. 14).
- (f) Secure the electrical feeders with existing clamps and new clamp (47-fig. 14), using attaching parts (48 and 49-fig. 14).
- (g) Restore the interior accommodation.

**FALCON 7X**

**7X -056**

I. SECTION 7X-056-6 (Baggage compartment) (M980)

Refer to **fig. 15**, **fig. 16** and **fig. 17**

NOTE: On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to **TASK 20-90-00-200-801**).

REMARK: OTHER SIMILAR PTFE PROTECTIVE SHEETS ARE INSTALLED ON THE LOWER GUTTERS IN PART 7X-056-2 (M897, POINT 1).

(1) Remove:

- (a) The LH rear FCS rack (**2003CZ**) (Refer to **TASK 27-90-05-900-802**).
- (b) The RH rear FCS rack (**2004CZ**) (Refer to **TASK 27-90-05-900-804**).
- (c) The LH rear SPDB (**L2000PM**) (Refer to **TASK 24-62-09-900-801**).
- (d) The RH rear SPDB (**R2000PM**) (Refer to **TASK 24-62-09-900-801**)

(2) On LH and RH side of the FR40, move the bundles away from the gutters and bond the new PTFE protective sheets (50-fig. 16) and (50-fig. 17) over the head rivets, using **interlaying sealant** (Refer to **TASK 20-32-00-390-801**).

(3) Allow to cure.

(4) Secure the electrical bundles.

(5) Install:

- (a) The LH rear FCS rack (**2003CZ**) (Refer to **TASK 27-90-05-900-802**).
- (b) The RH rear FCS rack (**2004CZ**) (Refer to **TASK 27-90-05-900-804**).
- (c) The LH rear SPDB (**L2000PM**) (Refer to **TASK 24-62-09-900-801**).
- (d) The RH rear SPDB (**R2000PM**) (Refer to **TASK 24-62-09-900-801**)

J. SECTION 7X-056-7 (M1021) (Fuselage, FR24 under fairing)

Refer to **fig. 18**

NOTE: On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to **TASK 20-90-00-200-801**).

- (1) Remove the CTR fuel tank booster pump fairing (**147FL**) (Refer to **TASK 53-60-00-900-801**).
- (2) Locate the two clamps to be re-oriented. Unscrew them.
- (3) Orient the clamps in accordance with the **fig. 18** and screw them.
- (4) Install the CTR fuel tank booster pump fairing (**147FL**) (Refer to **TASK 53-60-00-900-801**).

K. SECTION 7X-056-8 (M1470, RH electrical cabinet / toilet crew, FR 9, STR 8)

Refer to **fig. 19**

NOTE: On each area to be checked and modified, perform an inspection/check of the electrical components (wiring bundle and feeders) (Refer to **TASK 20-90-00-200-801**).

- (1) Remove the forward lavatory (Refer to SMM) to gain access to the feeder FR 9,STR 8.



**FALCON 7X**

**7X -056**

- (2) Remove cockpit lateral lining No.5 (**222XZ**) (Refer to **TASK 25-13-13-900-801**).
- (3) Remove the conditioning duct P/N F7XC213240201A1 (c-fig. 19).
- (4) Cut the cable grips (a-fig. 19).
- (5) Remove and discard the cable holder (b-fig. 19).
- (6) Position the new bracket (**51-fig. 19**) and counterdrill the third hole (V1-fig. 19) for rivet (**52-fig. 19**) (Refer to **RPI 51-40-04**).
- (7) Attach the new bracket (**51-fig. 19**) with three rivets (**52-fig. 19**) (Refer to **RPI 51-40-04**).
- (8) Route the two feeders with cable grips (**53-fig. 19**).
- (9) Install the conditioning duct (c-fig. 19).
- (10) Install cockpit lateral lining No.5 (**222XZ**).
- (11) Install the forward lavatory (Refer to SMM).

L. RECORDING

Record compliance with this Service Bulletin in the appropriate aircraft documents.

Fill out the electronic Service Bulletin reply form located in the "Service Bulletin" page on the Falcon portal.



**FALCON 7X**
**7X -056**
**3. MATERIAL INFORMATION**

**NOTE:** Compliance with this Service Bulletin by substituting the part numbers below with interchangeable subsequent part numbers is acceptable, provided the subsequent part numbers have the same effectivity as the original ones.

**A. MODIFICATION OF SECTION NUMBER**

NEW P/N	KEY WORD	OLD P/N	Section
F7XC535136214A2	Stiffener	F7XC535136214A1	7X-056-3, point 2

**B. MODIFICATION KIT**

(1) Section 7X-056-1 (M876):

Kit P/N SBF7X0056A				
NEW P/N	Q T Y	KEY WORD	OLD P/N	ITEM/FIG
<b>SBF7X0056FLAT</b>	<b>1</b>	<b>Kit including:</b>		
• F7XC535556615	1	Bracket	F7XC535556263	- (1-fig. 1) -
• NAS6403A14	2	Screw		- (2-fig. 1) -
• 23116DG050A	2	Washer		- (3-fig. 1) -
• HW17-3DF	2	Nut		- (4-fig. 1) -

(2) Section 7X-056-2 (M897, point 2):

Kit P/N SBF7X0056B			
NEW P/N	Q T Y	KEY WORD	ITEM/FIG
<b>SBF7X0056DLAT</b>	<b>1</b>	<b>Kit including:</b>	
• 30x14x0.5	4	PTFE protective sheet (Std 28-260)	- (5-fig. 2) -
• 33x27x0.5	2	PTFE protective sheet (Std 28-260)	- (6-fig. 2) -

**FALCON 7X**
**7X -056**

(3) Section 7X-056-3 (M900, points 1, 2, 3 and 4):

Kit P/N SBF7X0056C				
NEW P/N	QTY	KEY WORD	INSTRUCTIONS DISPOSITION	ITEM/FIG
78651RH8CG	1	Clamp	Point 1	- (7-fig. 3) -
NAS9203-13	1	Screw	Point 1	- (8-fig. 3) -
78652CT12CF	1	Clamp	Point 2	- (9-fig. 4) -
HW17-3DF	1	Nut	Point 2	- (10-fig. 4) -
<b>SBF7X0056ELAT</b>	<b>1</b>	<b>Kit including:</b>	Point 2	
• F7XC535554341	1	Bracket	Point 2	- (12-fig. 4) -
• 21215DE3206	2	Rivet	Point 2	- (13-fig. 4) -
F7XC531521579A1P02	1	Bracket	Point 3	- (15-fig. 6) -
F7XC531521581P01	1	Bracket	Point 3	- (16-fig. 6) -
78651RH11CG	1	Clamp	Point 3	- (17-fig. 6) -
78651RH6CG	1	Clamp	Point 3	- (18-fig. 6) -
NAS6203-2	2	Screw	Point 3	- (19-fig. 6) -
NAS1149G0316P	2	Washer	Point 3	- (20-fig. 6) -
HW17-3DF	1	Nut	Point 3	- (21-fig. 6) -
35053-4008ID	2	Rivet	Point 3	- (25-fig. 6) -
35053-3207ID	1	Rivet	Point 3	- (26-fig. 6) -
35053-3206ID	1	Rivet	Point 3	- (27-fig. 6) -
TEFLON-1FCOL	1	Protective sheet (100x70x0.25)	Point 4	- (28-fig. 7) -

(4) Section 7X-056-4 (M937, points 2, 3 and 4):

Kit P/N SBF7X0056D				
NEW P/N	QTY	KEY WORD	INSTRUCTIONS DISPOSITION	ITEM/FIG
<b>SBF7X0056BLAT</b>	<b>1</b>	<b>Kit including:</b>		
• F7XC535553791A1	1	Protector	Point 2	- (30-fig. 9) -
• F7XC535553030	1	Bracket	Point 3	- (31-fig. 10) -
• 35053DE3207	4	Rivet	Point 3	- (32-fig. 10) -
• HW17-3DF	1	Nut	Point 3	- (33-fig. 10) -
• F7XC535553061A1 (see NOTE)	1	Bracket	Point 4	- (34-fig. 11) -

**FALCON 7X**
**7X -056**

Kit P/N SBF7X0056D				
NEW P/N	Q T Y	KEY WORD	INSTRUCTIONS DISPOSITION	ITEM/FIG
• 35053-4009ID	2	Rivet	Point 4	- (35-fig. 11) -

**NOTE:** Order kit SBF7X0056L below if the bracket cannot be installed due to rivet center distance.

Kit P/N SBF7X0056L				
NEW P/N	Q T Y	KEY WORD	INSTRUCTIONS DISPOSITION	ITEM/FIG
F7XC535553061A1SB1	1	Bracket	Point 4	- (34-fig. 11) -

(5) Section 7X-056-5 (M954, points 1, 2, 3):

Kit P/N SBF7X0056E				
NEW P/N	Q T Y	KEY WORD	INSTRUCTIONS DISPOSITION	ITEM/FIG
F7XC535556619--1	1	Locking plate	Point 1	- (40-fig. 12) -
NAS1351-4H12P	1	Screw	Point 1	- (41-fig. 12) -
NAS1351-5H12P	1	Screw	Point 2	- (42-fig. 13) -
520801-24	2	Washer		
AN315-5-R	1	Nut		
F7XC533534262	1	Bracket	Point 3	- (45-fig. 14) -
35053DE4008	2	Rivet	Point 3	- (46-fig. 14) -
78651RH15CF	1	Clamp	Point 3	- (47-fig. 14) -
NAS9203-8	1	Screw	Point 3	- (48-fig. 14) -
NAS1149D0316K	1	Washer	Point 3	- (49-fig. 14) -

**FALCON 7X**
**7X -056**

(6) Section 7X-056-6 (M980):

Kit P/N SBF7X0056F			
NEW P/N	Q T Y	KEY WORD	ITEM/FIG
<b>SBF7X0056CLAT</b> • 30x14x0.5	1 8	<b>kit including:</b> PTFE protective sheet (Std 28260)	- (50-fig. 16), (50-fig. 17) -

(7) Spare kit for installation of FCS rack conditioning pipes

Kit P/N SBF7X0056G			
NEW P/N	Q T Y	KEY WORD	
120513	2	O-ring	
W932-12F	4	Seal	
W932-20F	2	Seal	

(8) Spare kit to be ordered in case of thermal plate damage during removal of FCS rack

Kit P/N SBF7X0056H			
NEW P/N	Q T Y	KEY WORD	
T84011010501	1	Thermal plate	

(9) Section 7X-056-8

Kit P/N SBF7X0056J			
NEW P/N	Q T Y	KEY WORD	ITEM/FIG
F7XC531524461P01	1	Feeder bracket	- (51-fig. 19) -
35053-3207ID	3	Rivet	- (52-fig. 19) -
78750A06	3	Cable grip	- (53-fig. 19) -
78750-02	20	Cable grip	

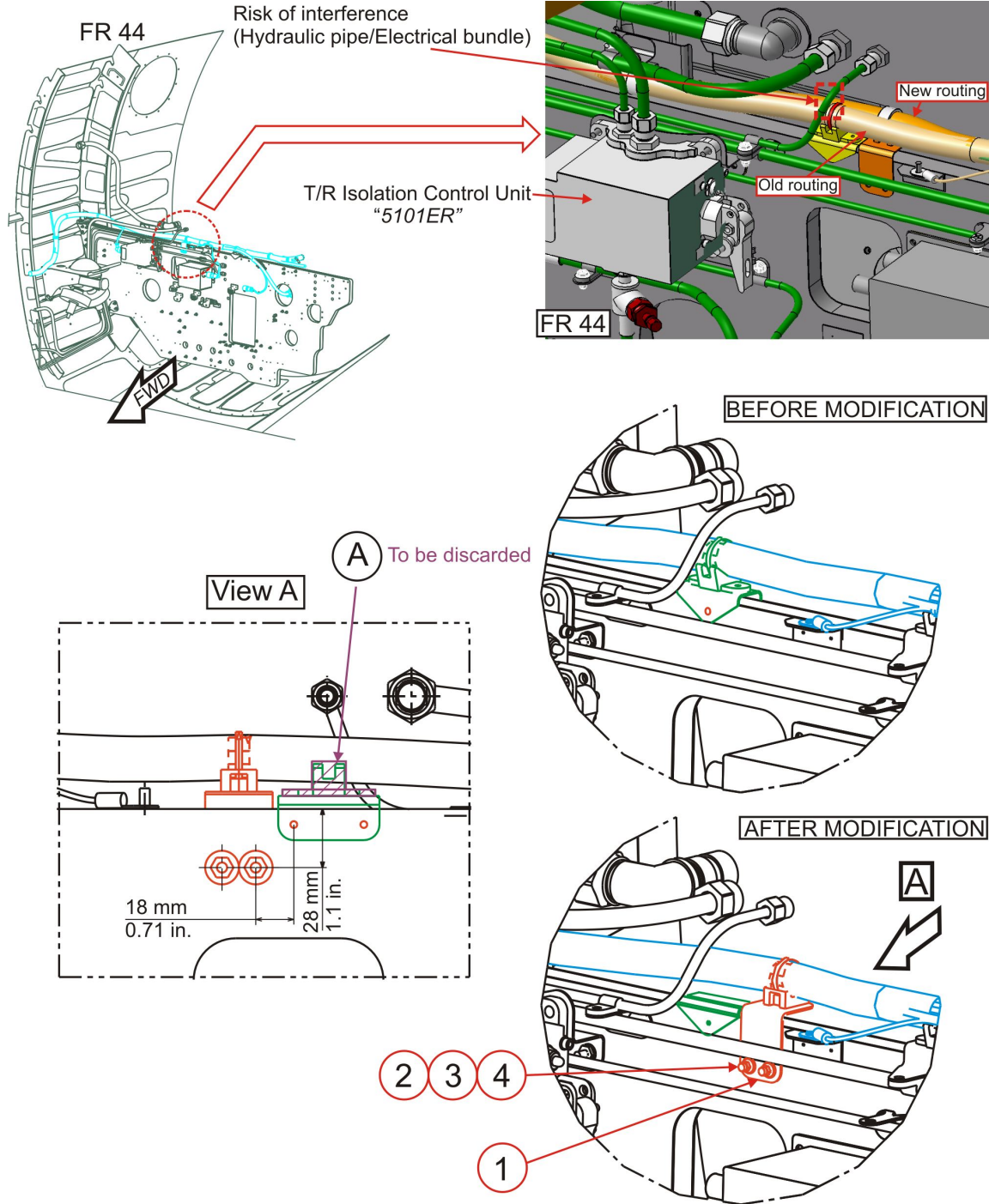
C. LOCAL PROCUREMENT

As indicated in Consumable Products Manual (CPM):

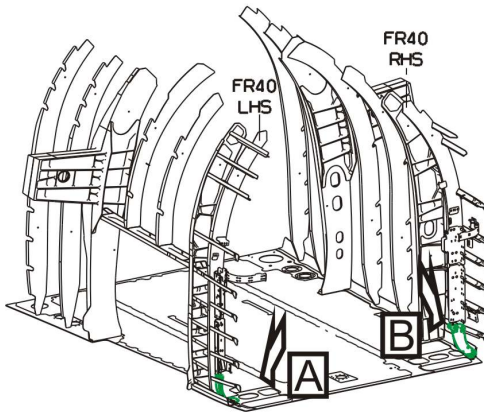
- **cleaner.**
- **adherence promoter for PR.**
- **interlaying sealant.**
- **araldite LY5052.**
- **hardener HY5052.**
- **vinyl varnish.**
- **ROUNDIT type sheath (01PR0NMR19G)**
- **White binding (01FFFTF6W)**

**FALCON 7X**

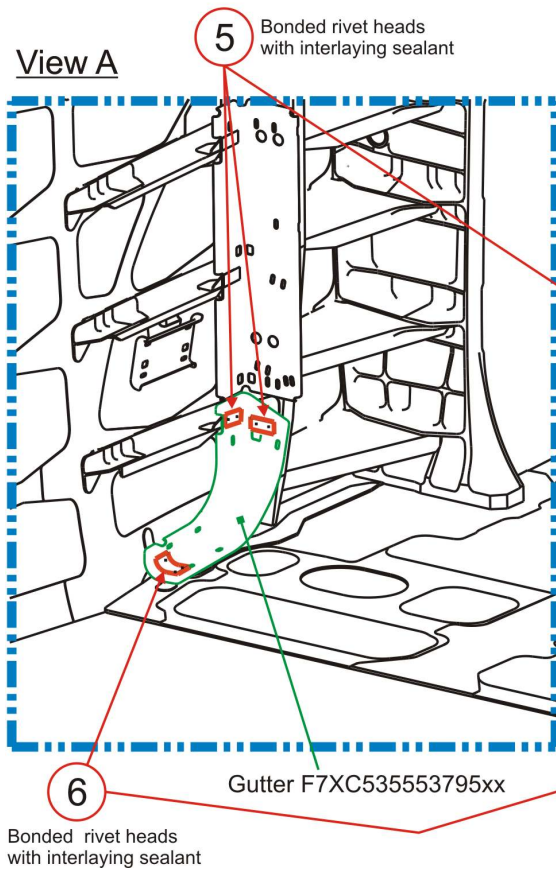
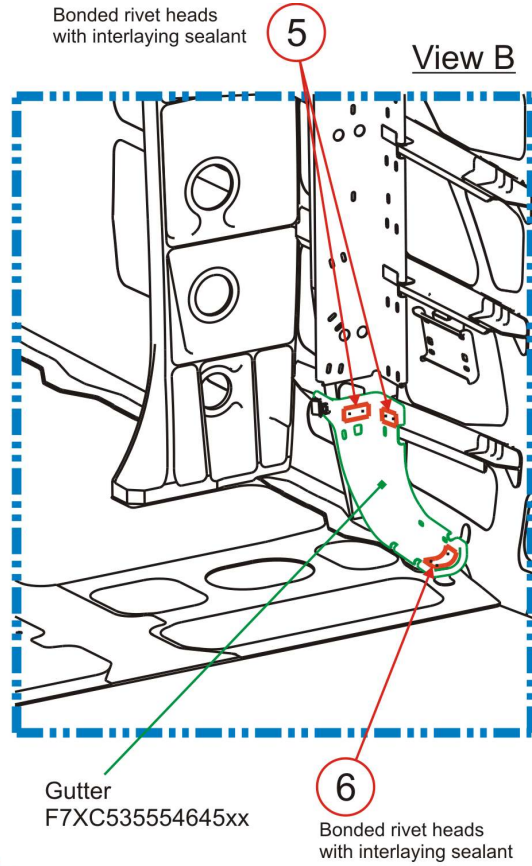
**7X -056**



**Figure 1: SECTION 7X-056-1 (M876)**

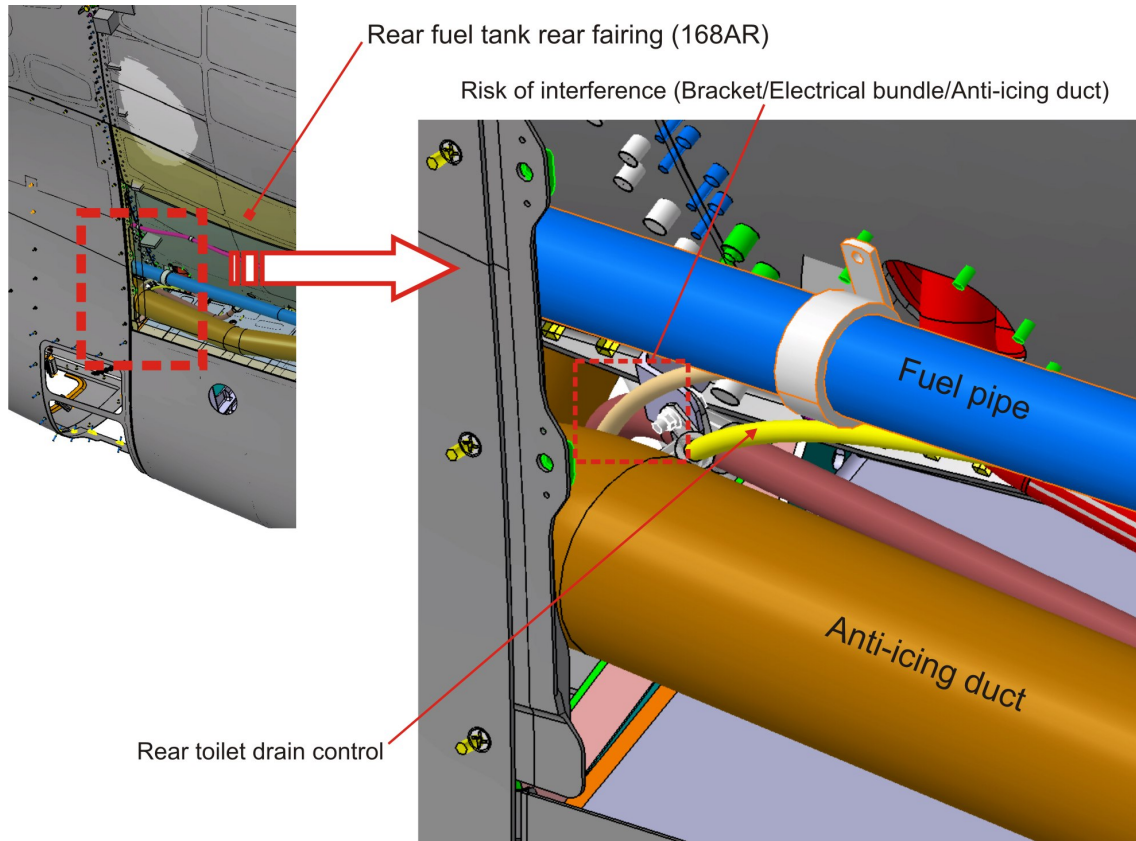


**BAGGAGE BAY**  
(View from Fr 41)

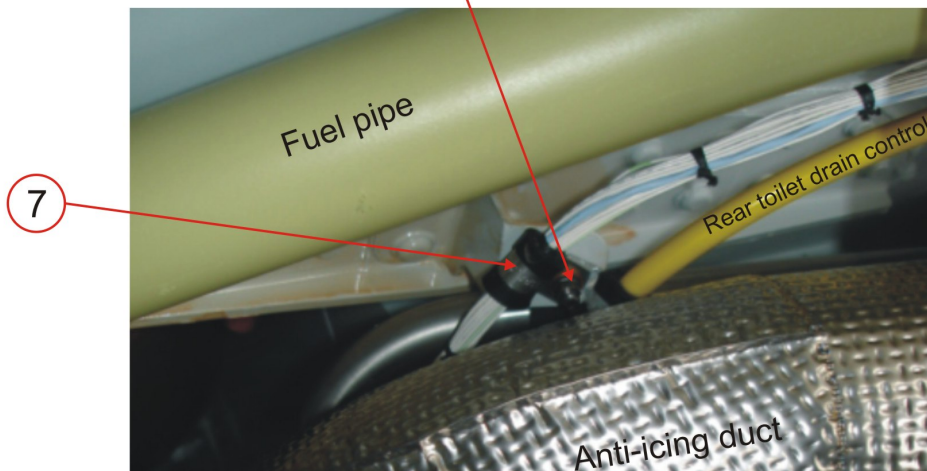


**Figure 2: SECTION 7X-056-2 (M897, POINT 2)**



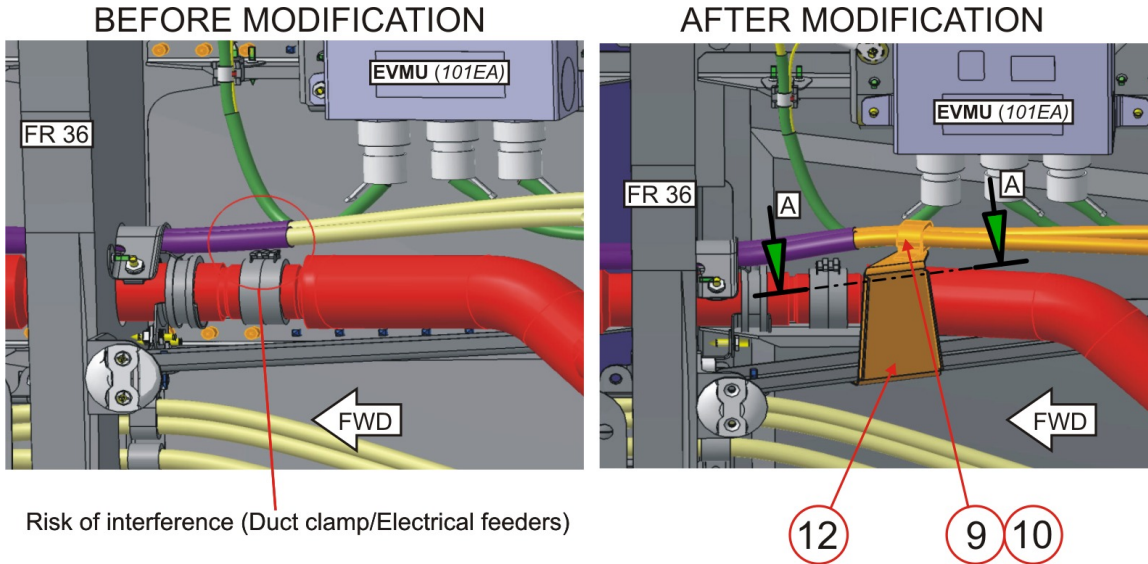


8 Screw NAS9203-13

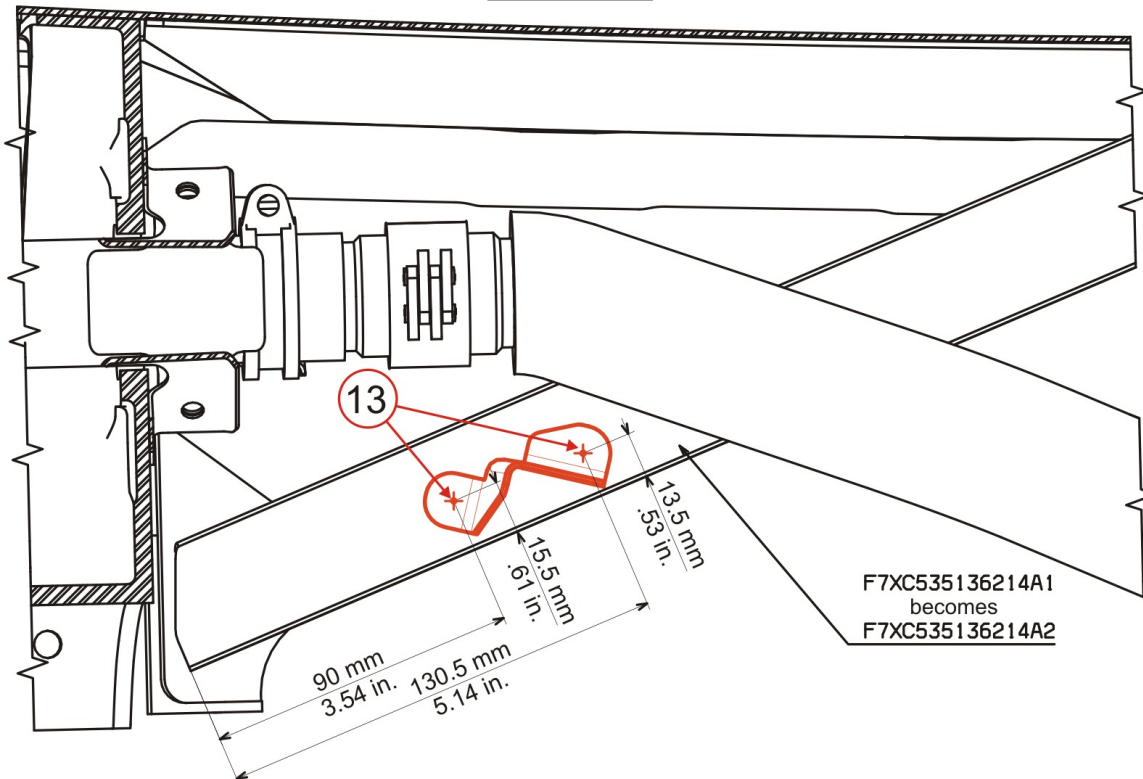


**Figure 3: SECTION 7X-056-3 (M900, POINT 1)**



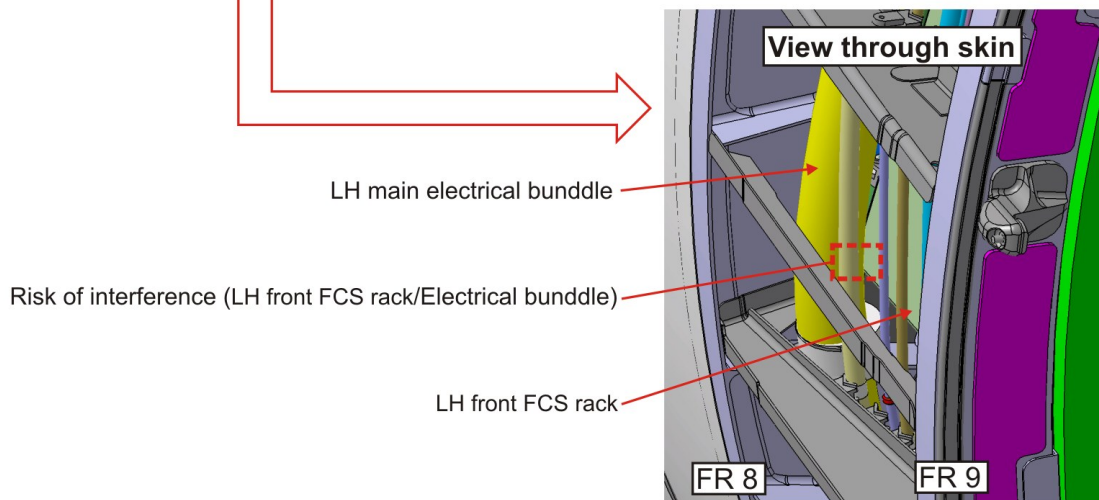
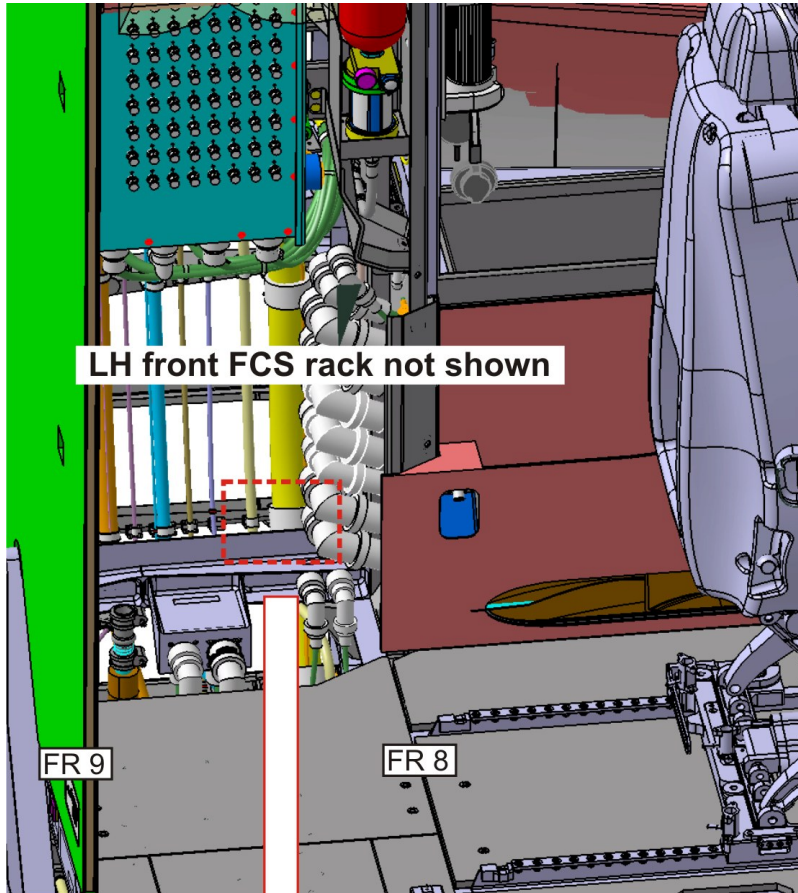


Section AA

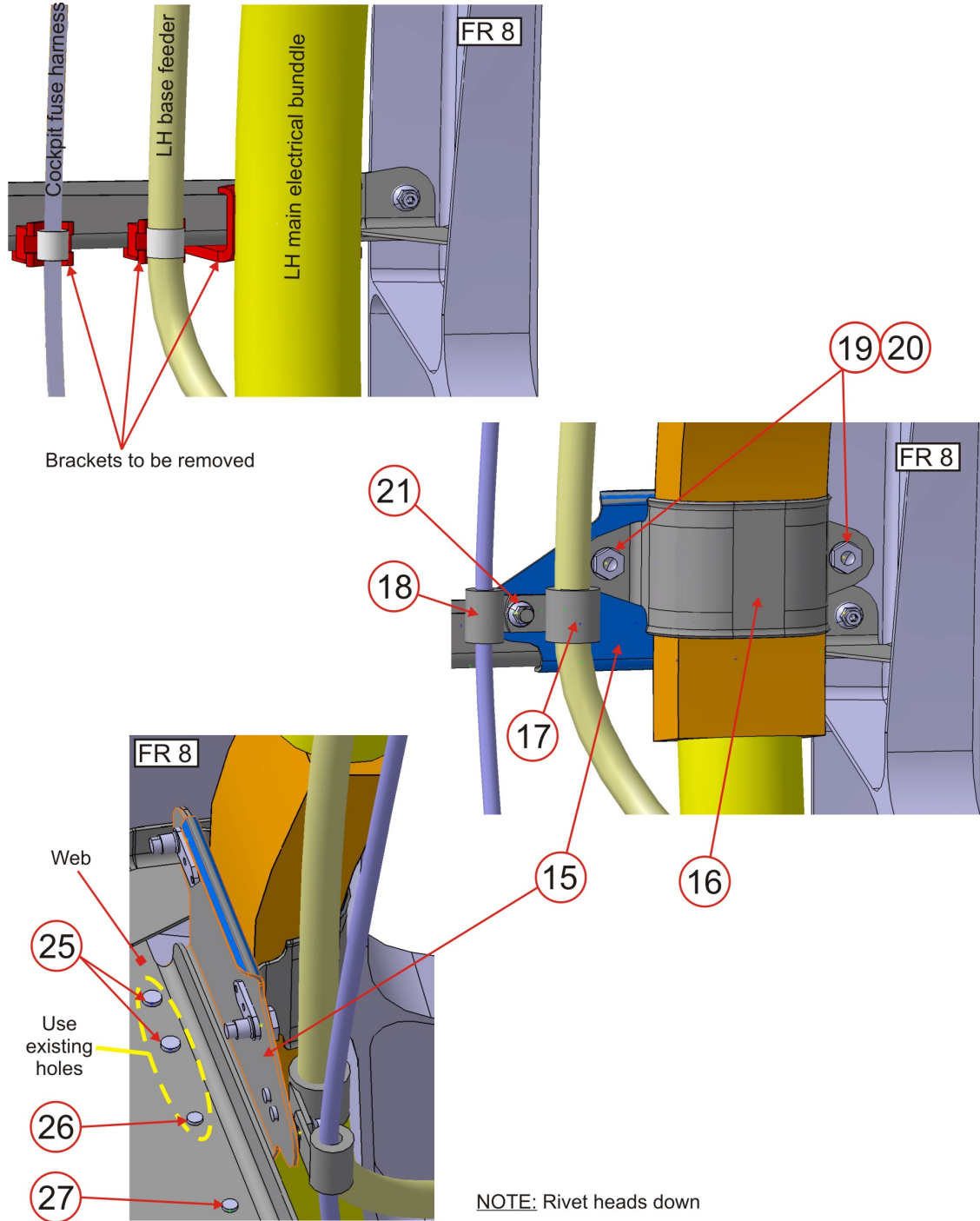


**Figure 4: SECTION 7X-056-3 (M900, POINT 2)**

**FALCON 7X** **7X -056**

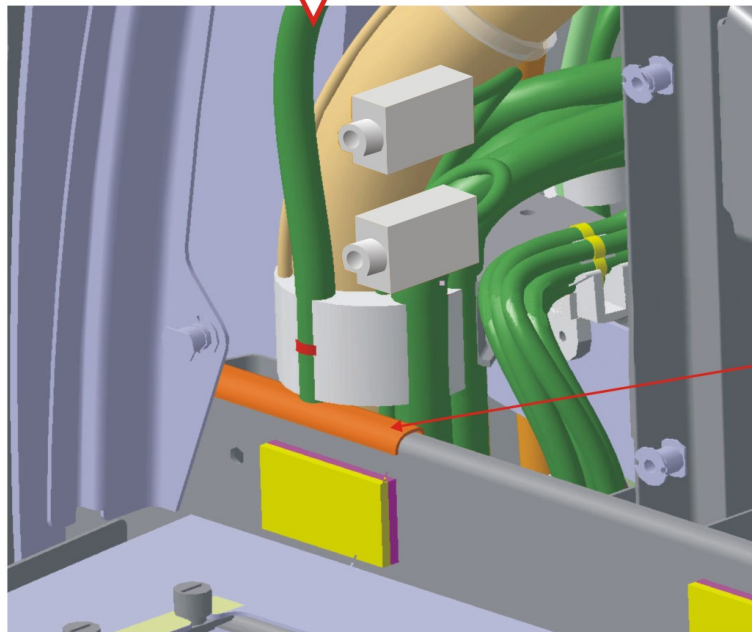
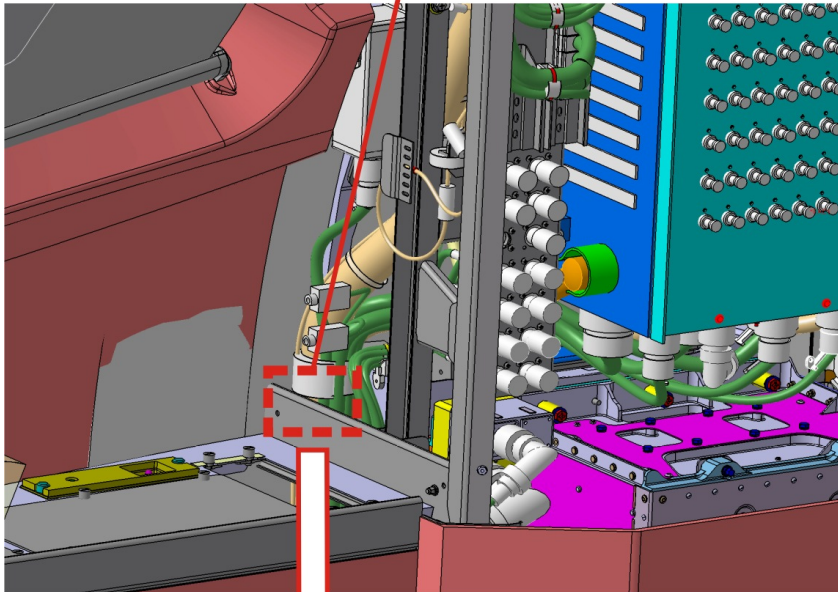


**Figure 5: SECTION 7X-056-3 (M900, POINT 3)**



**Figure 6: SECTION 7X-056-3 (M900, POINT 3)**

Risk of interference (RH cabinet structure/Electrical bundle)



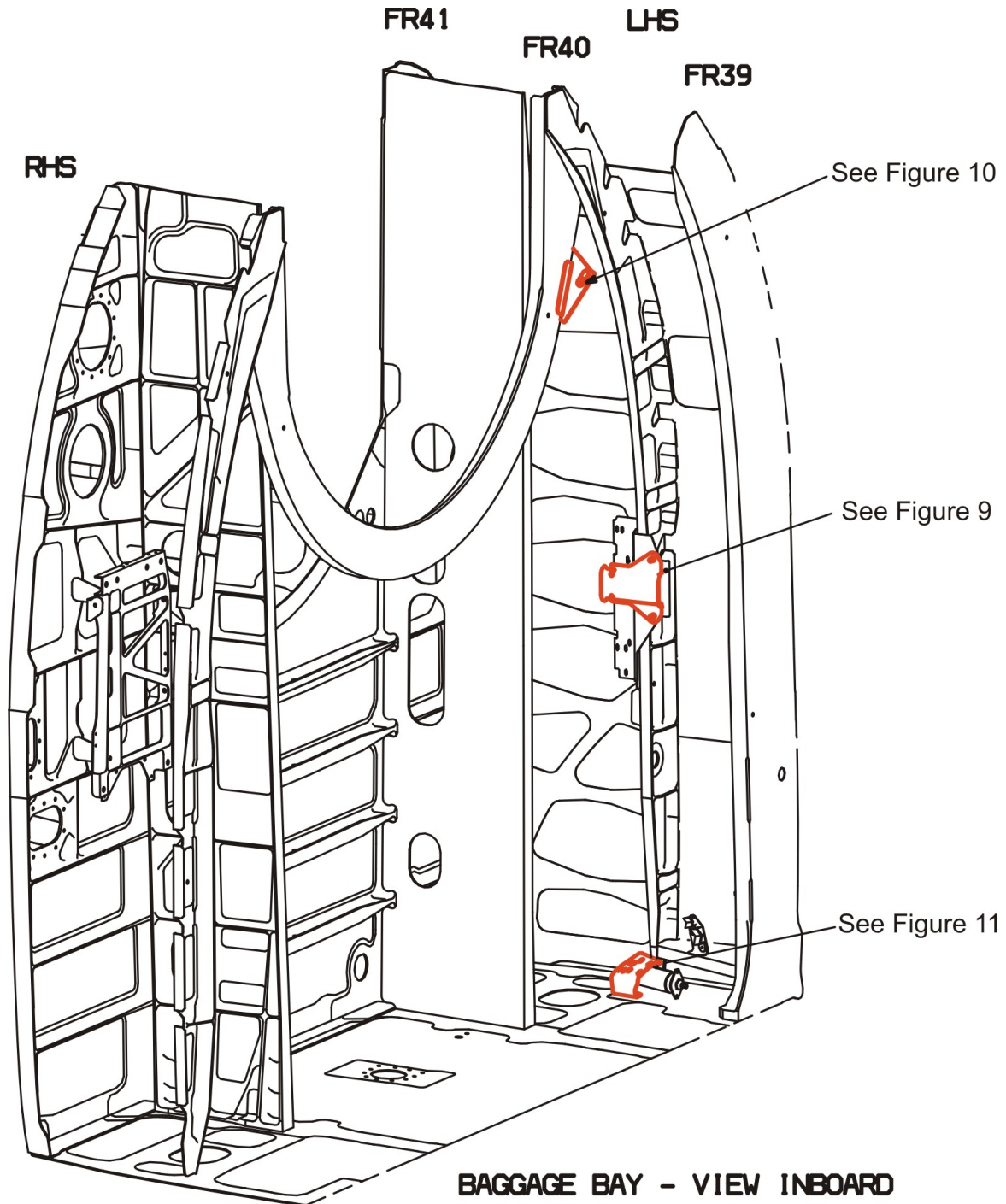
28

Bonded with interlaying sealant

**Figure 7: SECTION 7X-056-3 (M900, POINT 4)**

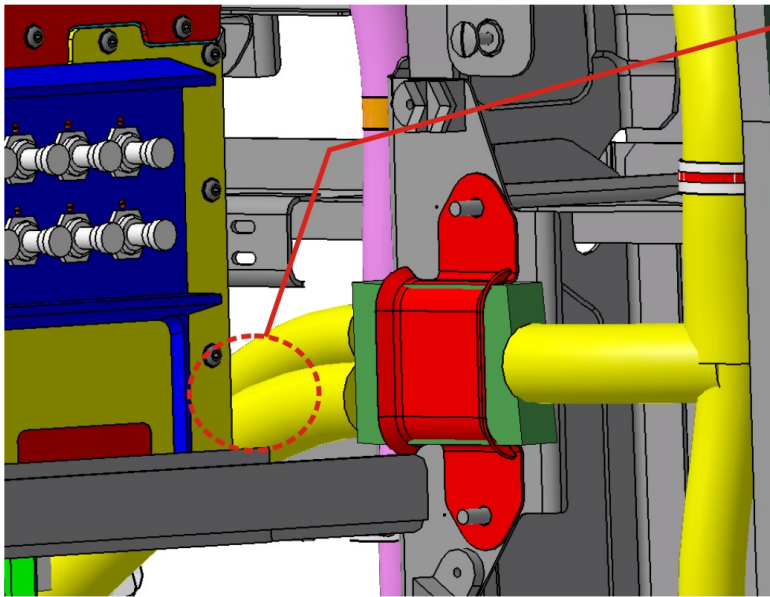


**FALCON 7X** **7X -056**

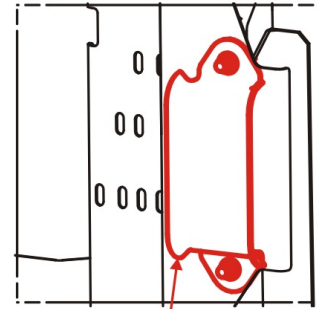


**Figure 8: SECTION 7X-056-4 (M937, POINTS 2, 3 AND 4)**

**BEFORE MODIFICATION**

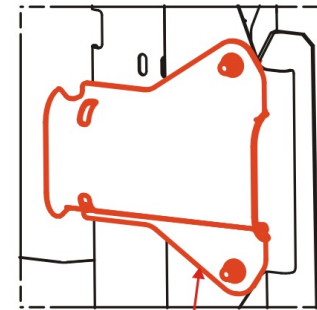
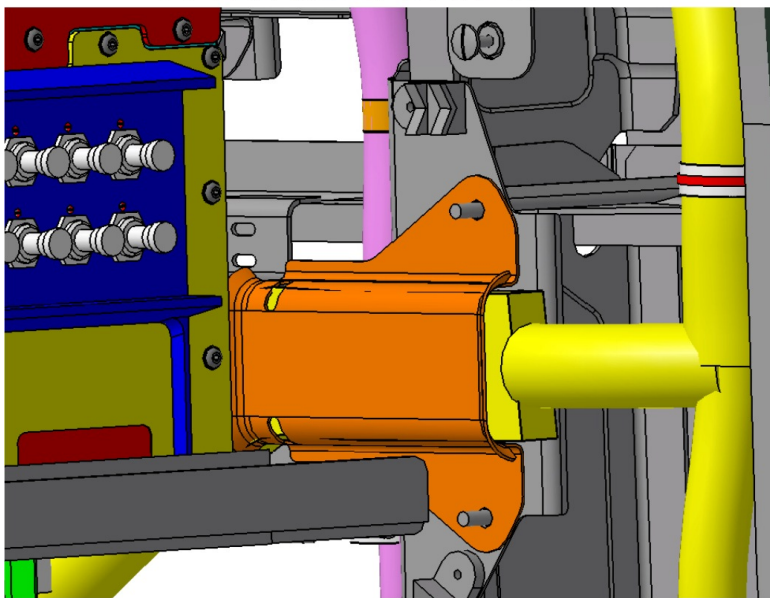


Risk of interference  
(LH Rear SPDB/Electrical bundle)



Protector to be removed

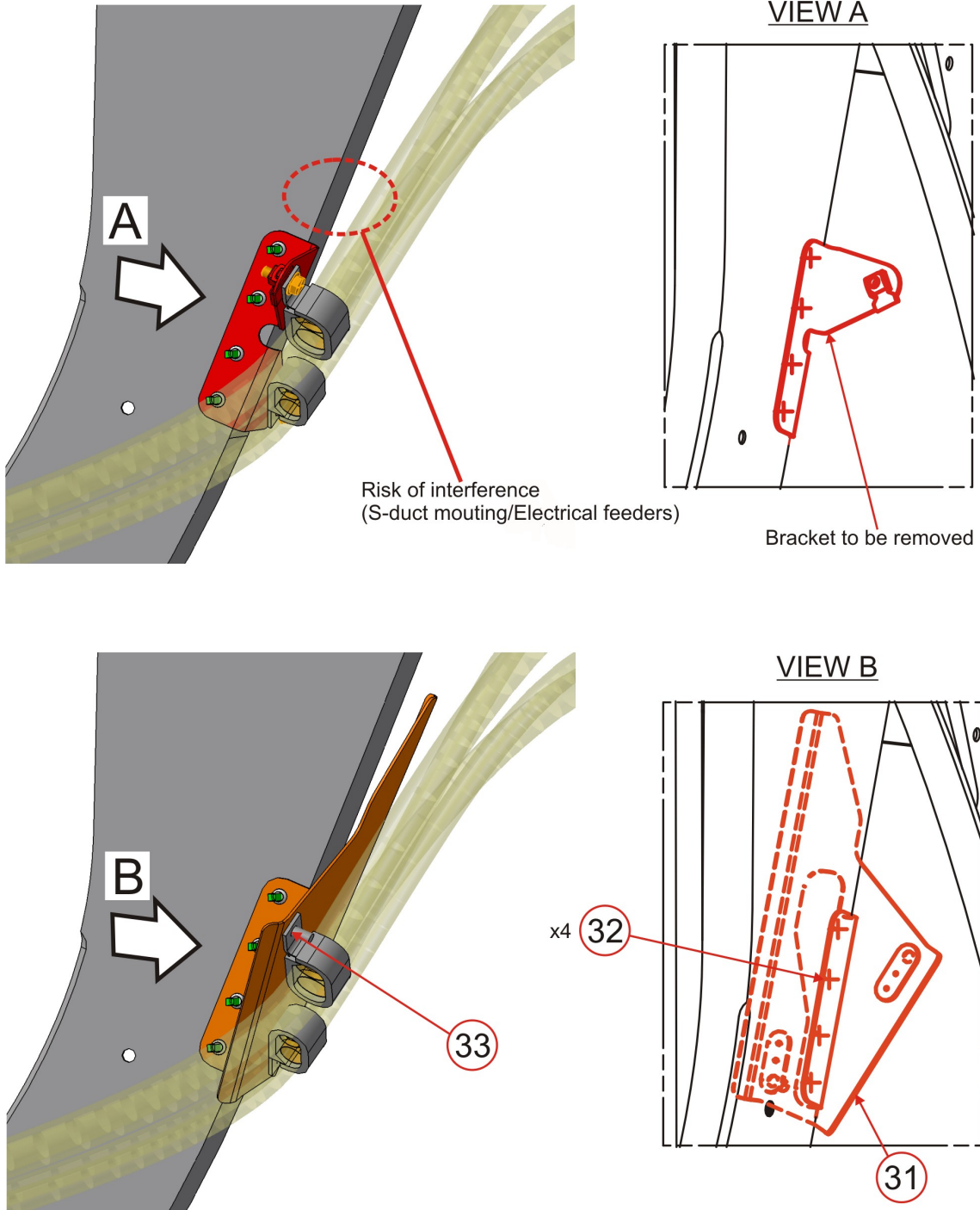
**AFTER MODIFICATION**



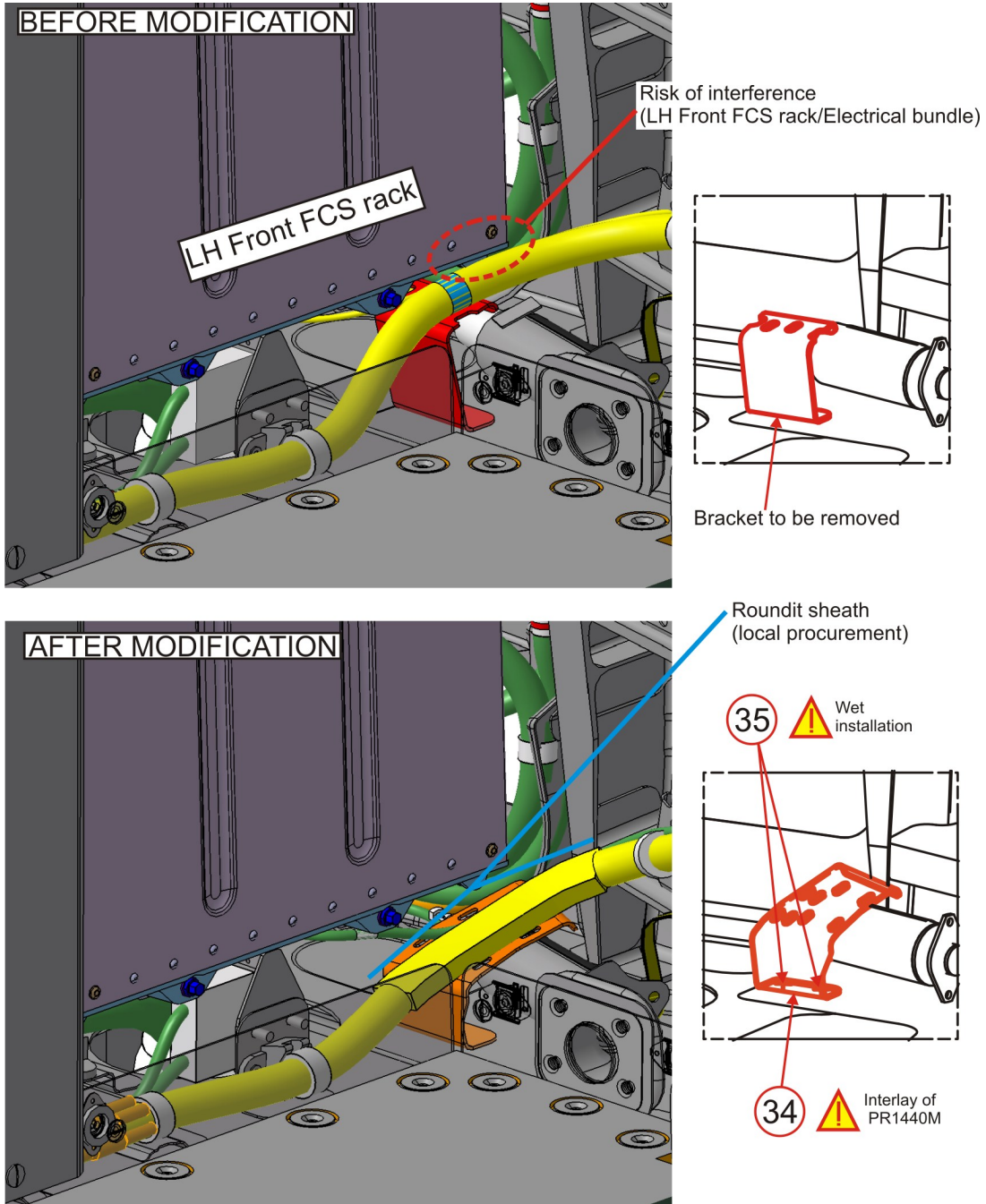
30

**Figure 9: SECTION 7X-056-4 (M937, POINT 2)**

**FALCON 7X** **7X -056**

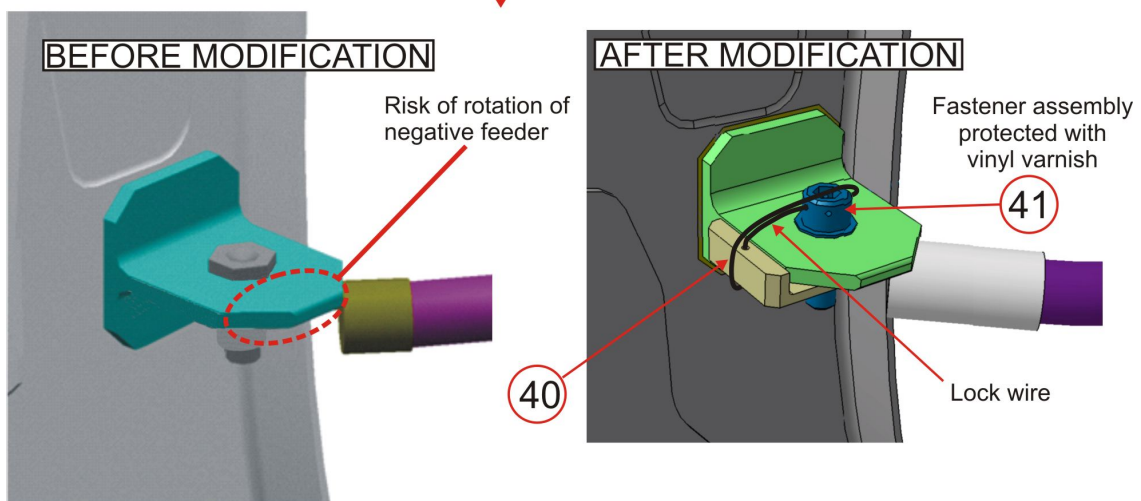
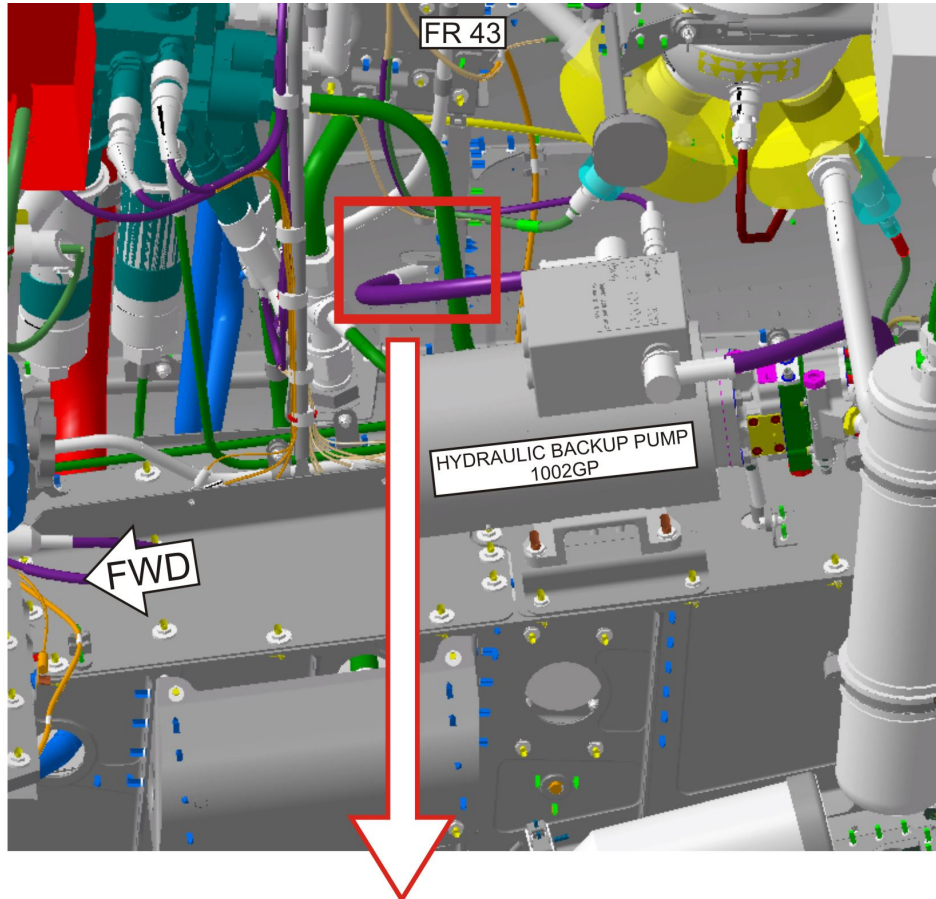


**Figure 10: SECTION 7X-056-4 (M937, POINT 3)**

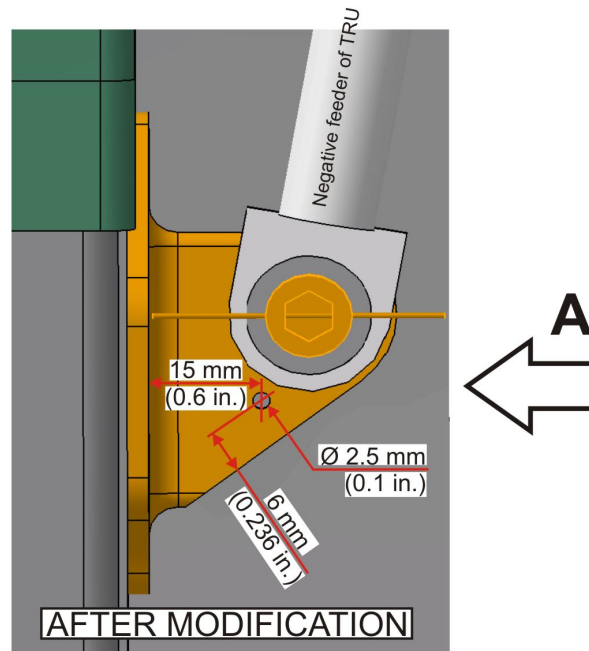
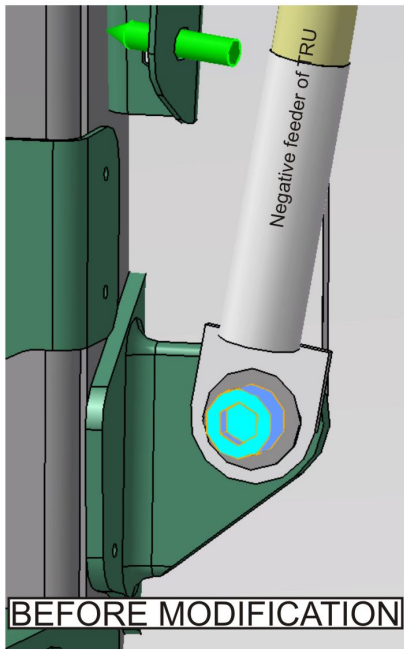
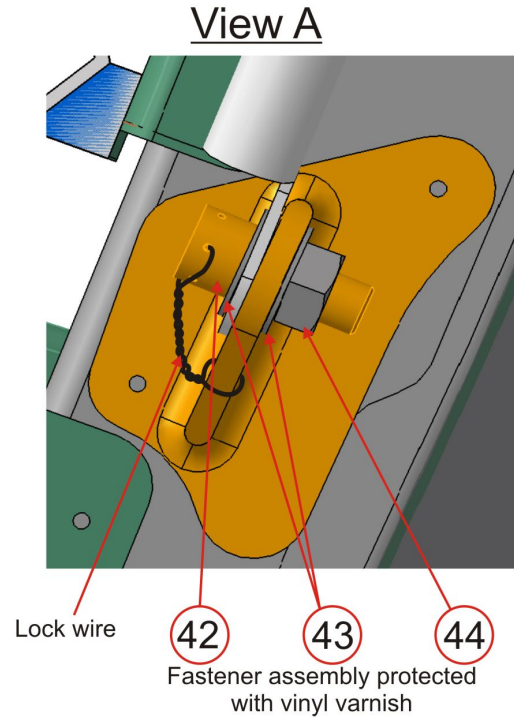
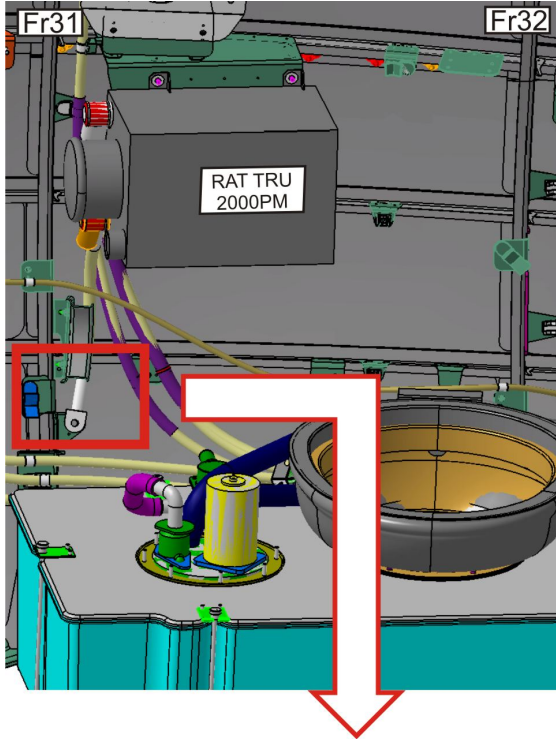


**Figure 11: SECTION 7X-056-4 (M937, POINT 4)**





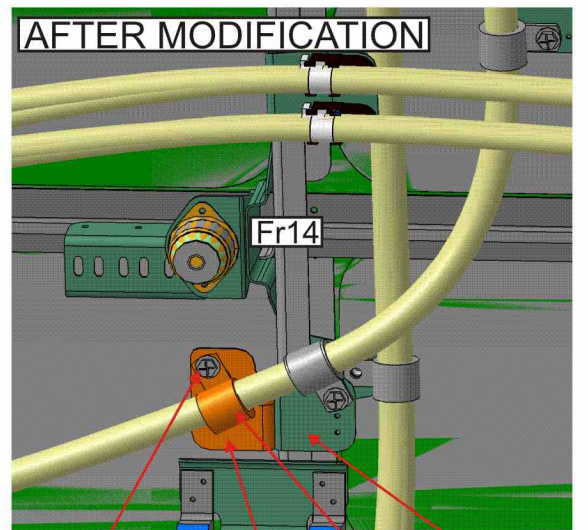
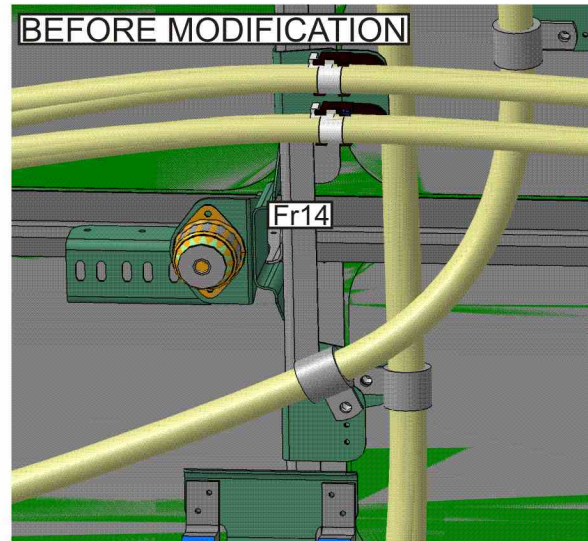
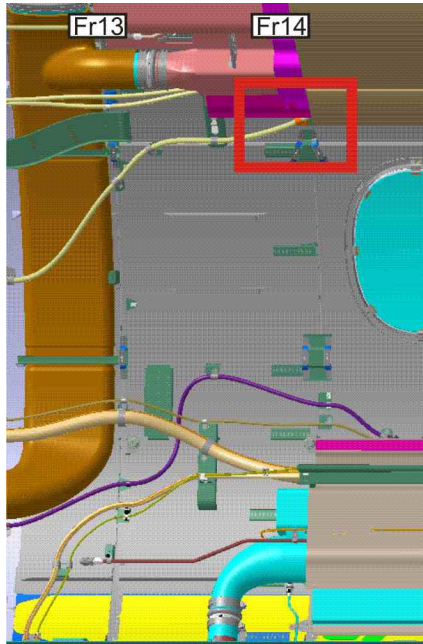
**Figure 12: SECTION 7X-056-5 (M954, POINT 1)**



**Figure 13: SECTION 7X-056-5 (M954, POINT 2)**

**FALCON 7X**

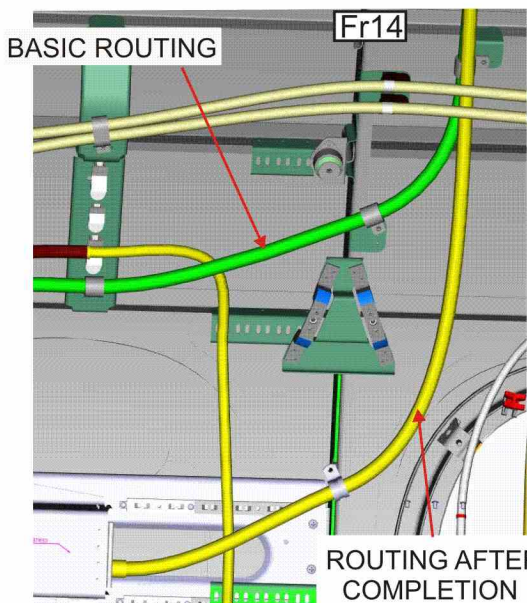
**7X -056**



- 48
- 49
- 45
- 46
- x2
- 47
- A

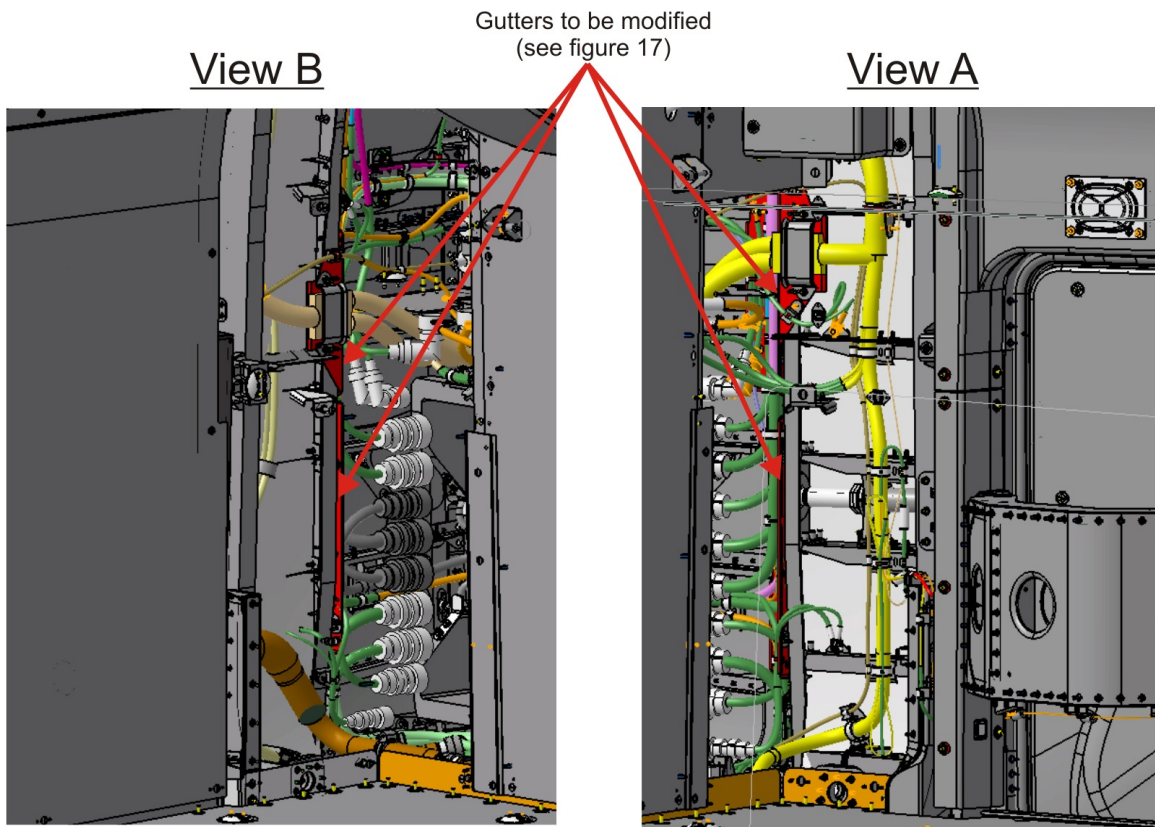
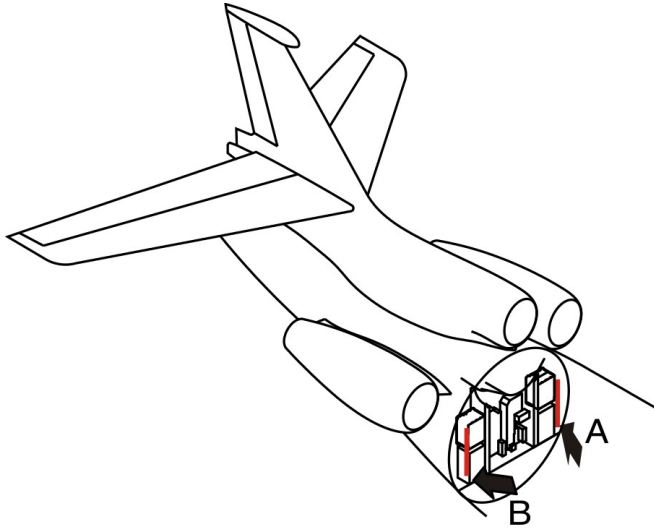
Replace existing rivets of bracket (item A)

**FEEDER REROUTING IN COMPLETION**



**Figure 14: SECTION 7X-056-5 (M954, POINT 3)**

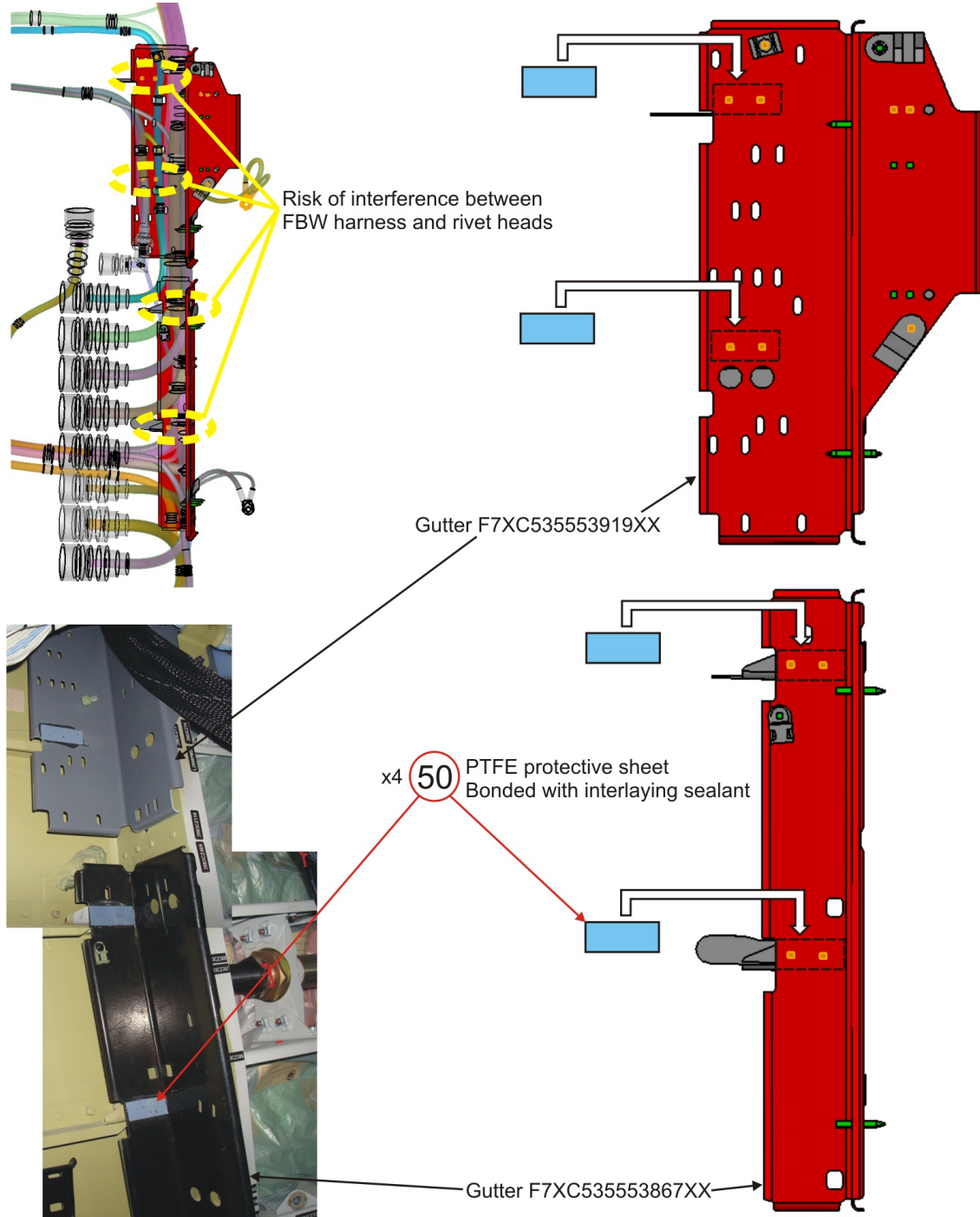




**Figure 15: SECTION 7X-056-6 (M980)**

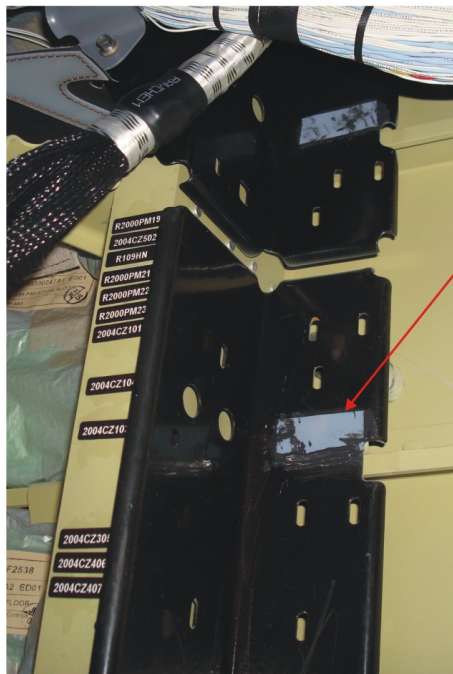
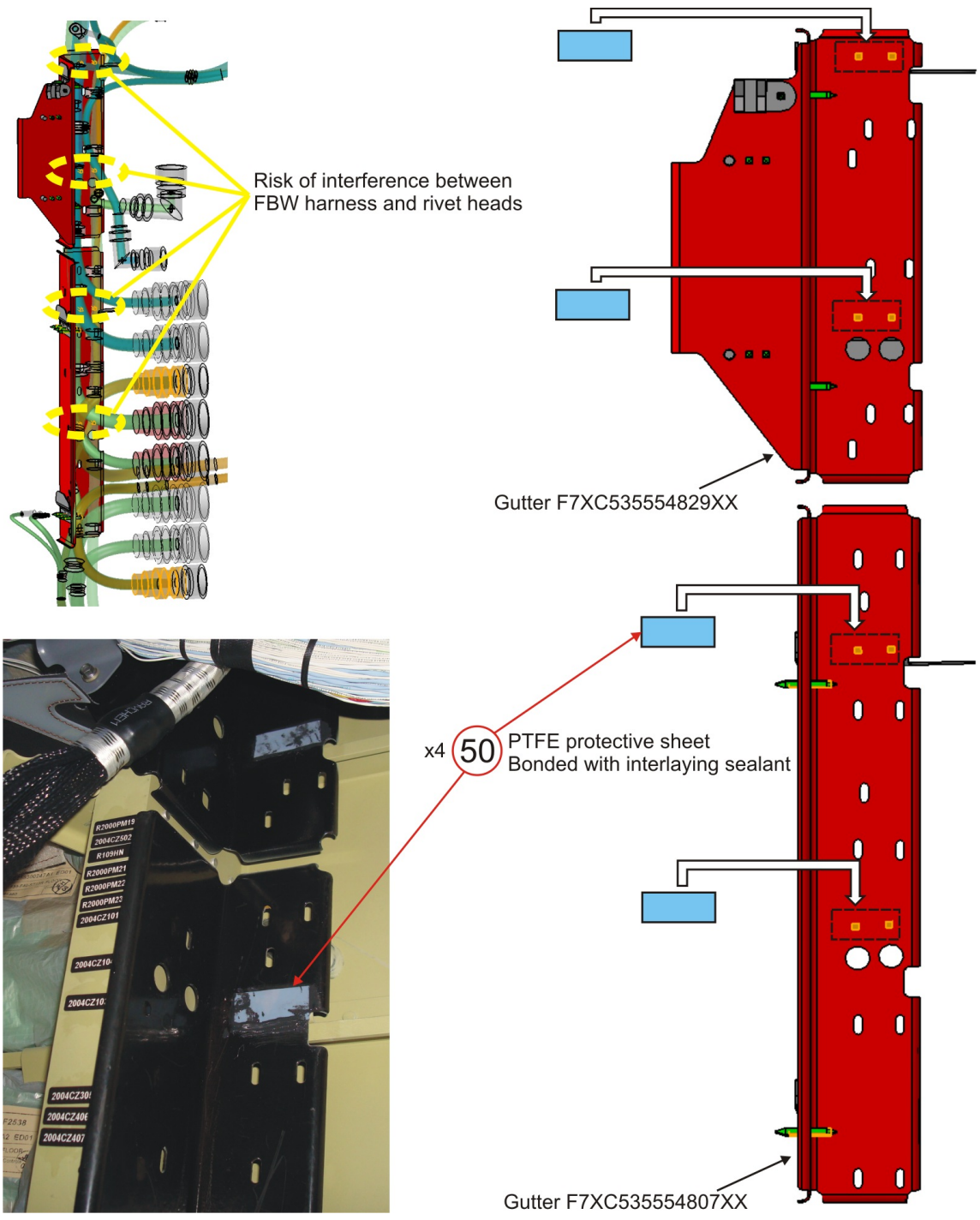
**FALCON 7X**

**7X -056**

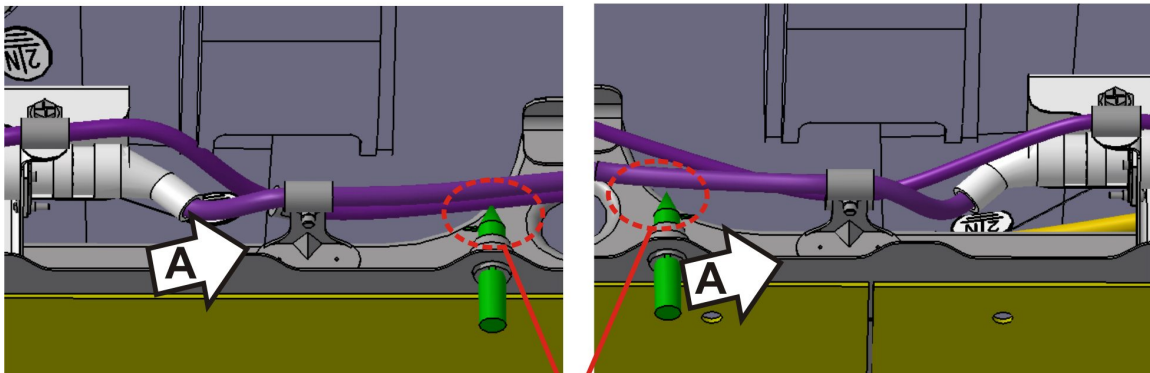
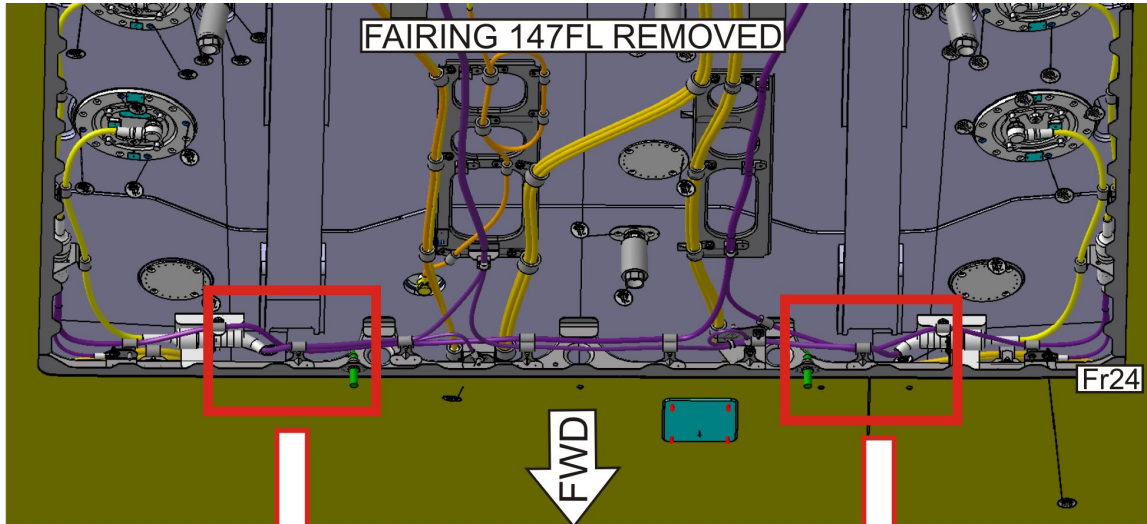


**Figure 16: SECTION 7X-056-6 (M980)**

**FALCON 7X** **7X -056**



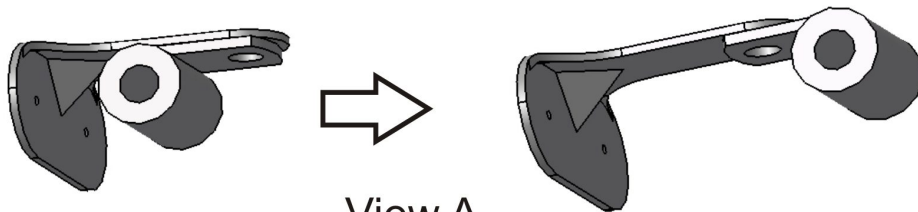
**Figure 17: SECTION 7X-056-6 (M980)**



Risk of interference between fuel booster pump harnesses and door screw

**BEFORE MODIFICATION**

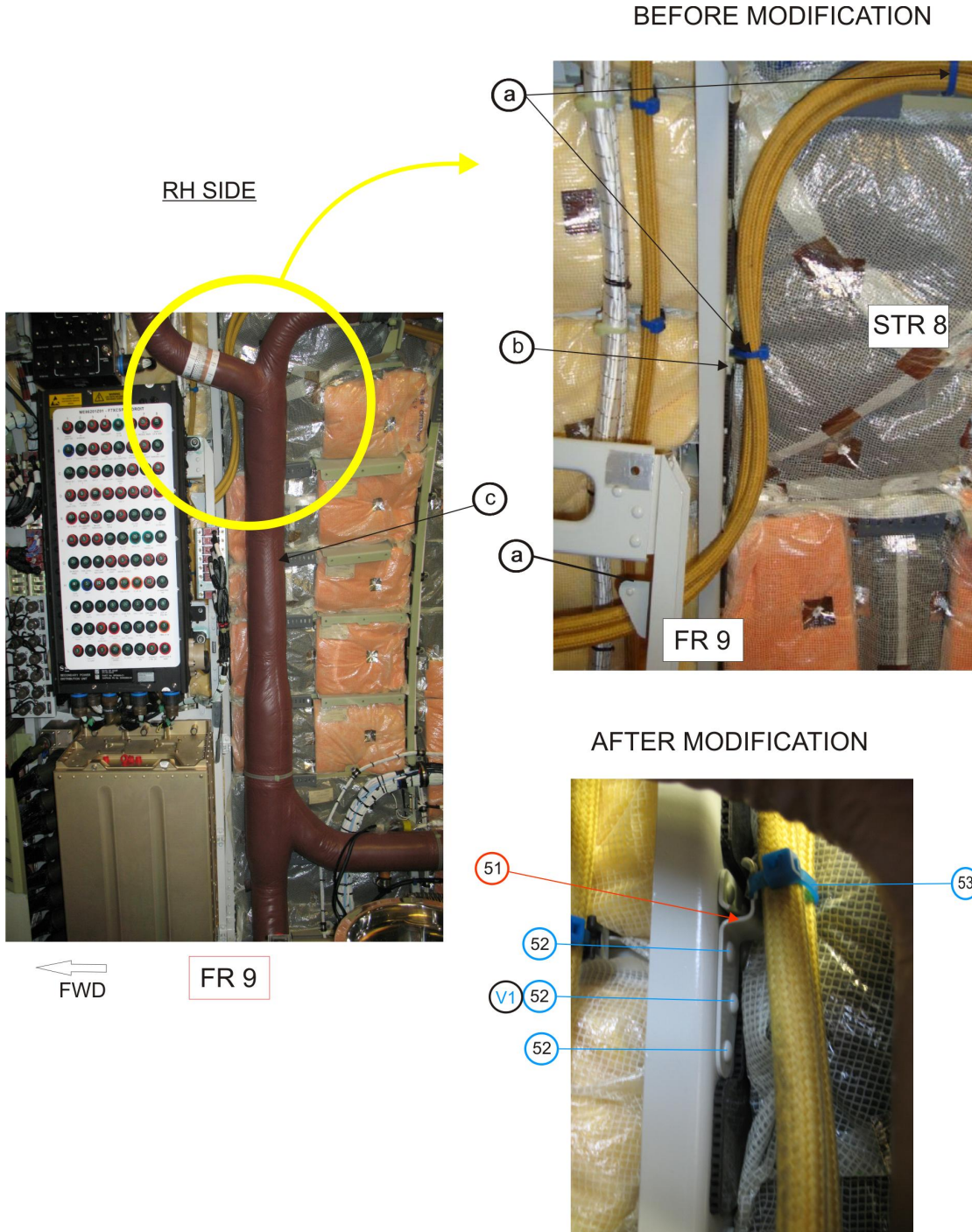
**AFTER MODIFICATION**



**View A**

**Figure 18: SECTION 7X-056-7 (M1021)**





**Figure 19: SECTION 7X-056-8 (M1470)**